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CORONA J  
PERFORMANCE EVALUATION REPORT  
MISSIONS 1006-1 & 1006-2  
FTV 1176 - J9

26 October 1964

Approved by [REDACTED]

11-18-64  
Date

Advanced Proj [REDACTED]

Approved by [REDACTED]

11/18/64  
Mgr. Date

Program [REDACTED]

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FORWORD

This report details the performance of the payload section of the Program  
[REDACTED] Vehicle 1176.

Lockheed Missiles and Space Company has responsibility for evaluating  
payload performance under the System Integration and the "J" System  
Contracts.

This document is the final payload test and performance evaluation report  
for Missions 1006-1 and 1006-2 which was launched on 4 June 1964.

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## INTRODUCTION

### A. PURPOSE

The purpose of this Performance Evaluation Report is 1) to define and establish the performance parameters of Mission 1006-1 and 1006-2, and 2) to identify the problem areas associated with these missions and recommend the appropriate corrective action.

### B. INFORMATION SOURCES

The information employed in this report is from government facilities and their sources are:

AFSPLL - Diffuse Densities, RES, and Edge Trace

NPIC - Vehicle attitude data, Frame Correlation, [REDACTED] Edge Trace Reports and Processing Summary

ACIC - Special Yaw Error Reduction

### C. PHASES OF EVALUATION

The evaluation was conducted in two phases. Phase one, the in-line, utilized the original negative and duplicate positive material generated during these missions and was conducted at customer facilities (AFSPLL, NPI, ACIC). The total mission material was available for the in-line evaluation. Phase two, the off-line evaluation, utilized only that material which was generated from the engineering passes over the continental United States and was conducted at the Advanced Projects facilities of LMSC.

The effort involved in conducting such an evaluation, and the personnel involved in such an effort, may be more than is readily apparent. The in-line evaluation was conducted at the customer facilities during other evaluation activities, utilizing personnel supplied by LMSC Advanced Projects and their personnel whose knowledge and experience makes them qualified to conduct the evaluation. The clients were [REDACTED]. Its participants were [REDACTED]

The off-line evaluation was conducted at the Advanced Projects facilities with equally capable personnel utilizing material obtained from the engineering passes over the Zone of Interior. Advanced Projects' participation, in addition to the above, includes [REDACTED] and [REDACTED]

SECTION I  
SYSTEM PERFORMANCE

**A. MISSION OBJECTIVES**

Flight Test Vehicle No. 1176 payload consisted of two Panoramic cameras, two Stellar-Index cameras, two Mark 5 recovery capsules, and a space structure to enclose the cameras and provide mounting for the recovery subsystems. The space structure also includes the payload clock, payload telemetry, and electrical cabling and junction boxes. This equipment is designed to obtain search reconnaissance photography of selected areas of the earth.

The planned mission was a two-phase 8 day camera operational mission with no deactivate period.

**B. FLIGHT RESULTS**

The mission was launched from Vandenberg Air Force Base at 3:59:22 PM PDT on 4 June 1964. The first phase was completed and a successful air catch was made on orbit 65 on 8 June 1964. The second phase of the mission was completed on orbit 128 on 12 June 1964 with a successful air catch recovery of the second capsule.

The comparison between predicted and actual orbit parameters is tabulated below:

	<u>PLANNED</u>	<u>ACTUAL</u>
Inclination	80.00°	79.97°
Period	90.87'	90.59'
Perigee Altitude	99.92 nm	84.00 nm
Initial Perigee Location	40.60°N	63.20°N
Apogee Altitude	250.50 nm	261.00 nm
Eccentricity	0.0209	0.0239

The space structure door covering the slave panoramic camera did not eject on command and remained in place through pass D02. Prior to the photographic operations during pass D03 the door was ejected.

### C. PANORAMIC CAMERAS

The Master and Slave panoramic cameras functioned satisfactorily throughout both missions and produced excellent photography despite the V/H errors caused by orbit mismatch. In part this is due to the 10-15% scale improvement resulting from the low perigee. It was possible to observe motor vehicles and identify aircraft types throughout the mission indicating that the cameras produced high resolution photography. No ground resolution targets were photographed. Microdensitometer edge trace measurements by [redacted] reduced to resolution in lines per millimeter indicate that the Fwd camera averaged 84 lines and the Aft camera averaged 87 lines. However, these measurements show a high coefficient of dispersion which indicates caution should be used in interpreting the results. Detailed comments on the edge trace measurements may be found in Section VII of this report.

Intermittent failures in the center-of-format switch closures in both cameras caused the loss of some auxillary data and some Stellar-Index photography.

### D. STELLAR INDEX CAMERAS

The Stellar-Index camera operations during Mission 1006-1 were satisfactory for approximately 60% of the mission life. The stellar shutter either failed to close, failed to open or produced multiple exposures during the remainder of the mission. The index camera operated properly throughout Mission 1006-1 however film depletion occurred twenty frames early. Both the stellar and terrain photography were rated good during proper camera operation.

The Stellar-Index camera functioned properly through frame 274 of Mission 1006-2. The faulty Master camera center-of-format switch permitted only five additional camera operations to the end of the mission. All of the photography required was rated good.

### E. MISSION SUMMARY

A brief summary of Missions 1006-1 and 1006-2 is shown on Tables 1 and 2 on pages 4 and 5. A schematic inboard profile is shown on page 5A.

### F. CONCLUSIONS

Missions 1006-1 and 1006-2 achieved the major objective of obtaining search reconnaissance photography.

GRADE  
[redacted]  
[redacted]

Payload No. J-9	Mission No. 1006-1	Booster No. 403	Vehicle No. 1170	SRV No. 638
Launch Date 6-4-64	Launch Time 2259Z	Recovery Date 6-8-64	Recovery Time 0600Z	Mode A
Orbit Parameters -	Planned Actual (Rev. 1)	Vehicle Attitude -	Error (90%)	Range
Inclination	80.00°	Pitch	0.41°	0.41° - 0.63°
Period	90.87	Roll	0.92°	0.92° - 0.98°
Perigee Altitude	59.92 NM	Yaw	1.14°	0.14° - 0.31°
Perigee Location	84.00N 63.20W	Resolution Limit	Along Track (90%)	2.1. 8°/NM
Apogee Altitude	10.60N 250.50W	Cross Track (90%)	13.8°	0.1° - 24.5°
Eccentricity	0.0289		6.7°	Range 0° - 12.4°
Instrument	MASTER (F/T)	SLAVE (AFT)	STELLAR	INDEX
Serial No.	148	149	D45/47 / 45	D45/47 / 45
Pre-Flight Resolution	181 W.C., 110 L.C. 0.200"	181 W.C., 106 L.C. 0.200"	71 W.C.	71 W.C.
Slit or Exposure Time	W. 21/4400	W. 21/4404	1/500 (64.5)	1/500 (64.5)
Filter/Film	2883 frames (1943)	2881 frames (1922)	W-21/4400	W-21/4400
Footage Recovered	All - 83.7 HIGH 91.9 350.4-63.45-80.83-90	All - 86.7 HIGH 90.9 350.4-71.43-90, RES-BY Very Good	417 frames	317 frames
Photo Quality E&E traces	Very Good	Very Good	Fair. Shutter failed to open 36 times.	Good. Static shot one frame.
Exposure	Majority good, some over exposed 35%, minor 60%	Majority good, some over exposed 35%, minor 60%	Intergalactic star field patterns aberrant	Good
Density	Primary L	Primary L	Full	Low 8%, med 50%, high 60%
Processing	227.0	147.0	160	15.4% (- 2.31% to +2.0%)
V/H Error (90%)			OK	
Time Track			OK	
Data Block			OK	OK
End of Pass Mark		OK	OK	OK
Port H. O. Quality (6/6.8)	6/6.8	6/6.8	6/6.8	6/6.8
Port H. O. Fiducials	O.K.	OK	OK	OK
Starboard H. O. Quality (6/6.8)	Slight out of focus. D-35	Soft overexposed	OK	OK
Starboard H. O. Fiducials	OK	OK	OK	OK
S/I Correlation Lamp	Normal	OK	One grossly overexposed	OK
Temperature			OK	OK
Remarks	Minor edge static. No camera	Slave does not have power at D-35-1. Camera fails D-10 through D-33-17. No static or camera	Slave goes to power out due film to frame break.	out due film to frame break.

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Mission No.	Launch Time	Booster No.	Vehicle No.	SRV No.
Launch Date	Actual (Rev.)	Recovery Date	Time Rev	Mode AIR Range
Orbit Parameters - Planned	Inclination	Vehicle Attitude	Error (90%)	Range
J-9	2259 Z	403	1176	639
Launch Date 6-4-64	Planned	6-12-64	128	Recovery Mode AIR Range
Orbit Parameters -	Inclination	Pitch	0.19°	0.10°/sec 100' sec
Period	90.81°	Roll	0.19°	0.08°/sec 74.7 sec
Perigee Altitude	43.12 nm	Yaw	1.08°	1.10°/sec 0.39 sec
Perigee Location	40.00°N	Resolution Limit	Alng Track (90%)	3.0°/hr
Apogee Altitude	260.50 nm	Cross Track (90%)	10.1°	Range 0.1° to 15.0°
Eccentricity	0.010		7.0°	Range 2.1° to 15.0°
Instrument Serial No.	MASTER (F/D)	SLAVE (AFT)	STELLAR	INDEX
Serial No.	148	142	D49/53 /42	D49/53 /42
re-Flight Resolution	181 H.C., 110 L.C.	181 H.C., 106 L.C.	72 H.C.	72 H.C.
Limit of Exposure Time	0.200"	0.200"	2 sec	2 sec
Filter/Film	W-21 / W404	W-21 / W404	None / W401	1/100 (4/45)
Voltage Recovered	2471 frames (1879)	3021 frames (8005)	219 frames (28')	YY-U/4400
Photo Quality	EDGE TRACE 3/4	ALL 83.6, HIGH 93.9	FAIR. FIVE TRANSMITTER	119 frames (58')
Photo Quality	EDGE TRACE 3/4	3504, 72, 43.4, 40, RES. 89	FRAMES AFTER Diag-1	GOOD. 4/12 PROBLEM.
Exposure Intensity	Very Good	Very Good	AGGREGATE	GOOD
Exposure Intensity	Good. Best obtained to date	Same	Same	Same
Processing	New 5% more 45%, High 50%	Primary 33	Same 18	High 45%
Processing	Primary 30	Full 24	Same 25	High 45%
Line Error (90%)	11.67% (-16.07 to +6.07%)	OK	OK	OK
Line Track	OK	OK	OK	OK
Data Block	OK TO DATA SCENIC	OK	OK	OK
End of Pass Mark	Vehicle after due to failure	c/e switch	c/e switch	c/e switch
Port H. O. Quality	Same as Data Block	OK	OK	OK
Port H. O. Fiducials	QAD. SCENIC & SMOOTH	OK	OK	OK
Starboard H. O. Quality	APPRO. Diag-1	OK	OK	OK
Remarks	Starboard H. O. Fiducials	51°F to 69°F	51°F to 69°F	51°F to 69°F
	// Fiducials	c/e switch SCENIC	c/e switch	c/e switch
	// Correlation Lamp	APPRO. Diag-1	APPRO. Diag-1	APPRO. Diag-1
	Temperature	INTERRUPTED DUE SCENE	INTERRUPTED DUE SCENE	INTERRUPTED DUE SCENE
		STRET-UP COCKAUA	STRET-UP COCKAUA	STRET-UP COCKAUA
		SCREATCHES NONE D	SCREATCHES NONE D	SCREATCHES NONE D

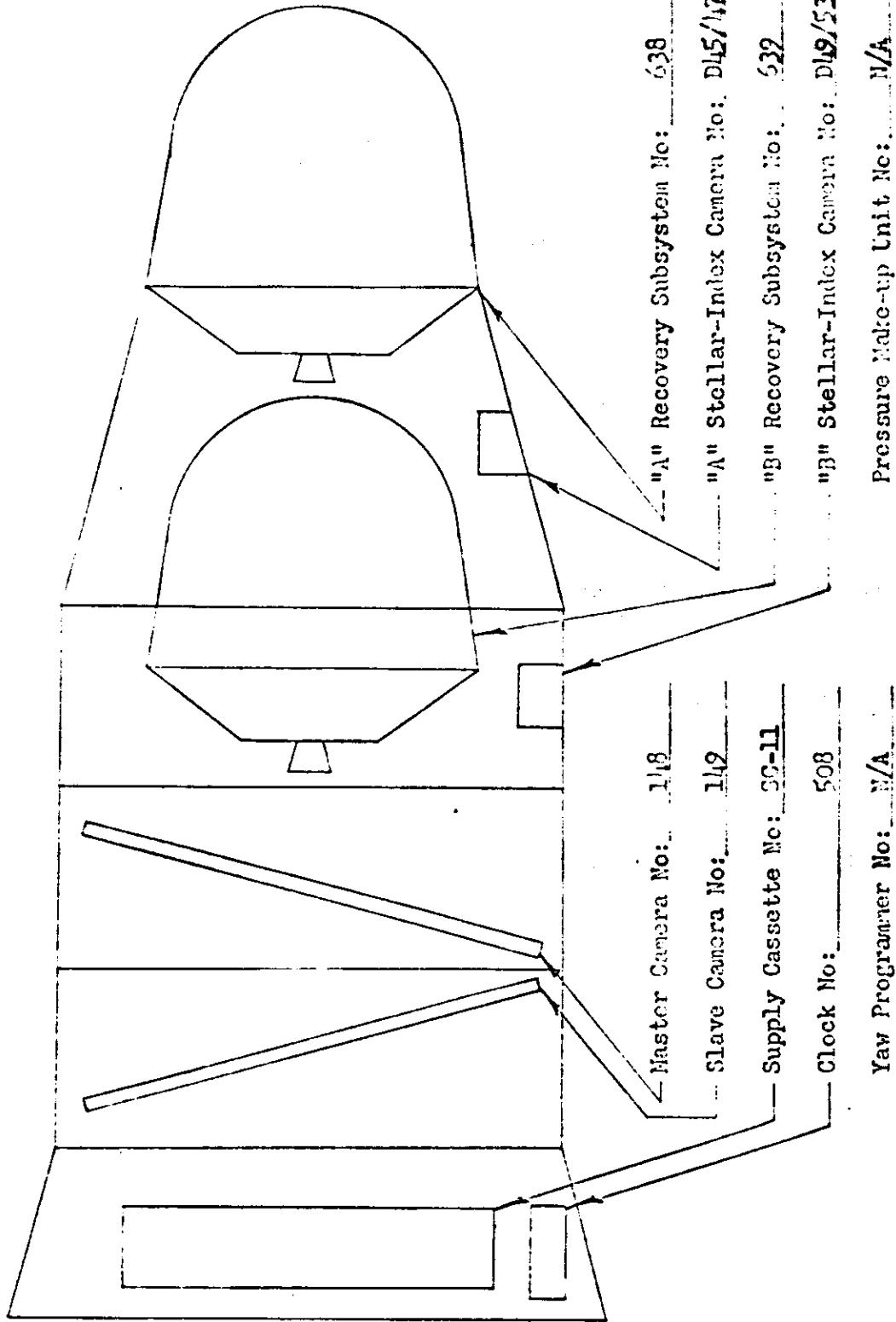
"DANGER" WHEN WEAPONS ARE IN FLAMES.  
INTERMITTENT CUE SCENE.  
COULD IN FLAMES 4 SEC. & LAST TWO FLAMES.

1" DANGER" WHEN WEAPONS ARE IN FLAMES.

c/e switch SCENIC  
INTERRUPTED DUE SCENE.  
STRET-UP COCKAUA.  
SCREATCHES NONE D.

"DANGER" WHEN WEAPONS ARE IN FLAMES.  
INTERMITTENT CUE SCENE.  
COULD IN FLAMES 4 SEC. & LAST TWO FLAMES.

Schematic Diagram Profile - COMM-A System



Failure of the Stellar camera shutter during Mission 1006-1 and failure of the master camera center of format switch caused loss of stellar-index photography during the latter part of Mission 1006-2. This will seriously affect cartographic use of the photography obtained during the mission.

G. RECOMMENDATIONS

Investigation and failure mode analysis resulting from the past flight analysis have resulted in the following actions:

1. Perform complete re-evaluation of the factors that produce uncompensated image motion in the Corona System.
2. Analyze and modify the main door and surrounding structure design to assure normal door injection. LMSC has accomplished this for all future flight units.
3. Modify the center of format switch to reduce future failures. Itek has accomplished this on all future flight units.
4. Study the problems of horizon camera exposure to determine if correct settings are used or whether "fuzzy" horizon images are due to external causes. Joint effort by Itek and LMSC is underway.
5. Review horizon camera boot installation to assure proper tension. LMSC accomplished this review and verified procedures are proper.
6. Improve reliability of stellar camera shutter and eliminate shutter open condition during shutter cock. Itek is phasing in an improved shutter as rapidly as possible.
7. Review Stellar-Index camera supply spool capacity requirements to preclude early depletion.
8. Modify fiducial lamp neutral density filter attachment to assure permanent filter positioning.

## SECTION II

### SPACE STRUCTURE SUBSYSTEM

#### A. COMPONENT ASSIGNMENT

<u>Component</u>	<u>Serial Number</u>
Clock	508
Telemeter "A"	T/M 107
Telemeter "B"	T/M 104

#### B. PREFLIGHT TESTING

The panoramic cameras were installed into their respective barrel sections on 6 March 1964. Command System Test and Pyro tests were completed on 16 March 1964 with no problems. The clock functional testing was accomplished with no recorded anomalies.

The System Light Leak Test was successfully completed on 6 April 1964. Subsequent investigation disclosed a light leak at the Felt Door Assembly. The component was replaced and the light leak removed.

A new Flight Commutator was installed on 26 April 1964 to cure the presence of some bad data points. Point 1152 (Separation Monitor) showed that the wire had pulled from the pins in connectors J100 and J61.

#### C. ENVIRONMENTAL TESTING

##### Summary

The J-09 Payload System completed a six-day orbit simulation test at the Sunnyvale HIVOS Chamber. A seven-day test was planned but was shortened as all primary objectives had been met. The chamber program was set for on-orbit temperature and pressure environment.

Pressure data as recorded by an alphatron gage indicated that the instrument pressures were below the suspected corona range. As the instruments were operated, the pressure increased through the corona region. Corona markings were reported to be acceptable on both panoramic instruments and the "A" Stellar/Index. The "B" Stellar/Index was reported as unsatisfactory.

The Electrical and Mechanical operation of the system was generally acceptable, except for the following: (1) The "B" Stellar/Index failed; (2) Cycle rate predictability exceeded limits; (3) Instrument #1 failed to meter

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for 1/2 frame on one occasion; and (4) The V/H programmer potentiometer developed an electrical open.

The instrument temperatures varied from 73° F to 82° F for Instrument #1 and 66° to 75° F for Instrument #2 after a nominal 10° self-heating correction.

Two self-heating tests were conducted and showed good repeatability.

A pressure make-up system was tested on selected operations throughout the HIVOS test. Results are reported in T9-4-028, "J-9 Pressure Make-up System Test (4-9-64)".

#### Instrument Performance

Cycle Counter versus film footage pot deviations ranged from -445 feet on #1 Instrument to + 275 feet on #2 Instrument during the "A" Operation. These deviations are not a good indication of calibration accuracy, since the "A" spools loose wrapped during the test.

Cycle Counter versus film footage pot deviation ranged from -143 feet on the #2 Cassette to +122 feet on the #1 Cassette, during the "B" Mission.

Separation monitor (11-52) read 5V+ throughout the "A" Mission and 0V throughout the "B" Mission. Normal is 0.5 in "A" and 1.2 in "B".

A 3V 60 CPS Voltage appeared on top of the tape recorder channel #1 during the cut and wrap sequence.

Tape recorder Channel #2 was extremely noisy. Data (Status Channel #11) would be unusable.

The "A" S/I Index metering Idler was noisy.

Both film footage pots had as much as .15V level change during a commutator sample.

Serious ground loop problems exist during Instrument Operates. T.M. monitors referenced to 28V regulated return shift as much as 3.0V when both instrument center formats fall together.

#### Clock Performance

Clock correlation with IRIG "C" showed a maximum of 87 milliseconds slow for the first three days of operation. In the "B" Phase of the HIVOS test, a 3600 second error was detected in the IRIG "C" time generator. Correlation

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between the clock and IRIG "C" in the "B" Phase of operation showed the clock to be 96 milliseconds fast after the 3600-seconds error had been corrected.

#### Temperature Summary

Three self-heating tests were conducted to determine repeatability of self-heating corrections. These were conducted near the beginning, prior to soak, and prior to second recovery during the HIVOS test. The test prior to soak was not a good test as the first part of the test was not recorded. The first and third tests showed good repeatability. The self-heating versus time curve to be used is shown in Figure 1, page 10. Sample instrument temperatures are listed. They are the average temperature and a 10° self-heating correction is applied:

<u>Day</u>	<u>Orbit</u>	<u>Instrument #1</u>	<u>Instrument #2</u>
1	1	70°	70°
2	6	76°	68°
3	11	82°	77°
5	1	70°	62°
5	16	64°	56°
6	11	66°	60°

#### Pressure Summary

An alphatron gage was used to monitor the internal pressure of the system. A pressure make-up system was used for selected orbits during the "A" Mission. The pressure of the system passed through the suspected corona range during instrument operation without the use of the Pressure-Make-up System. Data taken from Selected Orbits are shown below:

<u>Mission</u>	<u>Orbit</u>	<u>Mode</u>	<u>No. of Cycles</u>	<u>Pressure - Microns of Hg</u>	
				<u>Start</u>	<u>End</u>
A	2	Mono 1	40	15	33
A	2	Mono 2	40	23	40
A	42	Stereo	30	2	10
A	42	Mono 1	16	2	4.5
B	Confidence After Soak	Stereo	17	1.6	4.9
B	1	Stereo	23	1.4	5.0
B	Last	Mono 1	38	0.6	2.7
B	Last	Mono 2	29	0.6	3.4

Note: Pressure make-up system was not used during the operations listed above.

J-9 SELF HEATING TEST, MINES

7/5 103, 103, 105, 106, 112, 113, 203, 201, 205, 207, 208, 209, 212, 213  
7/5 102, 108, 109, 110, 210

7/5 111, 211

7/5 205

7/5 THRUST CONE  
7/5 SUPPLY 5000L

0 0 0 0

25

20

15

10

5

TIME FROM TURN ON {MINUTES}

100

## D. VAFB TESTING

One main door ejector was replaced as it would not release within the specified pressure range.

## E. IN-FLIGHT DATA

### Instrumentation and Command Performance

The command system operated properly. No problems were reported. The Instrumentation System functioned normally, except the recovery system separation monitor (11-1-52), and the film footage pot consumption indicators (11-1-25, 31). All temp sensors, status points and continuous channels functioned normally.

The recovery system separation monitor (11-1-52) stepped to 0.9 volts at in-flight reset. The normal level was 0.5 volts. Preflight calibration indicated that 1.0 volts corresponds to "A" SRV demated and "B" SRV mated, and 0.75 volts correspond to "A" SRV mated and "B" SRV demated. Each condition could be caused by either one of two switches being improperly adjusted, allowing a switch to relax during vibration at inflight reset. Changes being considered to prevent recurrence are as follows:

1. Install a new type switch to provide more over-travel capability.
2. Change mounting to produce more over-travel.
3. Epoxy bond switch to mounting bracket after adjustment to prevent slipping.

The film footage pots, both missions, indicated approximately 10% less payload consumption than the cycle counter monitors. The voltage reading was approximately 5% less than was expected. It was determined that the tape recorder input impedance was 250K ohms and the VCO input impedance was 1.0 megohms. This results in a 200K load being connected to the film footage pot wiper during commutator sample time. The pot impedance is 20K maximum and this load reduces the output from a normal 5.0 volts to 4.75 volts. This loading would not appear on channel 8-2-56 and 8-2-47 as the load is only connected while channel 11 commutator is sampling the points. Future system calibration techniques will be revised to allow for VCO and T/R loading.

### Panoramic Camera Performance

Both Panoramic cameras operated throughout both missions. Camera operation was monitored on 10 orbits during the flight. Significant items

of operation and malfunctions are as follows:

1. Slave center-format switch failed open on orbit 1 [REDACTED] through mid-orbit 3.
2. Slave door failed to eject at inflight reset but ejected after orbit 2 [REDACTED]
3. Master center-format switch failed closed intermittently after orbit 103.
4. First and second recovery performance was normal.
5. Maximum cycle rate deviations were 2.6 and 3.5 percent for the master and slave respectively.
6. Payload in both instruments was depleted at D-Timer start on orbit 128 over [REDACTED].

Two anomalies were noted with center-format switches. The first failure was noted on the Aft camera on orbit 1 over [REDACTED] where the center-format switch failed to close from frame 10 to the end of a 16-frame operation. Comparison of cycle counter readings and post-flight payload analysis showed that the switch is considered as the most probable cause of failure.

The second failure occurred on the Fwd camera center-format switch just after an operation on orbit 103 over [REDACTED]. The center-format switch failed closed intermittently. The switch would close at the appropriate time, but would open randomly. A broken switch is considered to be the most probable cause of failure.

The following steps are being taken to prevent recurrence:

1. Modified techniques to check switch adjustment and operation.
2. Engineering investigation for a new type switch.

The Aft camera door failed to eject at inflight-reset. Telemetry data indicated that the squibs were initiated properly. The payload data report indicated that the slave camera door separated on frame 18 of orbit #3. Thermal deformation is considered as most likely cause of failure. The ejector held pressure and ejected the door when thermal equilibrium in the structure was reached.

The following changes were made to prevent recurrence of this problem:

1. Provide additional relief in areas where binding could occur.
2. Change attachment bolts from aluminum to titanium close tolerance bolts.



3. Increase torque from 40 - 50 inch/pounds to 100 + inch/pounds.
4. Install modified lifter bars.
5. Study paint change to modify thermal response.

Variations of the cycle period repeatability were evident throughout the mission with maximum errors of 2.6 and 3.5 percent for the master and slave instruments respectively. A tabulation of the cycle period data and percent error from the predicted cycle period are included in the listing below.

#### CYCLE PERIOD DATA

<u>ORBIT</u>	<u>TIME UP RAMP</u>	<u>MASTER</u>			<u>SLAVE</u>		
		<u>Nominal</u>	<u>Actual</u>	<u>% Error</u>	<u>Nominal</u>	<u>Actual</u>	<u>%Error</u>
1	2010	2.307	2.290	0.75S	2.322	2.322	0%
9	1010	3.791	3.777	0.37F	3.780	3.843	1.63S
25	1075	3.639	3.640	0	3.629	3.680	1.4 S
40	1136	3.503	3.488	0.43F	3.495	3.536	1.18S
56	1200	3.368	3.360	0.24F	3.361	3.413	1.54S
72	1266	3.220	3.230	0.3 F	3.237	3.290	1.6 S
88	1330	3.117	3.123	0.19S	3.160	3.193	1.04S
103	1390	3.003	3.01	0.2 S	3.004	3.070	2.2 S
110	2975	2.302	2.353	2.11S	2.318	2.400	3.54S
119	1457	2.860	2.923	2.20S	2.862	2.948	3.00S

The cut and wrap sequence occurred over [REDACTED] All operations, functions and switchover was normal.

The second recovery was initiated over [REDACTED] Tracking Station and all functions performed normally. At initiation both cameras were observed to be out of film. The 99/101% shuttles were on the take-up side indicating normal film depletion. The film length summary is:

	<u>MASTER</u>	<u>SLAVE</u>
Off Spool Length	210	210
Control Strips (Run through camera and retrieved)	82	64
Recovered in "A" Capsule	7,943	7,862
Recovered in "B" Capsule	7,879	8,005
<b>TOTAL ON SPOOL</b>	<b>16,114 Ft.</b>	<b>16,141 Ft.</b>

Note: Film length should be 16,000 ± 80 Feet

REPORT

## Stellar/Index Camera Performance

### Mission 1006-1

The stellar camera functioned normally for 60% of the mission. The stellar shutter failed open, closed and double-pulsed for 40% of the frames taken during the mission.

The index camera functioned properly throughout the mission with no anomalies reported. Film depleted approximately 20 frames prior to the end of the Mission 1006-1.

### Mission 1006-2

The stellar and the index cameras operated properly with no shutter, metering or recording abnormalities reported. The master center-format switch failure after orbit 103 caused the S/I to expose 279 frames instead of the nominal 428 frames.

## Clock Performance

The clock operation was normal with the time accuracy within the reading accuracy of the analog records used for correlation. A linear clock offset of .209 seconds was indicated between orbits 9 and 126.

## Recovery System Performance

### Mission 1006-1

A successful air catch of the capsule was made on orbit 65. The impact point was within normal tolerances. All capsule re-entry events, except the thrust-cone physical separation switches, occurred within tolerance. This monitor indicated a 0.75 second delay between the separate signal and the switch actuation. Similar delays have been noted on previous recovery systems with delays ranging up to 1.0 seconds.

The condition of the recovered capsule was satisfactory with damage limited to normal paint blistering. Post-flight inspection and test showed no anomalies.

Below is a tabulation of the sequence to re-entry and recovery event times.

RECOVERY SEQUENCE OF EVENTS (Mission 1006-1)

<u>Event</u>	<u>System Time</u>	<u>Actual</u>	<u>Delta Time</u> <u>Nominal</u>
Transfer	3703.84		
Elect. Disconnect	3704.91	1.07	0.90 <sup>+.430</sup> -.400
Separation	3705.83	1.99 *	2.0 <u>+.25</u>
Spin	3708.35	3.44 **	3.4 <u>+.30</u>
Retro	3715.98	7.63	7.55 <u>+.45</u>
Despin	3726.58	10.60	10.75 <u>+.54</u>
T/C Separation	3728.84	2.26	1.5 <u>+.15</u>
"G" Switch Open	4252.32	-	-
Parachute Cover Off	4285.71	33.39	34.0 <u>+.1.5</u>
Drogue Parachute Deployed	4286.38	.67	0.75 <u>+.08</u>
Drogue Parachute Released	4296.72	10.34	10.05 <u>+.1.5</u>
Main Parachute Deployed	4297.26	.54	1.2 <u>+.15</u>
Main Parachute Disreefed	4301.35	4.09	4.0 <u>+.1.7</u>

\* From Transfer

\*\* From Electrical Disconnect

Spin Rate - 64.1 RPM

Despin Rate - 11 RPM

Retro Velocity 911 Ft/Sec

Recovery Battery Voltage At Arm 13.7 V

Mission 1006-2

The second recovery unit was successfully recovered by air catch on orbit 128. The impact point was within normal tolerances.

Post-flight inspections and tests showed all events to be normal. Damage to the recovery capsule was limited to normal blistering of paint.

Below is a tabulation of the sequence of monitored re-entry and recovery event times.

### RECOVERY SEQUENCE OF EVENTS (Mission 1006-2)

Event	System Time	Delta Time	
		Actual	Nominal
Transfer	85696.72		
Elect. Disconnect	85697.76	1.04	.9 $\pm$ .43 -.40
Separation *	85698.82	2.10	2.0 $\pm$ .25
Spin **	85701.19	3.43	3.4 $\pm$ .30
Retro	85708.79	7.60	7.55 $\pm$ .4
Despin	85719.49	10.70	10.75 $\pm$ .81
T/C Separation	851721.04	1.55	1.5 $\pm$ .15
Volt Monitor Closed	85806.96	85.92	100 $\pm$ 40
Volt Monitor Open	-	-	-
"G" Switch Open	86299.20		

Spin Rate - 65 RPM

Despin Rate - 12 RPM

Retro Velocity - 884 Ft/Sec.

\* From Transfer

\*\* From Electrical Disconnect

#### De-Activate/Re-Activate Performance

A vehicle de-activate sequence was initiated on orbit 133. All functions performed normally. The vehicle was re-activated again on orbit 149. The proper attitude and stability was regained and maintained.

A Lifeboat Real-Time Recovery Sequence was initiated on orbit 174. All vehicle functions performed as required.

The vehicle power was expended between orbit 191 and orbit 199 where all electrical functions were lost.

The vehicle orbital life ended on 6/17/64 or 6/19/64.

## Temperature Summary

Temperature environment of the system is presented in Tables 3 and 4 on pages 18 and 19. Data were obtained from real-time orbits acquired at [REDACTED] Tracking Station. Self-heating corrections were made based on self-heating data taken during the HIVOS Environmental Chamber Tests. Self-heating correction curves were shown in Figure 1 on page 10.

J-9 1176 RECOVERY

<u>Stator</u>	<u>Fwd L2 Turret 1</u>	<u>L</u>	<u>2</u>	<u>16</u>	<u>25</u>	<u>31</u>	<u>40</u>	<u>47</u>	<u>56</u>	<u>63</u>	<u>72</u>	<u>79</u>	<u>88</u>	<u>95</u>	<u>103</u>	<u>110</u>
1	240+	42	93	45	106	42	106	45	103	11	-	11	37	11	34	7
2	240+	18	21	25	25	15	54	18	24	1	11	4	14	4	24	102
3	240+	6	35	10	35	10	35	10	32	21	107	24	115	24	116	116
4	240+	56	81	56	78	56	75	56	75	44	130	41	133	41	78	78
5	240+	83	113	83	111	82	105	76	99	49	87	40	87	40	78	78
6	240+	73	154	70	154	67	118	67	138							
<u>Barrel No. 2</u>																
1	177	79	107	76	101	70	99	57	93	47	82	44	79	41	76	76
2	164	56	142	59	145	56	140	43	134	43	126	40	124	37	115	115
3	208	14	87	20	95	20	95	23	95	17	87	17	95	17	87	87
4	210	6	13	7	13	7	13	10	13	0	7	3	3	7	7	7
5	185	21	47	21	44	25	44	25	44	12	31	9	34	15	31	31
<u>Conic Adapter</u>																
1	178	59	92	59	92	59	86	66	83	43	71	36	71	36	59	59
<u>Clock</u>																
1	99	71	75	73	73	71	73	73	73	73	62	62	60	65	60	58
2	103	69	73	71	73	73	72	71	71	58	60	60	60	60	60	56
<u>Thrust Cone</u>																
1	130+	46	46	44	44	42	44	44	44	44	48	46	47	46	48	43
2	67	69	63	64	60	64	59	62	55	60	56	56	55	55	55	54
<u>Stellar Index</u>																
1	88	67	70	64	67	64	64	64	64	64	61	61	65	51	51	51
2	77	67	64	67	64	64	64	64	64	64	61	61	62	42	45	39
<u>Recovery Patt. 11</u>																
1	67	68	68	65	66	65	64	64	64	64	78	75	74	84	74	82
2	94	58	54	54	53	53	53	53	53	53	53	53	53	52	52	52

TABLE 3

כט

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<u>Slaver</u>	<u>Master</u>	<u>Owner</u>
3	4	112
4	5	47
5	6	53
6	7	53
7	8	56
8	9	56
9	10	55
10	11	55
11	12	55
12	13	55
13	Slave	55
1	2	110
2	3	103
3	4	25
4	5	88
5	6	72
6	7	72
7	8	56
8	9	56
9	10	56
10	11	56
11	12	56
12	13	56
13	Supply	56
1	2	57
2	3	57
3	4	57
4	5	57
5	6	57
6	7	57
7	8	57
8	9	57
9	10	57
10	11	57
11	12	57
12	13	57
13	Supply	57
1	2	58
2	3	58
3	4	58
4	5	58
5	6	58
6	7	58
7	8	58
8	9	58
9	10	58
10	11	58
11	12	58
12	13	58
13	Supply	58
1	2	59
2	3	59
3	4	59
4	5	59
5	6	59
6	7	59
7	8	59
8	9	59
9	10	59
10	11	59
11	12	59
12	13	59
13	Supply	59
1	2	60
2	3	60
3	4	60
4	5	60
5	6	60
6	7	60
7	8	60
8	9	60
9	10	60
10	11	60
11	12	60
12	13	60
13	Supply	60
1	2	61
2	3	61
3	4	61
4	5	61
5	6	61
6	7	61
7	8	61
8	9	61
9	10	61
10	11	61
11	12	61
12	13	61
13	Supply	61
1	2	62
2	3	62
3	4	62
4	5	62
5	6	62
6	7	62
7	8	62
8	9	62
9	10	62
10	11	62
11	12	62
12	13	62
13	Supply	62
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3	4	63
4	5	63
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13	Supply	63
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8	9	64
9	10	64
10	11	64
11	12	64
12	13	64
13	Supply	64
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3	4	65
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8	9	65
9	10	65
10	11	65
11	12	65
12	13	65
13	Supply	65
1	2	66
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4	5	66
5	6	66
6	7	66
7	8	66
8	9	66
9	10	66
10	11	66
11	12	66
12	13	66
13	Supply	66
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2	3	67
3	4	67
4	5	67
5	6	67
6	7	67
7	8	67
8	9	67
9	10	67
10	11	67
11	12	67
12	13	67
13	Supply	67
1	2	68
2	3	68
3	4	68
4	5	68
5	6	68
6	7	68
7	8	68
8	9	68
9	10	68
10	11	68
11	12	68
12	13	68
13	Supply	68
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2	3	69
3	4	69
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9	10	69
10	11	69
11	12	69
12	13	69
13	Supply	69
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3	4	70
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7	8	70
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9	10	70
10	11	70
11	12	70
12	13	70
13	Supply	70
1	2	71
2	3	71
3	4	71
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11	12	71
12	13	71
13	Supply	71
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11	12	72
12	13	72
13	Supply	72
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9	10	73
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11	12	73
12	13	73
13	Supply	73
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9	10	74
10	11	74
11	12	74
12	13	74
13	Supply	74
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12	13	75
13	Supply	75
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13	Supply	76
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9	10	77
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13	Supply	77
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8	9	78
9	10	78
10	11	78
11	12	78
12	13	78
13	Supply	78
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5	6	79
6	7	79
7	8	79
8	9	79
9	10	79
10	11	79
11	12	79
12	13	79
13	Supply	79
1	2	80
2	3	80
3	4	80
4	5	80
5	6	80
6	7	80
7	8	80
8	9	80
9	10	80
10	11	80
11	12	80
12	13	80
13	Supply	80
1	2	81
2	3	81
3	4	81
4	5	81
5	6	81
6	7	81
7	8	81
8	9	81
9	10	81
10	11	81
11	12	81
12	13	81
13	Supply	81

**SECTION III**

**MASTER (FWD) PANORAMIC CAMERA**

**A. COMPONENT ASSIGNMENT:**

<u>Component</u>	<u>Serial Number</u>
Main Camera	148
Main Camera Lens	1242435
Supply Horizon Camera	161B
Supply Horizon Camera Lens	813549
Take-up Horizon Camera	157A
Take-up Horizon Camera Lens	812206
Supply Cassette	SC-11

**B. CAMERA DATA AND FLIGHT SETTINGS:**

**Main Camera:**

Lens	24" f/3.5
Slit Width	0.200"
Filter Type	Wratten 21
Film Type	Eastman Type 4404

**Supply (Port) Horizon Camera:**

Lens	55mm f/6.8
Aperture Setting	f/6.8
Exposure Time	1/100 seconds
Filter Type	Wratten 25

**Take-up (Starboard) Horizon Camera:**

Lens	55mm f/6.8
Aperture Setting	f/8.0
Exposure Time	1/100 second
Filter Type	Wratten 25

### C. PRE-FLIGHT DATA

Camera number 148 was received at A/P on 27 January 1964 for contractor Receiving Inspection and Acceptance Testing. During these phases the following problems were encountered and action taken:

1. The horizon camera fiducial lamps were inoperative. The burned out lamps were replaced.
2. The intermediate roller assembly would not turn freely. The assembly was replaced as a defective bearing was located.
3. The shuttle assembly did not move freely. This anomaly was waived by Itek.

The camera underwent the normal System Functional Test during which the sticky shuttle assembly caused Fail-Safe #1 to trip. Disassembly of the shuttle determined that some epoxy was in the bearings; the items were cleaned and reinstalled. The cycle rate check recorded rates that were approximately 3% slower than the original settings. The tachometer motor voltages were adjusted to produce the correct rates. The dynamic photographic resolution test results are shown in Figure 2, page 22.

The V/H potentiometer was replaced during Environmental Testing as noise developed at the top of the ramp. The corona discharge marking was confined to the third frame after start and was considered within the acceptable level.

A new design V/H potentiometer was installed during VAFB testing and the V/H transducer with gold plated stepper switch contacts was incorporated.

### D. IN-FLIGHT DATA

The detailed summary of the telemetry data obtained during the mission is in Section II. The failure of the center-of-format switch at the start of pass D103 during Mission 1006-2 was observed in-flight.

### E. POST-FLIGHT PERFORMANCE EVALUATION

The photographic quality of the FWD camera film was excellent. This is primarily attributed to the processing of the original negative which was much closer to the exposure-processing criteria than achieved on previous missions. The informational content of the photography was exceptionally high because of the low perigee which improved the scale factor by 10% to 15%. The abnormal V/H error that resulted from the orbital anomalies did not significantly affect

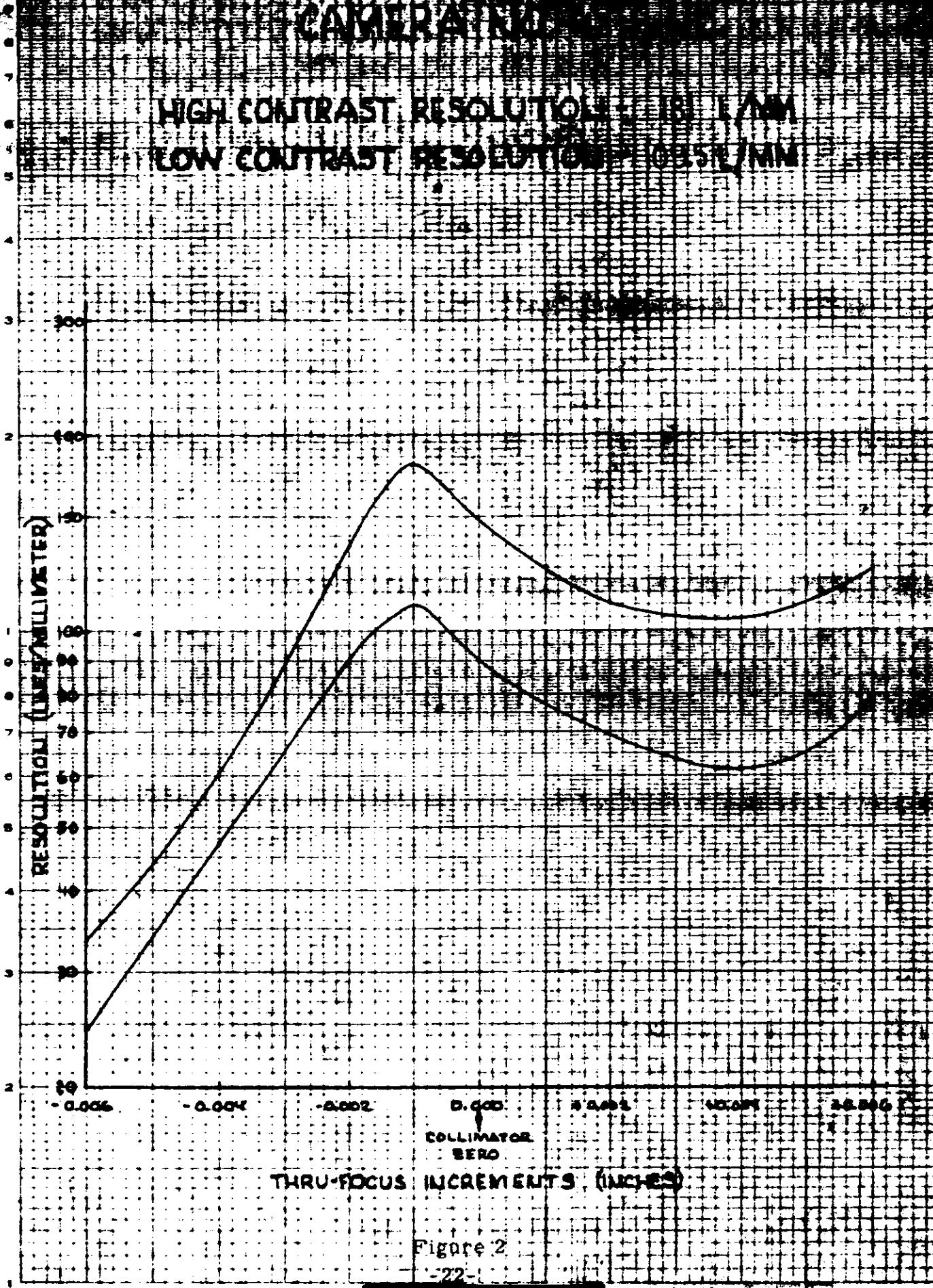


Figure 2

the resolution of the imagery as the error was below 15.4% during 90% of Mission 1006-1 operations and 11.6% for 90% of Mission 1006-2 operations. The analysis of image smear is contained in Section XV.

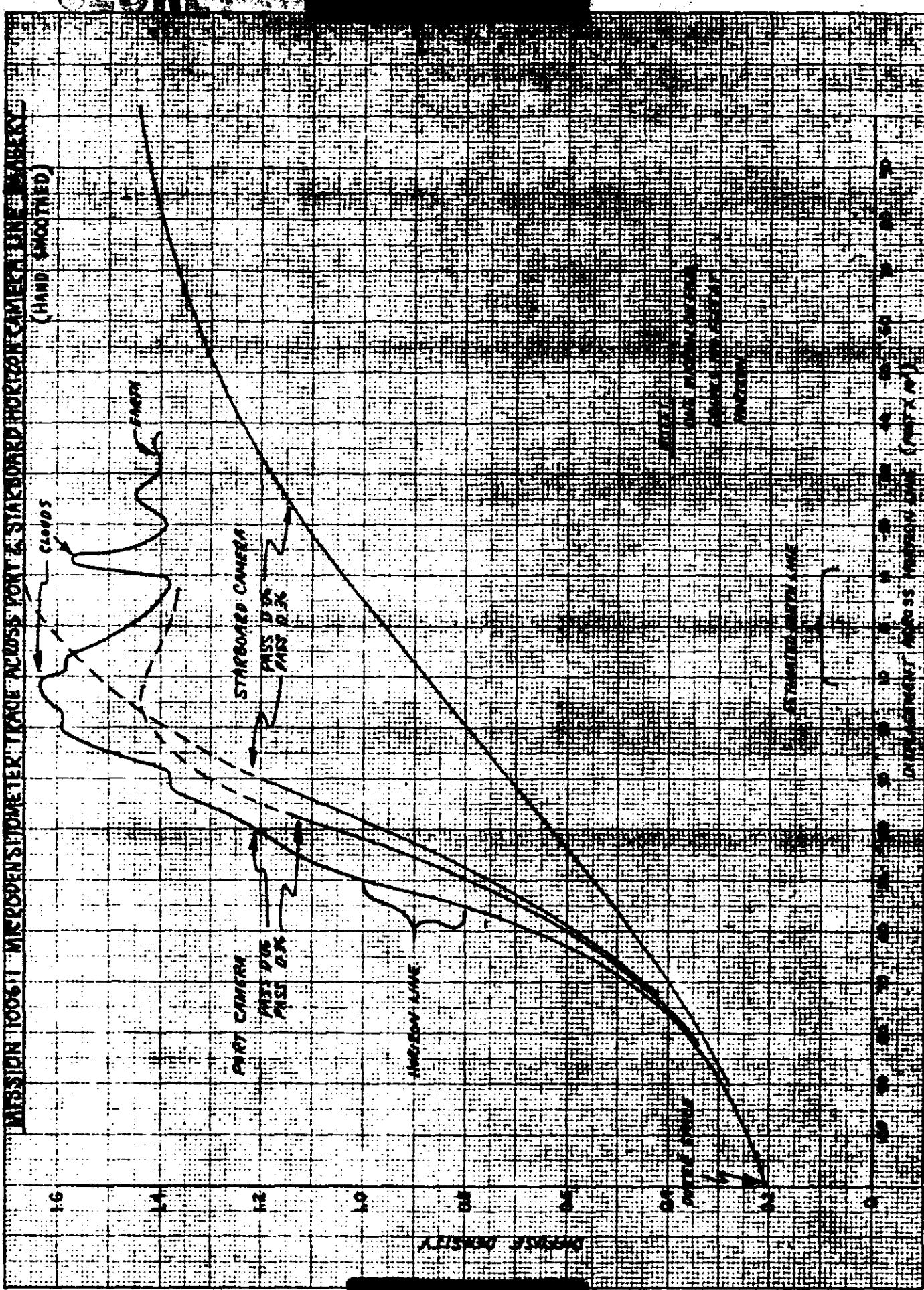
The camera center-of-format switch failed intermittently from the start of pass D103 to the end of the mission. This resulted in the loss of the binary data block time word, normal end of pass marks and horizon fiducials. The horizon cameras continued to function however the images were heavily smeared. Testing subsequently conducted by Itek simulated this failure mode when one of the dual center-of-format switches, #S-104, operated normally while the second switch, #S-105, was continually closed.

The dual switches were incorporated to improve reliability; however they actually reduced the reliability. As a result switch #S-105 has been removed from the circuit as recommended by Itek.

During the period of proper center-of-format switch operation the data block, end-of-pass mark and horizon cameras operated properly. The 200 cps time track functioned properly throughout both missions.

The starboard horizon camera photography was very soft through pass D35. The quality was somewhat improved thereafter however it was significantly inferior to the quality of the port camera photography. This was the first case of poor imagery since the incorporation of the 55mm horizon camera lenses. Several theories exist as to the cause of the poor quality however the current studies have been inconclusive to date.

The micro-densitometer traces made of the typical port and starboard horizon line photography are shown in Figure 3, page 24. The poor starboard photography is shown by the low slope of the pass D06 trace and the improvement after D35 corresponds to the increase in slope. The port image trace shows the normal slope that results from usual horizon camera photography.



## SECTION IV

### SLAVE (AFT) PANORAMIC CAMERA

#### A. COMPONENT ASSIGNMENT

<u>Component</u>	<u>Serial Number</u>
Main Camera	149
Main Camera Lens	1262435
Supply Horizon Camera	156B
Supply Horizon Camera Lens	813536
Take-up Horizon Camera	156A
Take-up Horizon Camera Lens	813559
Supply Cassette	SC-11

#### B. CAMERA DATA AND FLIGHT SETTINGS:

##### Main Camera:

Lens	24" f/3.5
Slit Width	0.200"
Filter Type	Wratten 21
Film Type	Eastman Type 4404

##### Supply (Starboard) Horizon Camera:

Lens	55mm f/6.8
Aperture Setting	f/8.0
Exposure time	1/100 second
Filter type	Wratten 25

##### Take-up (Port) Horizon Camera:

Lens	55mm f/6.8
Aperture Setting	f/6.8
Exposure Time	1/100 second
Filter type	Wratten 25

### C. PRE-FLIGHT DATA

Camera number 149 was received at A/P on 27 January 1964. No anomalies were reported during Receiving Inspection, Acceptance Testing and Systems Functional Testing. The dynamic, photographic resolution test results are shown in Figure 4, page 27.

The film passed through the camera during Environmental Testing showed a fog pattern in the third frame after start-up caused by corona discharge. This level of marking is within the flight acceptance criteria.

No anomalies were encountered during the Pre-Ship Tests at A/P and the testing at VAFB.

### D. IN-FLIGHT DATA

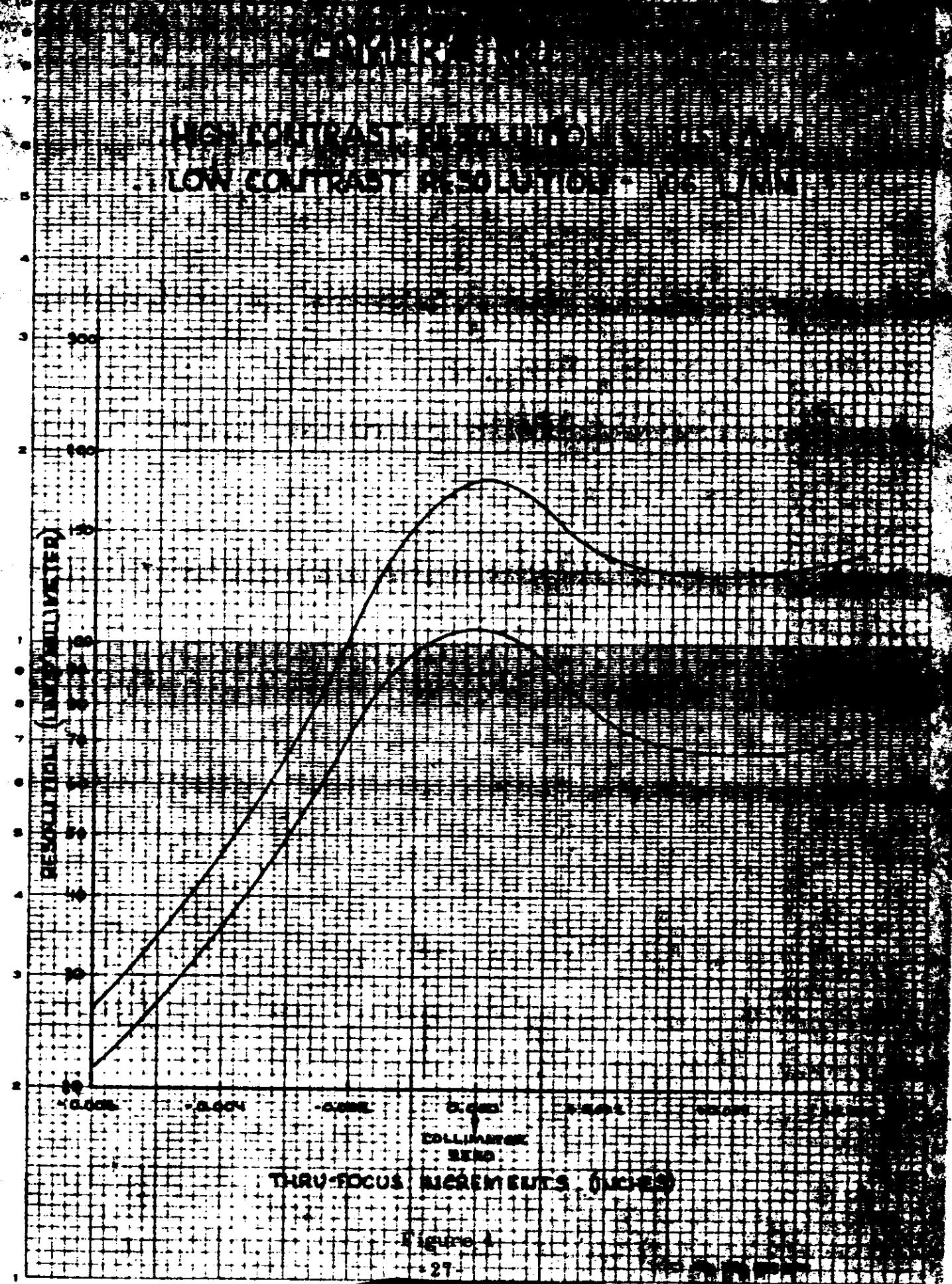
The detailed summary of the telemetry data obtained during flight is in Section II. The failure of the main camera door to eject on command, the eventual door absence and the center-of-format switch failure were observed in flight.

### E. POST FLIGHT PERFORMANCE EVALUATION

The photographic quality and informational content of the AFT camera was as excellent as the FWD camera. The excessive processing given to the Mission 1006-1 original negative was not readily apparent. This problem is discussed in more detail in Section VI.

The main camera door did not eject at the pyro firing command during ascent. The door did not eject until after camera operations during pass D02. It is postulated that manufacturing tolerances and ascent thermal aggravation created a bind between the door and structure. The piston chamber in the door ejection pin puller retained its pressure and caused door ejection after thermal stabilization during passes D02 and D03. A redesign of this area has been completed and incorporated on all future flights.

The center of format switch failed from frame D01-10 through frame D03-17. The binary data block, horizon camera imagery and fiducials were absent during this period however they operated properly throughout the remainder of both missions. It has been concluded that this malfunction was caused by a jammed switch that released after repeated cam action.



The horizon camera imagery was essentially the same as noted in the FWD camera. The port camera produced sharp photography throughout the missions while the starboard imagery was soft. The microdensitometer traces plotted in Figure 3, page 24, are also typical of the AFT camera horizon photography.

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## SECTION V

### PANORAMIC CAMERA EXPOSURE

The exceptionally low perigee resulted in the panoramic cameras operating at the maximum cycle rate for the majority of both missions. Figures 5 and 6 on pages 30 and 31 show the nominal exposure time versus latitude for the three levels of processing. If the orbital parameters had been as planned the actual exposure time would follow the "orbit" curve however the high speed cycle rate limit of the cameras would not permit exposure times above the dashed line marked "ramp".

The illumination conditions during the missions were relatively consistent as the flight took place near the summer solstice. The range of solar elevations and solar azimuth values are shown below for operations in the northern and southern hemisphere. The values of solar azimuth consider the heading of the payload at  $0^{\circ}$  with plus values clockwise and minus values counterclockwise. All values below are positive.

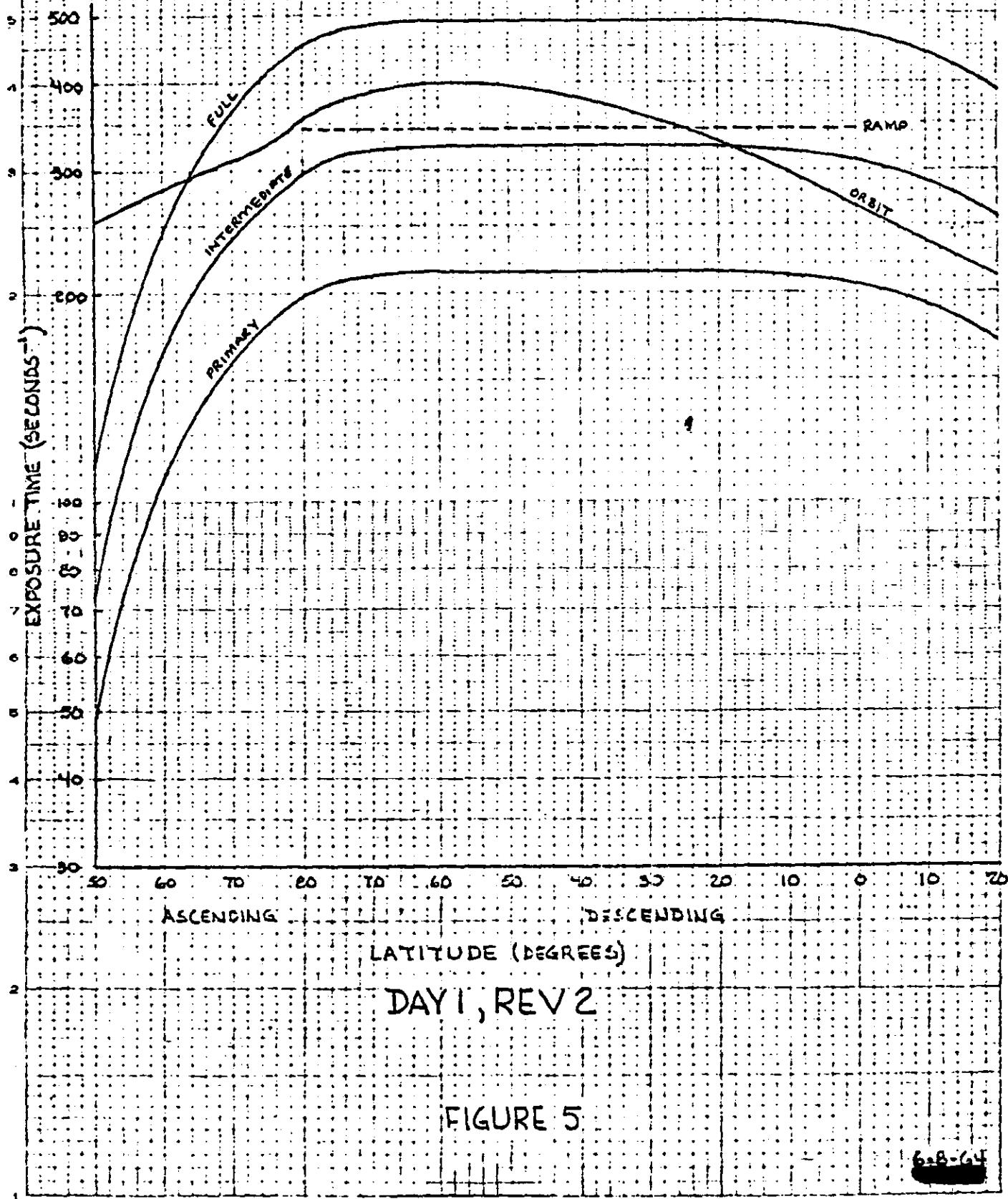
<u>MISSION</u>	<u>HEMISPHERE</u>	<u>RANGE OF SOLAR ELEVATION</u>	<u>RANGE OF SOLAR AZIMUTH</u>
1006-1	North	$38.2^{\circ}$ to $55.7^{\circ}$	$51.2^{\circ}$ to $111.9^{\circ}$
	South	$27.4^{\circ}$ to $30.6^{\circ}$	$138.3^{\circ}$ to $139.9^{\circ}$
1006-2	North	$42.6^{\circ}$ to $63.9^{\circ}$	$35.9^{\circ}$ to $112.3^{\circ}$
	South	$32.2^{\circ}$ to $38.0^{\circ}$	$142.4^{\circ}$ to $147.2^{\circ}$

The actual exposure time that occurred during passes D31 and D110 was calculated from the data block time word and the 200 cycles timing mark count. This analysis was limited to ZI engineering material due to availability.

The exposure time during pass D31 was programmed at a constant rate as controlled by the cycle rate limiter in the V/H programmer. The actual exposure times are shown in Figure 7 on page 32. The first six frames are not shown as they are significantly affected by instrument start-up; some effects of start-up are seen through frame 12. The slight variation in exposure time obtained from the data block is not considered significant as it is in the order of 1%. The variation in the timing mark exposure time is somewhat larger in magnitude. The shape of the anomaly is similar to cases previously observed when the V/H potentiometer center-tap point was crossed. The cause of this particular exposure time variation is under study.

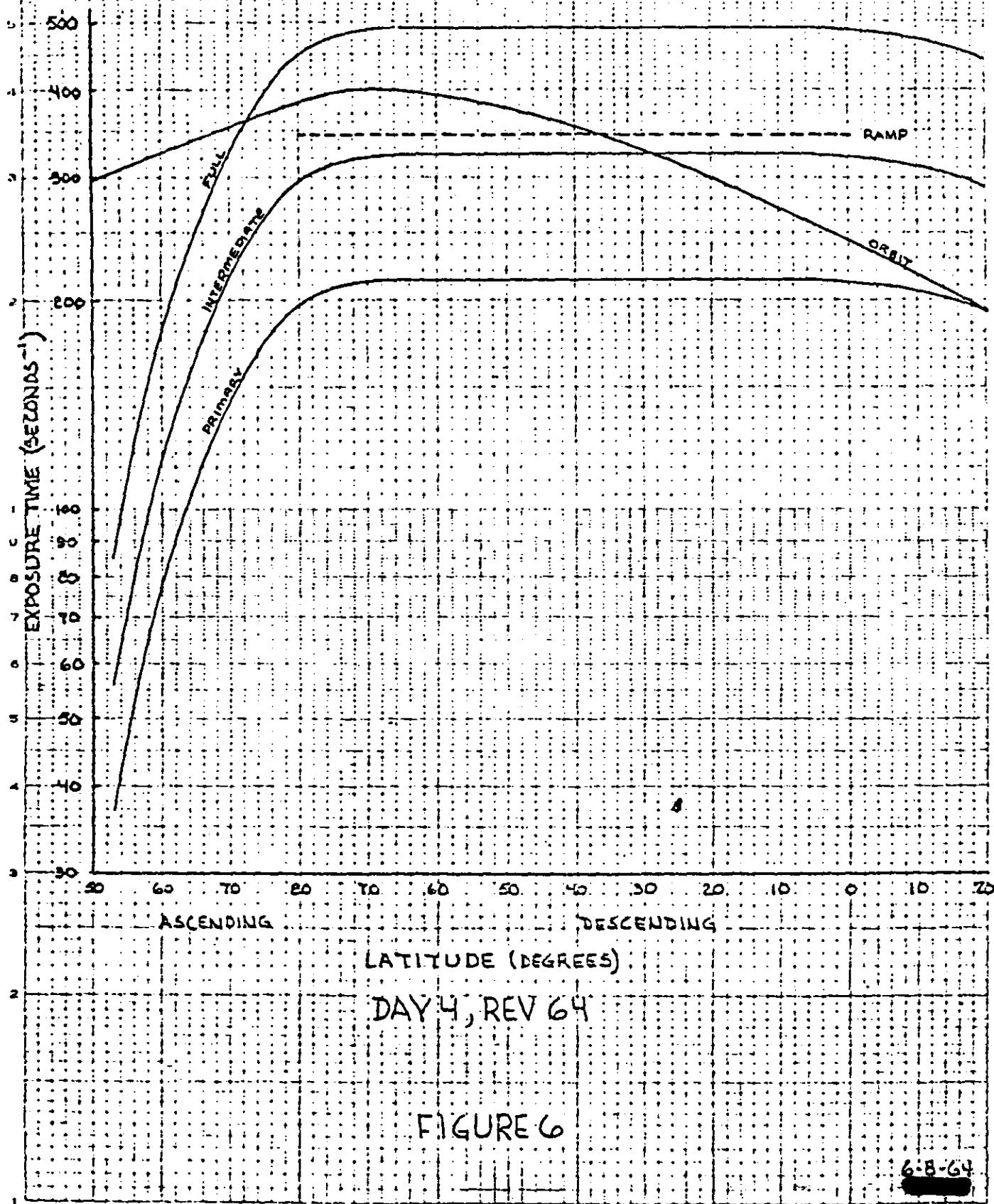
~~SECRET~~

1006-1 EXPOSURE POINTS  
0.200" SLIT



1006-12 EXPOSURE POINTS

0.200" SLIT



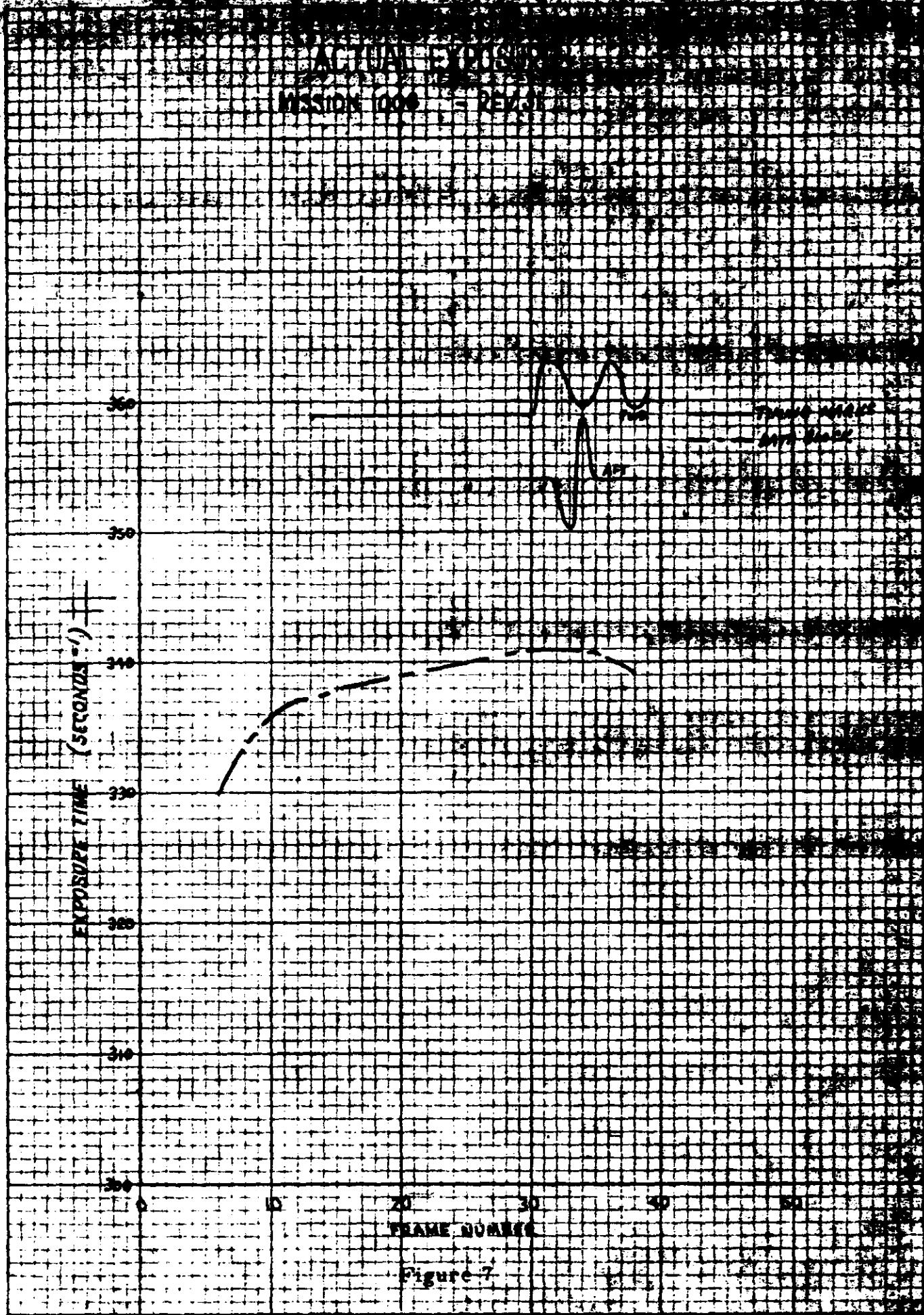


Figure 7

The exposure times that occurred during pass D110 are plotted in Figure 8, page 34. The general shape of the curves is normal as the V/H programmer was on the decreasing portion of the potentiometer.

The most significant factor shown in the two figures is the shorter exposure time, approximately 5%, calculated from the time marks. This calculation of exposure time is considered more accurate than the time derived from the data block as it is an absolute measure of the motion of the scan head, and hence slit, during photography. The exposure time calculated from the data block assumes uniform camera velocities throughout a cycle which is apparently incorrect.

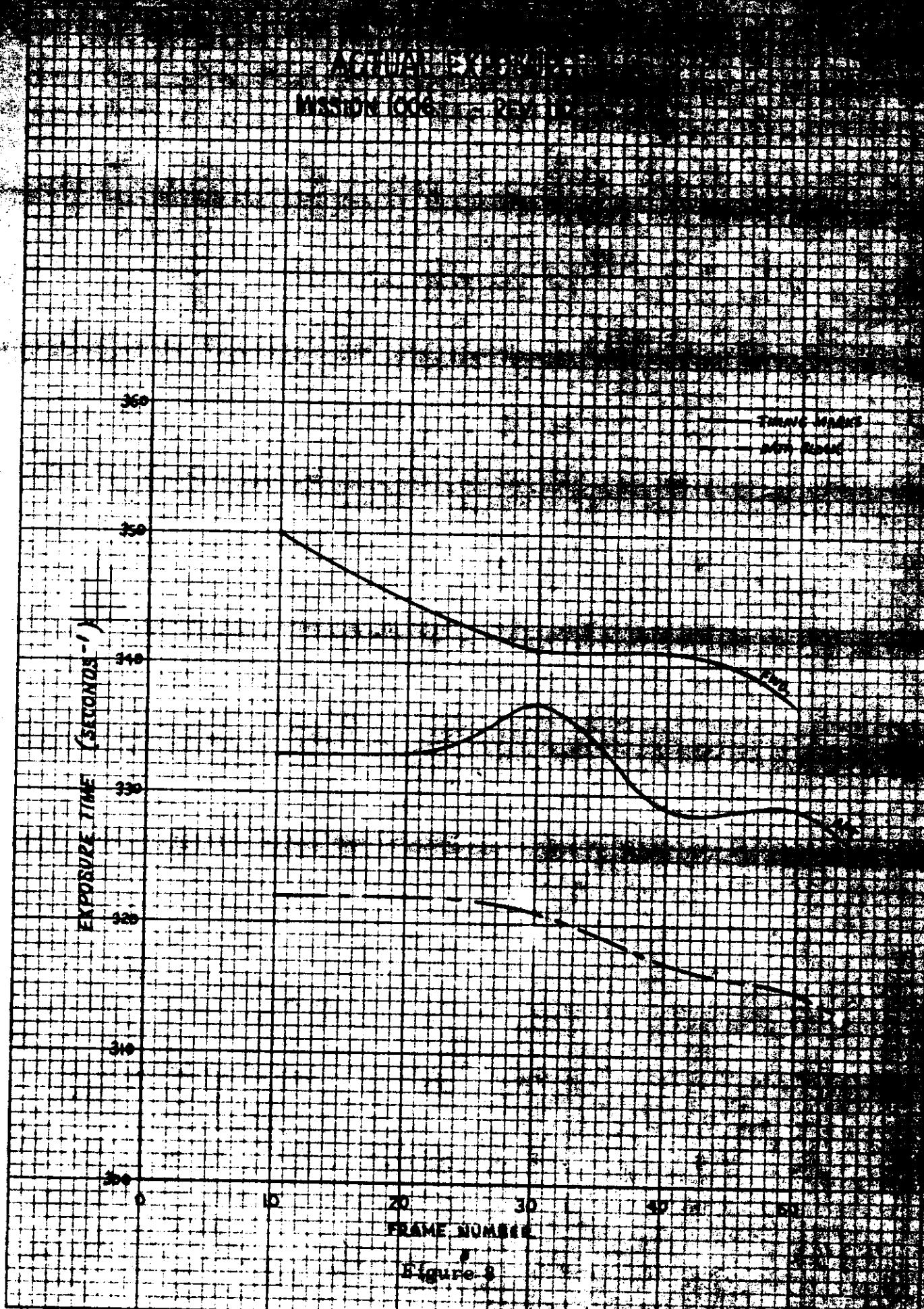
The non-uniformity of scan rate is discussed in more detail in Section VIII. The data shown in Figures 7 and 8 do indicate that at least a portion of the non-uniformity of scan rate is produced by the scan head only and not associated with the lens rotation.

The level of processing that is given to the film is predicted for each mission based on the known factors of camera cycle rate, hence exposure time, and solar elevation. The effects of the unknown factors of weather, such as cloud and snow cover, and mission anomalies are normalized by variations in the processing level.

The predicted and actual processing levels, in percent, for the missions are:

<u>MISSION</u>	<u>CAMERA</u>		<u>PRIMARY</u>	<u>INTERMEDIATE</u>	<u>FULL</u>
1006-1	FWD	predicted	1	99	0
		actual	1	51	48
1006-1	AFT	predicted	1	99	0
		actual	0	23	77
1006-2	FWD	predicted	2	98	0
		actual	30	41	29
1006-2	AFT	predicted	2	98	0
		actual	35	40	25

The significant variation in processing levels used on the four film spools is not normal. Generally, the processing levels used on Corona missions approximate the percentages used on the AFT camera film from Mission 1006-1. The reasons for the decrease in full processing on the remaining three film spools is not understood. However, it is considered to be one of the major contributors to the exceptionally good photography acquired during the missions.



**SECTION VI**  
**DIFFUSE DENSITY MEASUREMENTS**

Tables 4 and 5, pages 37 to 61, list mission data supplied by AFSPPE. This data includes the visual Reciprocal Edge Spread (RES) values, the area on the format in which the value was obtained and the general characteristics of the edge as shown on the data key page. The densitometric measurements of the base plus fog, minimum and maximum terrain densities and the maximum cloud densities are also listed with other general data such as solar elevation, latitude and overlap.

The columns are arranged in the following order:

<u>COLUMN NUMBER</u>	<u>HEADING</u>	<u>DATA</u>
1	-	Ascending or Descending pass
2-4	Pas Nbr	Pass Number
5	-	FWD or AFT camera
6-8	Frm Nbr	Frame Number
9-17	Area 1 RES	RES data in area 1
9-11	WWW	With flight RES value
12-14	AAA	Across flight RES value
15	S	Subject - see key
16	T	Terrain - see key
17	Q	Qualifiers - See Key
18-26	Area 2 RES	RES data in area 2
27-35	Area 3 RES	RES data in area 3
36-44	Area 4 RES	RES data in area 4
45-53	Area 5 RES	RES data in area 5
54-56	D min	Terrain minimum density
57-59	D max	Terrain maximum density
60-62	D B+F	Base plus fog density
63-65	LIM max	Cloud maximum density

CRAFT

<u>COLUMN NUMBER</u>	<u>HEADING</u>	<u>DATA</u>
66-68	LAT	Latitude
68	T	0 = North, 1 = South
69-71	Sun Ele	Solar Elevation
73-74	CLD	Percent cloud cover
75-76	OL	Percent overlap

The data key for the listings of the "Subject", "Terrain" and "Qualifiers" is shown below.

### I SUBJECT

1. Buildings
2. Roads, runways
3. Tanks, A/C, other man-made
4. Non-cultural

### II TERRAIN

1. Flat
2. Hilly
3. Mountains
4. Flat and snow
5. Hilly and snow
6. Mountains and snow

### III EDGE QUALIFIERS

1. Clear
2. Snow
3. Hazy
4. Shadow
5. Snow and Haze
6. Snow and Shadow
7. Haze and Shadow
8. Snow, Haze and Shadow

~~SECRET~~

MISSION 1006-1

PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D D LIM SUN

NBR NBRWWWAAASTQWWWAAASTQWWWAAASTQWWWAAASTQMINMAXB+FMAXLATECLECL

C001F011		012226500+4910099
D001F005		012230500+4910099
C002F006	067070411	121223011 640+4300000
C002F021	078078411	026134011 620+4500099
L002F030	075070411	023121011 610+4600099
D003F006	078085411	092231011 530+4600099
C003F016	072075431	035226011 570+4700099
C003F029		0700754320992140112C7540+4709099
DC03F037		011198530+4810099
C005F005	085078121	049147011219490+4905000
DC05F023067067112		098150017228470+4905599
C005F036	085085111	079175014227450+4905000
C005F046	072072111	044181010217430+4907000
C005F058	063065112	086135010214410+5009799
C005F066	063063111	046153010219400+5005099
C005F076		075078211047100011215380+5000599
C005F088		075075411069144011211380+5005099
C005F098		078072411069119011216350+5007099
C006F005		092112011230500+4709000
CC06F015		011233050+4710000
DC06F025		011240540+4810000
L006F032	078085433	05518_011230530+4307500
CC06F042	075072433	096168011189410+5002000
DC06F052		011221400+5000000
DC06F062		011211380+5000000
C006F072		011031370+4910000
C006F082		011231350+4909000
DC06F092	072078433	044105011215540+4909000
C006F099	063063432	046092011131300+4607500
C006F109		011100200+4910000
C006F119		011201200+4810000
C006F129		011238200+4710000
C006F132		017211200+4710000
CC07F005	078072212	05919_011227500+4703000
C007F015	082085112	057199019 540+4700000
DC007F025	094094111	071177020000520+4700000
C007F035	072065311	057194020000510+4700000
CC007F045	085089121	053158019198490+4700500
CC007F055		020217400+4710099
DC007F065	078059412	087196019228460+4703099
C007F075	085075111	096176018228450+4705000
C007F085078072121		078160017226440+4704000
D007F095	082075432	129231018225420+4707000
C007F105		019224410+4810099
CC07F115		019223390+4810099
CCC7F125		019231380+4810099
D007F135067082433		124226016230560+4809099
D007F145065072433		067112011218350+4608099
C007F155	082075433	053220011221330+4803099
DC007F165	094078433	048213010214320+4801000

~~SECRET~~

PAS FRMARE1 RESARE2 RESARE3 RESARE4 RESARE5 RES D D D LIM SUN  
 NBR NBRWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQMINMAXB+FRAXLATECLDC

DC07F175	082075422	052147011229300+480709
CC07F185	C82C82311	055163010224290+480050
A009F006		011000400- 49999
DC09F008	C75C78111	059119011220530+480909
DC09F018	C99090111	029114011220520+480200
CC09F029	C94090111	063149010230600+490050
LC09F039	C85085111	09114101e232400+490600
DC09F054	C67072112	C82152010236470+500750
DC09F060		065065411C9214901e232460+5001CS
CC15FC05	063055432	091151018234430+500850
DC015F015	065055131	094134019231420+500900
DC15F025	C85090111	060156014226400+500100
CC15F035	C85094111	052149012222390+500300
DC15F045	C78082111	039160012225370+500300
DC15F047	C82082111	042130012223370+500600
DC018F004	070072421	082212018000660+420000
CC018F014	C72C67411	042221018000660+430000
CC018F024	C82182421	067179018000650+440000
CC018F034	C85075431	072209018000650+450000
CC018F044	C72C75422	05218901219000745 1
CC19FC05	C62C61412	181216015000730+380000
DC019F015	C69072413	120216011000720+390000
DC019F025	C55062413	0491780099000710+410000
DC019F035	C57052412	072192019200700+420050
DC019F045	C63063412	049156019000620+430000
DC019F055067065412		129212019214540+490300
CC19FC65		019216520+500600
CC19F075		019229510+500000
CC19F082		019222500+511000
CC21FC10	063067412	120180020230500+500750
DC021FC24	C78067412	12315402023040+510600
CC21F039	C75075432	118162020234400+520500
CC21FC51	C82C85132	130152020230420+520700
DC021F062	C82082112	131172019204400+510500
CC21FC75	085094211	118182019170300+510920
CC21FC83	C67075111	110174018000370+510000
CC21FC95067072112	072C72111	100168020218350+510100
CC21F105		094158019000340+500000
LC21F115	C65063112	122170020226320+500500
CC21F125	C90078111	084145019210310+500400
CC21F135	C72C72112	095135019220290+500E5
CC21F145		020220280+49100
DC21F157	078078211	052130020223260+490200
CC21F169078067211		064100019225240+490400
LC21F179		013232230+48100
CC21F192		020233210+48100
DC22FC05		017228570+48100
CC22F015		013222560+49100
DC22F021	067059412	076110012223550+490950
CC22F031075061411		066104011221530+500900

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D C LIM SUN

NBR NBRWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQMINMAXB+FHAXLATECLDC

CC22F044	072065312	C86136011230410+520705
DC22F054	C72070431	C72187011224400+520505
DC22F064	070000222	082167011230380+520205
CC22F074	C70075411	C56162011214370+520100
CC22FC84	C67055432	C00134012226320+520955
DC22F094	072065132	C30137012223310+510905
DC22F104	082075131	C29153012225290+510500
DC22F114	070064332	C55159012229280+510905
DC22F124		012227260+501005
DC23F005	C90082211	042200012148480+510005
DC23F015	C92078421	C40179011145470+510350
DC23F025	C76072421	C47194010300450+510000
DC23F035	C85085111	C53186011000440+510000
DC23F045	C75085422	C55210010214420+510400
DC23FC55	C87082111	C35202010215410+520300
DC23FC65	C78069431	050210010215390+520200
DC23F075	C78082432	C70223012221380+520300
DC23F085	C72072432	C11164011222330+520400
DC23F095	C62050432	C127219011221350+520300
DC24FC05090085413		C5C117011220710+400000
LC24F015		011210690+41090
DC24F025		011215600+421000
DC24F035		011214600+431000
CC24F038	078085411	C46C90011220660+430500
DC24F048		014222640+441000
DC24F058	C75072112	C70110020220630+450700
CC24FC68	072065111	C45112020190620+450300
CC24F078	104C94111	C72122019229600+450400
DC24FC88	094099112	C049124019230590+470800
DC24F098	C99104111	C50099011229570+480400
DC24F108	C99094111	C621360119226560+480300
DC24F120	C85072112	C761360119232510+500200
DC24F130072085112		C221660119236100+400000
DC24F140	082072112	122152020218490+510400
DC24F150	C67059412	C02177020224470+510500
CC24F160065051412		C72190020230460+510400
DC24F170	C47000412	C55195020227440+520100
DC24F180		067061412180198020192430+520100
AC25F006		019000390- 7999
CC25F005	C94C94111	071132011222570+480100
CC25F015	C85090111	C52128020228560+480100
CC25F025	C82C90111	C48148020000540+490000
CC25F035	C85078111	C46134020000530+490000
DC25F045	C94C85111	C52159020225510+490000
DC25F055	C82C82111	C52150020222500+500005
LC25F065	C94C94111	C43152020220490+500005
CC25FC75	C85090111	C49155020223470+500005
CC25F085	C85085111	C69166020226460+510100
LC25F095	C78C85111	C60160020226440+510100
DC25F105	C90085111	C85132020229430+510200

PAS FRMAREAI RESAREAZ RESAREAZ RESAREAZ RESAREAZ RES D D D LIM SUN

NBR NBRWWAAA STQWWAAA STCKWAAA STCKWAAA STQWWAAA STQMINMAXB+FMAXLATECLDC

DC25F115	C82C75111	C7914202023341C+52015C
DC25F124	C82078111	C60132020228400+52020C
CC31F005	078075112	C69130014231490+50085C
CC31FC15		011224480+51095C
CC31FC23	C90090131	C48106011224470+51075C
DC31F033	085090132	111161017240460+52070C
LC35F005	C9C078431	056210020000620+46000C
DC35FC15	C85C90433	068210020220610+47000C
DC35F025	C90090433	C85229020176590+42010C
DC35F035		02G212580+49086C
CC36F005		021224530+51100C
CC36F015		020229520+51100C
DC36F025		020229500+52100C
DC36F035		02C227490+52100C
CC36FC45		019227470+52100C
CC36F055		02C232460+53100C
DC36F064085094412		C66152020226450+53090C
CC36FC70		020226440+53100C
CC36FC80		020236420+54100C
CC36FC90		020226410+54100C
CC37F005		020210540+50100C
CC37F015		020228530+51100C
CC37F025		C21226520+51100C
CC37F032	075082411	C70125021224510+51010C
CC37FC40085082411		068128019225500+52085C
CC37FC50	C78C75411	C7216801321430+50030C
CC37FC60	C75C85131	C56130013222420+53040C
DC37F070	C82C78431	C50104012222400+53040C
DC37FC80	C95C90111	C70142012219390+52020C
CC37FC90	C78C82111	C79140012220370+52030C
CC37F100	C82C85111	C84123012204390+52010C
CC37F110	C94C85111	C91127012180340+52010C
CC37F120	C94C85111	C75144013220320+52010C
CC37F130		012221310+51095C
LC37F141	C78C72412	C6410001321629C+51090C
CC37F151		07207841207713601622220C+51090C
CC37F161	C85C90411	C54148C20228260+51025C
CC37F171	C85C85111	C73163020224250+51010C
CC37F181		C20224230+50090C
CC37F191		C020222220+50090C
CC37F199		C2C226200+50090C
CC38FC10	C82C94111	C8C136020229580+49035C
DC38F020	104118111	C4917202Q000570+49035C
CC38F030	C95C90111	C52185320000550+50000C
DC38FC40	C90C82111	C4C227018160540+51005C
LC38FC50	C94C99411	C49223013228490+52000C
CC38FC60		C87166011224470+52000C
DC38F063		090094411045187011229470+52035C
CC38F073		063059412089169011223450+52040C
CC38FC83	C94104411	C2922201122644C+52055C

PAS·FRMAREAI RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D DLIN SUB

NBR NBRWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQMINMAX3+FMAXLATECLDC

CC38F093		011220420+521009
CC38F114	075078412	095197011225390+520905
CC38F124	072072431	077205011223340+520904
CC38F134	111085431	067223011225360+53090C
CC38F144	C78085411	055217011227350+53030C
CC38F154	C85078411	106223011227340+53C70C
CC38F164	C78082411	064177011231320+53080C
CC38F174		078075411099163012231310+53090C
CC39F184		011225290+53100S
CC38F194		011225290+53100S
AC40F006		011000390+53100S
CC40F006	061055112	080134015224530+43095C
CC40F016	C94104111	060115018217520+48065C
CC40FC26	C94111111	044142017000500+49000C
CC40F040	C85C90111	050140019000490+49000C
CC49F010061063411		088110018210220+52060C
CC49FC25055055412		090140019228161+28020C
CC49FC33		000052412114132019231191+27090C
CC49FC39		019220191+26090S
CC52F006	C61055411	049124018230530+51050C
CC52FC16	C78072111	049133018224520+52020C
CC52FC26	C67063111	068164018227510+52050C
CC52F036	C72072111	048104017211111
CC52F046	C65065111	044118013218480+53095C
CC52F056	C47055421	045110013220460+54C95C
CC52F066	C85085111	044C80012220450+54085C
CC52F076	067072111	049130012220440+55085C
CC52F086		012229520+52100C
CC52F095		012229520+52100C
CC53F006		017223580+49100C
CC53F012	C82085411	048123018191580+50010C
CC53F022	C72078411	039220018C00560+50009C
CC53F032	C94094111	046114011220500+55090C
CC53F042	085085111	044104016172530+52050C
CC53F052		012229520+52100C
CC53FC62		011227510+53100C
CC53F079		011217440+55100C
CC53FC89		011222420+53100C
CC53F099		011222410+55100C
CC53F113	067063422	105147011221380+55090C
CC53F123	C65065431	048142011220370+55060C
CC53F133	C72075112	079137011212350+55090C
CC53F145		011228340+54100C
CC53F155		011226320+54100C
CC53F165		011223310+54100C
CC53F176	C72078412	00212601122720+54090C
CC53F188	C75075411	046128011218270+54085C
CC53F198	C67070111	045C90011219260+54085C
CC53F205	C75067111	044110011220250+54085C
CC54F005067072412		106139015000450+54000C

# **SECRET SOCIETY**

PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D D LIP - SUN

PAS FRMARE1 RESARE2 RESARE3 RESARE4 RESARE5 RES D D D LIM SUR  
 NBR NBRWWAAASTQWWAAASTQWWAAASTQWWAAASTQMINMAXC+FRAXLATECEDC

CC05AC79		067060212112164020236390+500059
DC05A089		020224370+500309
LC05A096		02G226360+500000
CC06AC05		020232570+460903
CC06A015		020230560+471009
DC06AC25		020234550+471005
CC06AC35		020239530+480955
LC06AC42	067063422	C95162020204420+500600
DC06A052		021222410+501005
CC06AC62		020216390+500900
CC06AC72		021222380+491005
LC06AC82		021226360+490994
CC06AC92		020236350+490951
LC06AC95		C90145020228340+490951
CC06A108		C20234290+491004
DC06A118		C20236280+491005
DC06A128		020238260+480994
LC07AC06	C47052112	C79134022114550+470200
DC07A016	C90085111	05814022212540+470010
CC07A026	C75082211	082113020 520+47
CC07AC36	C67067411	070170320 510+47
CC07AC46	C82085111	072170320 490+47
DC07AC56	C63061412	13815602222680+470000
DC07A068	C63063412	10318002222540+470600
DC07A079	059061412	11117202223340+470400
DC07AC89	085085111	108167022232430+470200
CC07A100	063061412	130232022233410+429500
DC07A110		020235400+480000
CC07A120		022232380+481000
CC07A130		022230370+481000
LC07A141	055062413	103230016220150+480900
DC07A150	067067432	C94234018230340148085
LC07A160		09823502222930
CC07A170	C72067431	100229022235310+480100
LC07A180	C70075431	105144021235290+480800
DC07A190	057059431	102139022 280+48
AC09AC06	072070111	021 300- 6999
CC09AC06		130171019232540+470900
CC09AC16	C67075112	088162019231530+480700
CC09AC26	C78072111	056154019213510+480100
CC09AC36	C94094111	062166020233500+490100
DC09AC46	078072111	082134018232490+490000
DC09AC56072067112		114170019238470+500900
CC15A005	C65070222	128162022237440+500800
CC15A015		022235430+501000
LC15AC17	067060112	124192023236420+500900
CC15A027	078075112	092179022238410+500500
CC15AC37	C70075111	108198022238390+500100
CC15A047	C75072221	C94193021226380+500150
LC18AC05	C90094413	076210022 680+420000

PAS FRMAREAL RESAREAZ RESAREAS RESAREAL AREAS RES C D D LIN SUN

NBR NBRWWWAASSTQWWWAASSTQWWWAASSTQWWWAASSTQMINMAXB+FMAXLATECLDC  
0018A015 C94104413 084222022000670+430000  
0018A025 C94090413 08E1940200003650+440000  
0018A035 C94094413 070176020000640+450000  
0018A045 C94094413 096222020216520+450100  
0018A048 C90C94413 064108022164620+460100  
0019A005 014 740+389999  
0019A015 159218012000730+399999  
0019A019067063422 126214012000720+409999  
0019A029061055422 091202011000710+409999  
0019A039 C67070412 072201021217690+410100  
0019AC55 063069411 087231020186540+490000  
0019A065 020230530+500600  
0019AC75 019230520+500700  
0019A082 020230510+511000  
0021A005 022232510+501000  
0021A008 C61069312 119182020234510+500800  
0021AC18 000065212122178020220500+510700  
0021A028 067062412131171021231410+510700  
0021AU38 072077111 110160320233440+520700  
0021AC48 C68C65112 156229020234420+520800  
0021AC58067071112 129182020204410+510100  
0021A068 078075212 110137022193390+510500  
0021A078075067312 079170020171380+510050  
0021AC88 C82079212 080171021000360+510000  
0021AC98067075212 076184022219350+500200  
0021A108 085082212 092180021000430+500500  
0021A118 078078212 142180020214320+500500  
0021A128 C67060312 128168020222500+500500  
0021A138072077322 078142020222290+490700  
0021A148 022228270+490700  
0021A158 024204100+490100  
0021A162078075211 0621960252152+490100  
0021A172065072211 068142020222390+490100  
0021A182 020234220+481000  
0021A189 019234210+481000  
0022AC15 020232170+481000  
0022A025 075075112 120150021234550+490900  
0022A033 078082112 120158021228540+500900  
0022A043070063423 103172014228420+520800  
0022A053 C72082111 078184013222410+520100  
0022A063 C67061433 096210014230390+520400  
0022A073 C78C72112 044186013230320+520400  
0022A083 C78C82413 102158012226340+520900  
0022A093 C63067433 058142012226320+520980  
0022A103 C82085433 042123012230300+510850  
0022A113 C85072433 051121012221290+510800  
0022A117 061054434 065146013226200+510950  
0022A126 020234270+501000  
0023AC06 C85C94111 077170019174480+510000  
0023A016 C72075411 099180020003470+510000

REPORT

PAS FRMARE1 RESARE2 RESARE3 RESARE4 RESARE5 RES 0 0 0 CLIX SUN

CC23AC26	C90C94111	122186020000450+510000
DC23A036	078072411	130152020000440+510000
DC23A046	057070111	077170019230420+510100
DC23A056	078078111	074162019229410+520100
DC23AC66	C67072431	126231018234400+520300
DC23A078	C67072431	120220021230380+520400
DC23AC88	C63061431	136201016232380+520800
DC23A100	C61061431	165230011230350+520900
DC24AC05067072433		070156012219710+400900
LC24A015	C78085411	060206011220700+410200
DC24A025		012220890+421000
DC24A035		013210670+431000
DC24AC45		013210660+440000
DC24A047		063055413006096012221650+440050
DC24AC57		012223640+440050
DC24A060	C85078113	080137019230640+450000
LC24A070	072072111	053104019220620+450200
DC24A080	C85072111	050120022230610+430150
DC24AC90	078072111	053124020226590+470000
DC24A100	C82070111	050144020230580+480200
DC24A110	C85082111	050158020226570+490200
DC24A120		010230510+500000
DC24A122	C75078112	092180020226520+490200
DC24A132	C67061112	126164018223510+510500
DC24A142	C67072112	086110021230410+510400
DC24A152	C61052412	128192012222400+510300
DC24A162		022230400+410600
DC24A172	059061411	112200020226500+510200
DC24A178		022000440+520000
AC25A006		000 310- 7999
DC25AC06	C45041212	120154020234570+470500
DC25AC16	C74170111	089180020234580+480050
DC25A026	C78078111	064180020226500+490050
DC25A036	C85094111	053190020 530+430000
DC25AC46	C90.90111	049164018 520+490000
DC25A056	C92085111	058189020220500+490000
DC25AC66	C94.99111	047179020224490+500000
DC25A076	C90.85111	057179020 470+500000
DC25AC86	C85085111	081171019232460+500050
DC25A096	C94090111	083181019228440+510100
DC25A106	C75085111	079174019231430+510400
DC25A116	C75075112	103182019233410+520500
DC25A121	C78078411	072140019229410+520200
DC31A010	059059412	134174018233500+500900
DC31AC20070072112		123202019204400+510950
DC31A034	065063412	128157018232480+510970
DC35A005	C63072411	069187019000620+460000
DC35AC15	C72072431	086215017000610+470000
DC35A025	C63059421	072157017000600+480000
DC35A034		017210590+490500

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D D LIM SUN

NBR NBRWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQMINMAXB+FRAXLATECLCD

DC36A005		063055422	130170018230530+500909
DC36A015			018228520+511009
DC36A025			018226510+511009
LC36AC29		C67057422	118151018232500+510909
DC36A039			018229490+521009
LC36A049			018233470+521009
CC36A056		065060422	114163018232460+520909
DC36AC66	07808C122		094162018227450+530709
DC36A076			018220430+530709
DC36A086			018230420+540955
DC37A005			019230550+491009
LC37AC15			018230540+501009
DC37AC20085072412			072153018229530+500955
CC37AC30			019232520+511009
DC37AC35	078082412		083163019224510+510909
DC37A040	075078122		080107019222510+520909
DC37A050		C65065421	142163019208490+530030
LC37A060		C99104112	083163019210470+530100
LC37AC70		C94104131	083170012224450+530200
DC37AC80076085132			083110181228430+530009
LC37A090			07806711212416013224410+520909
DC37A100		C90082112	070152012209300+520100
LC37A110		C94094111	072145012260370+520030
DC37A120		C90094111	050104012190360+520100
LC37A130			018224340+520009
LC37A140			018221320+520955
LC37A145			078067421076180011212310+520909
DC37A155			014226290+521009
LC37A165		C94099111	046145019229270+510300
DC37A175		C85082111	04515901822100+510100
LC37A185			018222-0+510909
DC37A195			019231220+511009
DC37A198			01822110+510909
FC38A005			019223550+481009
DC38AC14	078067412		084152018232500+480800
DC38AC24		C75085111	055154018000670+500000
LC38AC34		C94082111	053152019001650+500000
DC38AC44		C85078111	045229020205530+510100
LC38A054		C94099411	063116017230490+520100
LC38AC64		C67063412	087195019227400+520500
DC38AC74		C59061432	103210019223460+520150
LC38AC84063070412			086163018230440+510350
LC38A094			018230430+520909
LC38A104			018235410+521009
DC38A114			018234400+521009
LC38A119	070063432		150163019255390+520909
DC38A129	065063432		123215017234370+530800
LC38A139	063063432		089230012230360+530750
DC38A149			060101012231340+530800
DC38A159059063412		C67061412	055224011227300+530800

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D D LIM SUN

NBR NBRWHWAAASTQWWAAA\$TQWWAAA\$TCKWAAA\$TQWWAAA\$TQM\$INMAX3+FMAXLATECLDC

D038A169085075412		055201012230310+530000
DC38A179063063432		03721001122F300+530000
D038A189072085422		074129011230280+530959
CC38A192		011230270+531009
AC40AC06		012 370-109995
DC40AC05	C82085111	086162016208600+470909
CC40AC15	C65072112	018232590+481009
DC40AC18	1C4C88111	118162016228590+480600
DC40AC28	C72085111	066179016196590+490320
DC40AC38		056164016000560+500000
CC49AC05	063061411	017178230+520019
DC49AC13		017230210+510109
DC49AC23		019220141+290209
LC49AC32		072121015219161+280209
DC49AC37		018228171+270909
D052AC05	094094121	05311802C232540+510100
CC52AC15	08506121	062125016226530+500150
LC52AC25	089060111	031153019226510+520300
DC52AC35	C92092111	056120019226500+530150
LC52AC45		019230430+531009
DC52AC50		102154020226460+540100
LC52AC60	085077312	09013802L232460+540600
DC52AC70	C94090111	094151020235440+550600
LC52AC80		116154019226430+550500
CC52AC90		019225410+561009
DC53AC06		019229590+481009
LC53AC16	C63063411	07C101018198580+490580
CC53AC26	C63061411	047123018000560+500000
DC53AC36	C72078111	048101018000550+500000
DC53AC46	072078111	065130016140550+510600
LC53AC56		018222920+521009
CC53AC68		018222500+531009
DC53AC78		016230450+551009
CC53AC88		018230430+551009
DC53AC93		018226410+551009
CC53A108		018227400+551009
DC53A122		055052421120190016230380+550850
LC53A136		018226350+550900
CC53A146		018230340+541009
DC53A160		019226310+541009
LC53A170		019228300+541009
CC53A180	C67075112	130195020222460+540900
DC53A191	C70070422	088170019231270+540950
CC53A204	C63061422	110180016226240+540950
DC54AC05	C67071112	100260016224460+540020
DC54AC15	C60060112	088190018196450+540100
DC54AC25	C78070431	07C224016000430+540000
DC54AC35	072078411	082200026000420+540000
DC54AC45		124196018219400+540050
DC54AC55	C65057412	106186018000390+540000

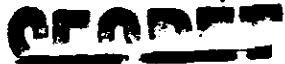
PAS FRMAREAI RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D U LIM SUN

NBR NBRWWAAA STQWWAAA STQWWAAA STQWWAAA STQWWAAA STGMINMAXE+FMXALATELCLOOL

CC54A065		05706241211222018226370+54999C
CC54A075067071421		098218012222350+55040C
CC54A085	078072423	088222012224340+55090C
CC54A095	C67055423	079210016222320+55030C
CC54A105	C72063413	054172010220310+55020C
CC54A115	C72062413	048170011222290+55030C
CC54A125	C75094423	044202012206200+55015C
CC54A132	Cc7063421	059140011220270+55045C
CC55A005		012200610+471009
CC55A013	C78067412	088150012210600+48095C
CC55A023		020220550+48095C
CC55A025	C78-7S112	094140019229580+49050C
CC55A035	C90090111	004154015000570+49000C
CC55A045	C94099111	058164013204550+50003C
CC55A055	C85078112	07610020212540+51020C
CC55A065	085094111	078132020220520+51050C
CC55A075	C82078411	062180020198510+52000C
CC55A085	C82075411	016214020222100+53015C
CC55A095	C85078411	103204018600480+53000C
CC55A105	C72072411	120202018000470+54010C
CC55A115	C82078411	090210020000450+55010C
AC56AC06		019 370-21999
CC56AC06	C78078111	077124019234628+47020C
CC56AC16		019226610+48100C
CC56AC26	057052111	170116019250550+49025C
CC56AC36	C94085111	0501-0019000500+49000C
CC56AC46	C75072111	058166019220560+50001C
CC56AC60	C78075111	080172019232540+51020C

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREAS RES D D C LIM SUN  
 ABR NBRWWHAAASTQWWHAAASTQWWHAAASTQWWHAAASTQMINMAXB+MAXLATECLDC

CC56F		++0
D065F009	082078111094199022229210+560309	
DC65F016	0222271801550305	
DC68F005	060156019228520+530400	
DC68F015	062130013222510+540300	
D068F025	059153013228490+540250	
D068F035	064145013223400+550250	
DC68F045	07013101823460+550400	
DC68F055	048150019229450+560350	
D068F065	05618201322830+560250	
DC68F075	047148016229420+570350	
DC68F085	011222400+570300	
DC68F092	038178009190390+570350	
DC68F102	062160017220380+580850	
DC69F006	059100022219580+500300	
D069F016	0380990222125703510000	
DC69F026	042110022213550+520300	
DC69F044	055151013228400+570200	
DC69FC54	072181016 420+570000	
DC69FC64	055150011 400+570000	
DC69F074	04308101176390+560950	
DC69Fc84	011186370+581000	
CC69F091	074122011204300+560950	
CC69F104	011214340+561000	
DC70FC05078085411	029124019230560+510050	
DC70FC15063078111	090131010231550+520970	
LC70F025	057200011228530+520850	
DC70F035	047167010221920+570300	
LC70Fc45	048122010220510+530500	
DC70F055	0291910101217400+530100	
LC70F065	039187011103400+530300	
DC70F075	044176010000480+530300	
DC70F085	057160017228450+520850	
DC70F095	052131016220400+540200	
DC70F105	092160010225420+550800	
DC70F115	007106400+550200	
DC70F125	037100300+560000	
DC70F135	058110007178370+560200	
DC70F145	005207360+561000	
DC70F147	07807241311416700720750+560950	
DC70F157	046192008212340+570800	
DC70F167	031189008212330+570150	
DC70F177	02119300610310+570100	
DC70F187	037216005220300+560000	
DC71FC05	039186020219500+480500	
DC71F015	039120020180590+490000	
DC71F025	043124019035580+500000	
DC71F035	049140019192560+510100	
DC71F045	060150019230550+520500	
DC71F055	061104019226530+530600	



PAS FRMARE1 RESARE2 RESARE3 RESARE4 RESARE5 RES\_D D\_C LIM SUN  
 NBR NBRWWAAASTQWWAAASTQWWAAASTQWWAAASTQWWAAASTQHINMAXB+FMXLATECLDC

D071F068	104104111	032130019231500+550250
D071F078	C67078422	052130013000480+550000
D071F088	C85075411	090177012000470+560000
DC71FC93	067065412	051170011000460+560005
AC72F007		011000391-119,99
DC72F005	C78078111	035128014224550+510200
DC72F015	C94085111	065162023229540+520300
DC72F025	C82078111	072166023222530+530100
DC72F035	111104111	074118022220510+540200
DC72F045	C99104112	082172022229500+540500
DC72F055	118118111	034130022228490+550400
DC72FC61		022231480+560000
DC73FC06	C85072111	064223022232540+520150
DC73F016	C85078112	056170020228530+530200
DC73FC26	C90C78111	054132014224520+540250
DC73F036		066140014226500+540900
DC73F005	C85087211	035150014219340+580100
DC78F015	C90C94111	081173020224330+580080
DC78F025	C85085111	076154021229310+580400
DC81F006		024225340+580050
DC81FC13		018223330+580050
DC84F005	C82085412	052104014226500+530650
DC84F015		014224540+540900
DC84FC25094104412		C25065010192520+550950
DC84FC35085090412		046095013208510+550950
DC84F045		010214400+561000
DC84F055070085412		048132013215460+570950
DC84F065		013222430+590000
DC84FC75		012226410+591000
DC84F085		010201300+581000
DC84FC95		010202300+581000
DC84F105		010205300+581000
DC84F115		010111340+581000
DC84F119		010204340+601000
DC85FC05		009224610+491000
DC85FC11	085082412	092146010232600+500000
DC85FD21	094094112	044112019220590+510400
DC85F031		019230570+521000
DC85FC41	C78090112	106150019230550+530950
DC85F048	067063412	064122017226550+530800
DC85FC58	C78085411	039140011223470+570100
DC85FC68	C85078411	040153011194450+570030
DC85FC78	C90C99411	063170011000430+580000
DC85FC88099104111		043143008000420+580000
DC85FC98	104094411	031125008163400+580002
DC85F108	C85082412	058119000186390+560200
DC85F118		008206370+580850
DC85F121	C85090131	0411010008206370+590400
DC85F131		077161018233350+590850
DC85F141	104090121	043191021230340+590100

PAS FRMAREAI RESAREAZ RESAREAZ RESAREAZ RESAREAZ RES 0 0 0 LIM SUN  
 NBR NBRWWAAASTQWWAAASTQWWAAASTQWWAAASTQWWAAASTQMINFAXB+FMALATECLDC

DC85F151094094412		063110015226320+590009
DC85F161		0102103006501009
DC85F171		010206290+601009
DC85F180		010201270+601009
DC86F005	085090411	035119012222500+55C40C
DC86F015	C90095411	043158014223490+56040C
DC86F025	104118111	059172014000470+56000C
DC86F035	C78078411	051166014000460+57000C
DC86F045	C75082411	040160014229440+57000C
DC86F055	104104111	042152014224430+58C40C
DC86F065		010211410+581009
DC86F068	078075431	051122010212410+58040C
DC86F078		010205390+590209
DC86F088	078072411	04411201000058C+590009
DC86F098	085094433	046198010210960+600009
DC86F102090078433		048202010208380+600009
LC87F005		008139850+451009
DC87F007	094085411	039091007131650+450909
DC87FC14	067078412	05212901121643+460750
DC87FC24		011222610+461009
DC87F034		096163019226000+460009
DC87FC40	C94094411	044140019142590+490020
DC87F050	118125111	031158017173530+500150
DC87F060	394099111	050110017159280+500020
DC87F070	C85078311	039127017156550+510010
DC87F080	C99099111	039179015000330+520009
DC87F090	104118111	04913201700520+530009
DC87F100	C90104111	054178015000510+54000
DC87F110	C90094111	034140010000410+54000
DC87F120	C82094112	056181010191400+55005
DC87F130	104099111	0531700000000+55000
DC87F140	C82094112	042122000201400+57030
DC87F150	C67070412	041130010206410+58245
DC87F1600	C76082412	036155010210400+58255
DC87F170	C82078412	04413801020810+59045
DC87F173	C78078411	00000001-13999
DC88FC07		010190131+36050
DC96F005		024106010212141+35049
DC96FC10	072078411	010210171+33040
DC96F020		025007010194171+33025
DC96FC22067072411	075067411	037127011219191+31025
DC96F029		049090013219530+55050
DC99F005	C94087112	062098011210520+56050
DC99F015	C67060412	010186500+57102
DC99FC25		061069412052080010187490+57095
DC99FC35		010180470+58100
DC99FC45		010183460+59100
DC99F055		051081010212450+59095
DC99F060	C90075312	084209019225430+60080
DC99F070	C94087112	

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D O D LIM SUN  
 NBR NBRNWWAAASTQWWAAASTQWWAAASTQWWAAASTQMINMAXB+FMAXLATECLOC

D099F080	C99C90111	026166016215420+600400
D099F087		013217410+610089
C100F005	118090111	060168014220+20+600409
D100FC15	118104211	C24094010192410+610099
C100F025	118C94111	021C56010168390+610101
C100F035	C94C94211	047150010220370+610309
C100F045		015177360+620109
C100F055		015214300+621009
D100F065		010197290+620409
D100F075		013228270+611009
C100FC85		013220250+610609
C100FC88		094099211029135013216250+610409
D100F093		012194200+610209
C100F108		0121012104810109
C101F006		012213000+511009
D101F022	C78C94111	079132020225570+530400
C101F032	094099111	043141020228550+540090
C101F042	067072421	02416601421540+550700
C101F052	072072421	04313501620040+560600
D101F062	067067421	036116011221400+560100
D101FC72	075085111	030130014115400+610500
C101F082	078073431	03011201221600+610400
D101F092	085094111	055194014218370+610300
D101F102	072078431	041165012221350+610500
C101F112	075078431	044134012222340+610400
C101F122	075075431	031146012222320+620200
C101F132	067072431	033217012224310+620100
D101F137	082090121	04113201222300+620500
C102FC05	082085412	067140017207500+500600
C102F015	072060312	052149017202500+520050
C102F025	085072312	043147010001570+530000
C102F035	085C90111	045196017200500+540020
C102F050	099104111	037157011227500+530000
C102F060	085085211	030140010217400+580150
C102F070	085099111	03416701121470+590100
C102F085	067075432	049219010220410+610500
C102FC95	037085111	039227010231400+610100
C102F105	085094431	04517803220+5300+620500
C102F114	078082431	043204005197170+620400
A103FC07		008000001-14999
C103FC05		009201000+500600
C103FC11	C78090112	0501101121000+510700
C103FC21	075082112	061142016209500+520400
L103FC31	C82094111	047129016225570+530200
C103FC41		01521300+54100
L103FC51		010191040+55095
C103FC55	C82C90112	028076010210540+560700
C103FC65		012223500+56100
C103FC70	067072412	077122012210510+560200
C103FC80	C99C82411	050112011200500+570100

NBR NBRWWAAASTOWWWAAASTOWWWAAASTLWWRWAAASTOWWWAAASTOKINMAX3+FMXALATELC1DC

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PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D C S LIX SUN

NBR NBRWWHAAASTQWWHAAASTQWWHAAASTCHWAAASTQWWHAAASTQMINMAXL+FHAXLATECLOUDS

C116F197		613120200+60100
D117F006	072078431	6431720122204306030050
D117F016	067070432	538102010194450+600305
C117FC26	082085111	65213491011+450+610400
C117FC36	075078411	60011111111+104610200
D117F046	078078411	64111111111+45010100
C117F056	078078431	65301111111+34620100
C117F066055052411		607151010101070+020100
C117F076		61110101010+621000
C117F086	055-59432	605171010210300+630990
C117F096	067078432	611101010207000+630990
C117F106	059055431	611101010207000+630990
C117F115		600101010101010+630990
C117F127	075082411	630010101010101+704000
C118FC05	078072411	645210101010101+642000
C118F015		611101010101010+630990
C118FC21	085090411	602210101010101+440000
C118FC31	078085411	611101010101010+440000
C118FC41	085085411	611101010101010+440000
C118FC51		611101010101010+440000
C118F061		611101010101010+440000
C118FC71		611101010101010+440000
C118FC81		611101010101010+440000
C118FC91		611101010101010+440000
C118F101		611101010101010+440000
C118F111		611101010101010+440000
C118F114	087085112	602210101010101+440000
C118F124		630010101010101+630000
C118F126		611101010101010+630000
C118F136	094071111	611101010101010+630000
C118F146	111104111	602210101010101+440000
C118F156	090.090.1	602210101010101+440000
C118F166	111099111	602210101010101+440000
C118F176	090.094112	602210101010101+440000
C118F186	099099311	611101010101010+630000
C118F196	090.094111	646111010100400+690000
C118F206	104111111	607000001000+10400000
C118F216	094.085422	630210101010101+61015
C118F226	090.094111	630210101010101+61015
C118F236	076005432	651122010000000+630060
C118F246	085092431	64014201021000+64030
A119F007		611101010101010+10099
C119F008		61002000400+40100
C119F018		611100020+49100
C119F028		611101010101010+1100
C119F038	090.094111	602145010122000+451065
C119F048	085090111	60010201422.000+52060
C119F058	104118111	646171004421.000+53300
C119F068	113125111	653145044220000+54485
C119F078	104125111	646150020000500+55000

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PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D C C LIPF SPC

**CHART**

PAS FRMARE1 RESARE2 RESARE3 RESARE4 RESARE5 RES ARE6 RES ARE7 RES ARE8

C71A015	072072112	047147321222500+49040
C71AC25	067070112	041176022 590+50000
C71A035	095087111	049164022 570+51000
C71A045	094085111	083188022204500+52051
C71AC55	094094111	08014601723500+52051
C71A067	104.11111	064150014204500+54035
C71A077	085085212	049150011200 00+55022
C71AC87	082035412	047176013220 700+55022
C71AC91	063061412	055137011205400+55022
C72AC06		009 351-13999
C72A005	094095411	049130001121500+51015
C72AC15	094094411	041176011220500+52021
C72AC25	104111111	041142011122500+53031
C72AC35	104111111	083148001112 00+54011
C72A045	134134111	083130101111500+54035
C72A055	104094111	08312201222500+55055
C72A061	094104111	019148011231475+56035
C73A005	085082111	009146011220500+52051
C73A015	104.04111	083116011220500+53031
C73A025	094099111	0411701120 00+54011
C73A035	094104111	083150011220 00+53070
C73A039	104118111	049136014212340+58015
C78AC05	082094111	011 114200 00+51015
C78AC15	118125111	083147011211500+53031
C78AC25	094111111	0491370114225110+51015
C78AC31	094104111	017210220+56020
C81AC05		011224210+55020
C81A013		052130011222500+52065
C84AC05	059059422	083130011220 00+53031
C84AC15	063067242	083072412111220 00+53031
C84AC25		083130011212110+55035
C84AC35	063065412	108142011222500+56090
C84AC45	059067412	088116012220410+56090
C84AC53067072412		08211601120430+60080
C84AC63090078111		01020110+58100
C84AC73		0102020200+59100
C84AC83		010193010+59100
C84AC93		010171030+59100
C84A103		010171030+59100
C84A113		010163030+59100
C84A116		010201030+60100
C85AC05		010201030+48100
C85AC15		08019300+49090
C85AC18		043072009209000+50060
C85AC22	085176112	009145019200 00+51015
C85AC38		015226570+51100
C85AC46	072065122	083090110216500+52065
C85AC57	094111121	044144009110400+57010
C85AC67	065072411	080160011203430+57002
C85AC77		049174000 440+58
	072094421	

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PAS FRMAREA1 RESAREA2 RESÁREA3 RESAREA4 RESAREA5 RES D C U L I S

PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D D LIX SUR

NBR NBR WWWAAA STQWWAAA STQWWAAA STQWWAAA STQMINPAXD+FRAXLATELCLOC

DC99A050		000194470+501009
DC99AC60		009192450+591009
DC99A064	125111411	050130009221450+590959
DC99A074	C94094411	050120011211450+590959
DC99AC84	C78085411	050171014211420+590959
DC100AC06	C87C94111	061100011211420+590959
DC100AC16	C78085111	04014001120+590959
DC100A026	094094211	04415001122110+590959
DC100A036078082111		040141011 0004001071
DC100A044	C85C88111	05214601120+570+620019
DC100A056		050200011211420+590959
DC100A067057061411		0501120112220110+590959
DC100A077		0112210710+590959
DC100AC87		0111900100000000000000
DC100A092065065112		00010400110 000+000000
DC100A107		010160221+000029
DC11ACC9	059063412	0020720000000000000000
DC101A016		0000000000000000000000
DC101AC27	072072111	17014001120+700000000000
DC101AC36	C63059411	0000000000000000000000
DC101AC49	060157411	0000000000000000000000
DC101AC56	C59059411	0000000000000000000000
DC101AC66	065151411	0000000000000000000000
DC101AC76	C65073111	0000000000000000000000
DC101AC86	070073431	0000000000000000000000
DC101AC96	C72065421	0000000000000000000000
DC101A106	075070421	0000000000000000000000
DC101A116	070070421	0000000000000000000000
DC101A126	C78072431	0000000000000000000000
DC101A133	072067421	0000000000000000000000
DC102AC005		0000000000000000000000
DC102AC09082085111		0000000000000000000000
DC102AC19	C70053111	0701200220+5909590000
DC102AC29	04094111	002100022110500+590004
DC102AC39	084104111	0701200221 070+590004
DC102AC49	104111111	0001030000000000000000
DC102AC59	C78090411	07410001142214 04000000
DC102AC69	075082421	0000000000000000000000
DC102AC82	070073421	0072100000000000000000
DC102AC92	C94094413	0002100000000000000000
DC102A102	C94090111	071149 12+000000000000
DC102A112	70007433	0022000000000000000000
A103AC07		0000000000000000000000
DC103AC05		0000000000000000000000
DC103AC15	C94104111	0041111110000000000000
DC103AC25	099104411	0001400000000000000000
DC103AC35	C94111411	0001000000000000000000
DC103AC45		0401000000000000000000
DC103AC47	C59059412	0100000000000000000000
DC103AC57		0001000000000000000000

REPORT

PAS-FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D D E LIN SU

**CRAFT  
WORLD**

PAS FRMARE1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES C C D LTR SUP.

0940854110460963122100514470705



PAS FRMAREA1 RESAREA2 RESAREA3 RESAREA4 RESAREA5 RES D S E LIM SON

C119AC15		01122140-040.010.0
C119AC25		01122140-040.010.0
L119A028		067063412057090011190620+100989
C119A038	094094111	06600001103010400000
C119AC48	095000111	05011111-01-01-01-01
C119AC58	094104111	14117000-01-01-01-01
C119AC68		09409011210310-01-01-01
C119AC78	104094111	050112000000-01-01-01
L119AC88	090099111	05410-01-01-0000-01-01
E119AC98	104111111	067140021000510+0000000
L119A108	094094111	0711900010000000+070000
C119A118085082411		0701040010000000+070000
C119A128	072078112	061102012121-01-01-01
C119A138085078412		070105001121-01-01-01
C120AC05		100100-0021-0000-0000-00
C120AC15	090099111	070104002223-0000-0000-00
C120AC25		111100011021-0000-0000-00
C120AC35	9011111.	071100000000-0000-0000-00
C120AC45	091111011	071100000000-0000-0000-00
C120AC55	104111111	071100000000-0000-0000-00
C120AC65	.11104.112	071100000000-0000-0000-00
C120AC75	073094112	104100000000-0000-0000-00

END-OF-DATA ENCOUNTED IN SYSTEM INPUT FILE.

**SECRET  
GARDEN**

The following series of graphs shows the frequency distribution of minimum and maximum terrain diffuse densities and maximum cloud density values recorded in the original negative panoramic photography during Missions 1006-1 and 1006-2. Graphical plots, Figures 9 to 26 on pages 11 to 30, show the density frequency distributions as a function of primary, intermediate and full processing for the forward and aft looking panoramic cameras for these missions. Basic data was measured and tabulated by AFSPPL.

The processing of the panoramic camera film is controlled through the infra-red densitometric measurements of the minimum, terrain diffuse density values. These values are maintained within the density range of 0.40 to 0.90 in order to keep the scene densities on the straight line portion of the characteristic curve.

Analysis of the AFSPPL density measurements for the four film spools shows that the minimum density values from the AFT camera film of Mission 1006-1 went outside of the 0.40 to 0.90 density range more frequently than measured in the other three spools of film. The percentage of values that were outside of the minimum density control range are:

MISSION	FWD CAMERA	AFT CAMERA
1006-1	27.6	42.5
1006-2	27.4	23.0

The excessive deviation from the control range of the 1006-1 AFT camera film is strong evidence of overprocessing and the subsequent loss of information content of the film.

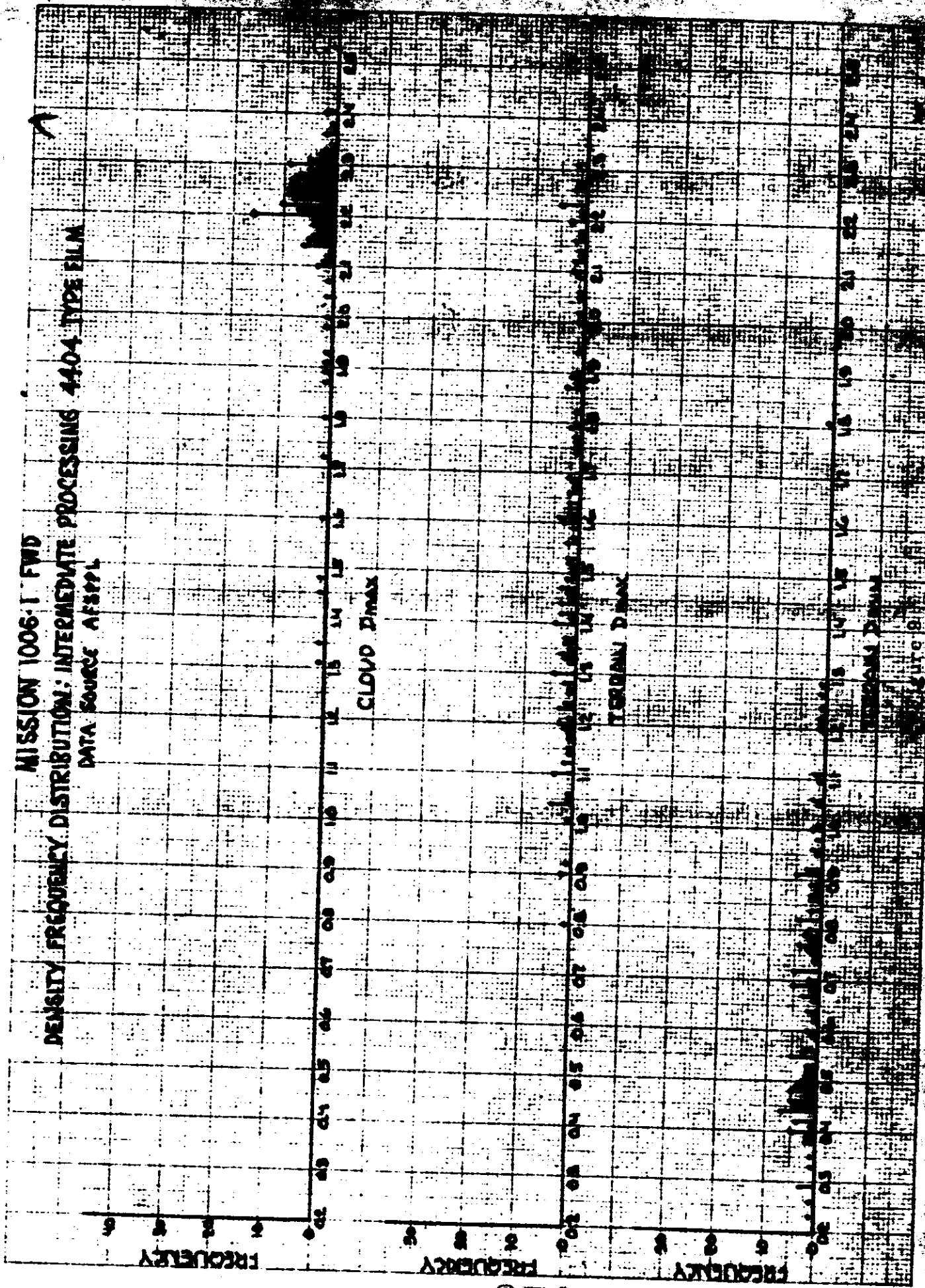
Film processing is varied to produce a minimum density ( $D_{min.}$ ) in the panoramic terrain scene ranging between 0.4 and 0.9 through use of the primary, intermediate and full processing levels employed by [REDACTED]

Minimum density values that cannot be brought within the 0.4 - 0.9 density range by either primary, intermediate or full processing are, by definition, either under- or over-exposed. These ranges are shown on the nominal [REDACTED] processing curves in Figure 27, page 31. Values that exceed the 0.4 - 0.9 density range because of the process employed but could have been placed within range by using a different level process are classified in Table 6, page 11, as either under or over processed, depending on whether the  $D_{min.}$  value is less than 0.4 or more than 0.9.

SECRET  
[REDACTED]

MISSISSIPPI FEDERAL BUDGET

DENSITY FREQUENCY DISTRIBUTION: INTERCENSUS PROCESSING 44-04 THREE ELM  
DATA SOURCE AFSCM



**CRONET**  
**DATA SHEET**

K-22 10 X 10 TO THE CM. 359.14  
REPORT. REPORT. REPORT.

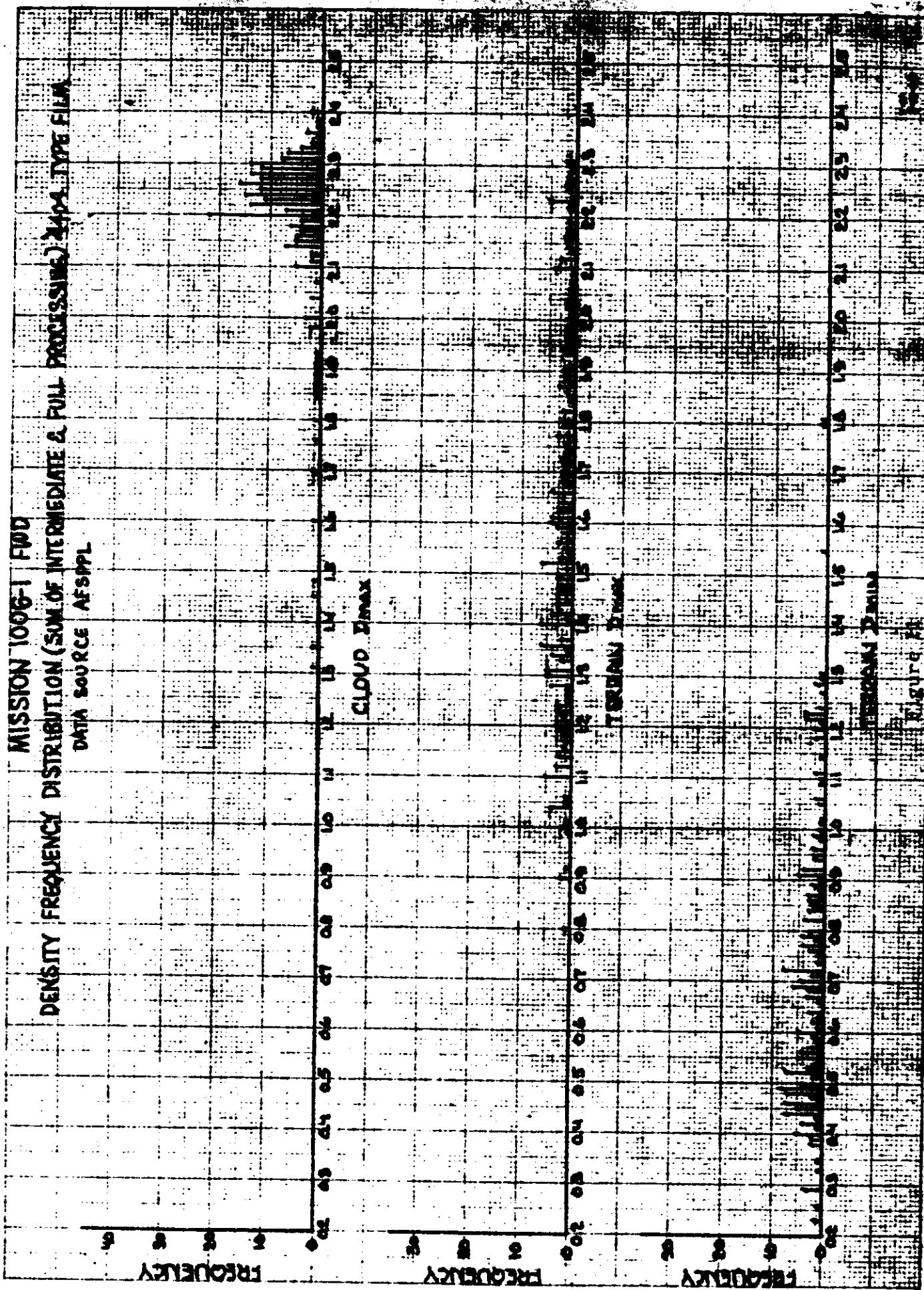
MISSION ID: 000-1110  
MISSION DATE: 1974-07-14  
DATA SOURCE: AFSPC

FREQUENCY READING FREQUENCY READING FREQUENCY READING

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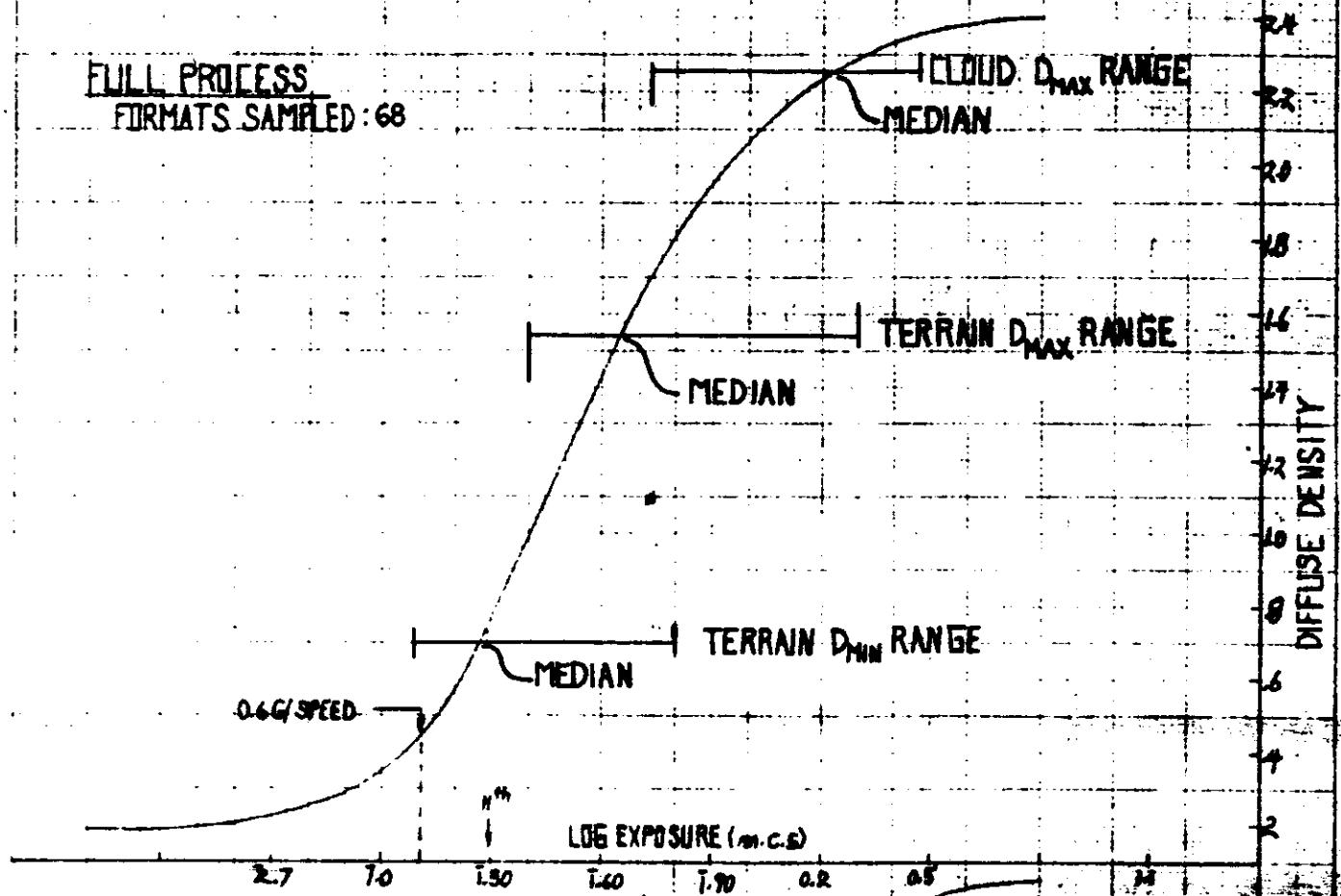
MISSION LOGIC I FPD  
DENSITY FREQUENCY DISTRIBUTION (SUM OF INTERMEDIATE & FULL PROCESSING)  
DATA SOURCE AFSPCL



~~SECRET~~

MISSION 1006-I FWD  
DENSITY DISTRIBUTION VS LOG E. RESPONSE OF 4404 TYPE FILM

FULL PROCESS  
FORMATS SAMPLED: 68



INTERMEDIATE PROCESS  
FORMATS SAMPLED: 140

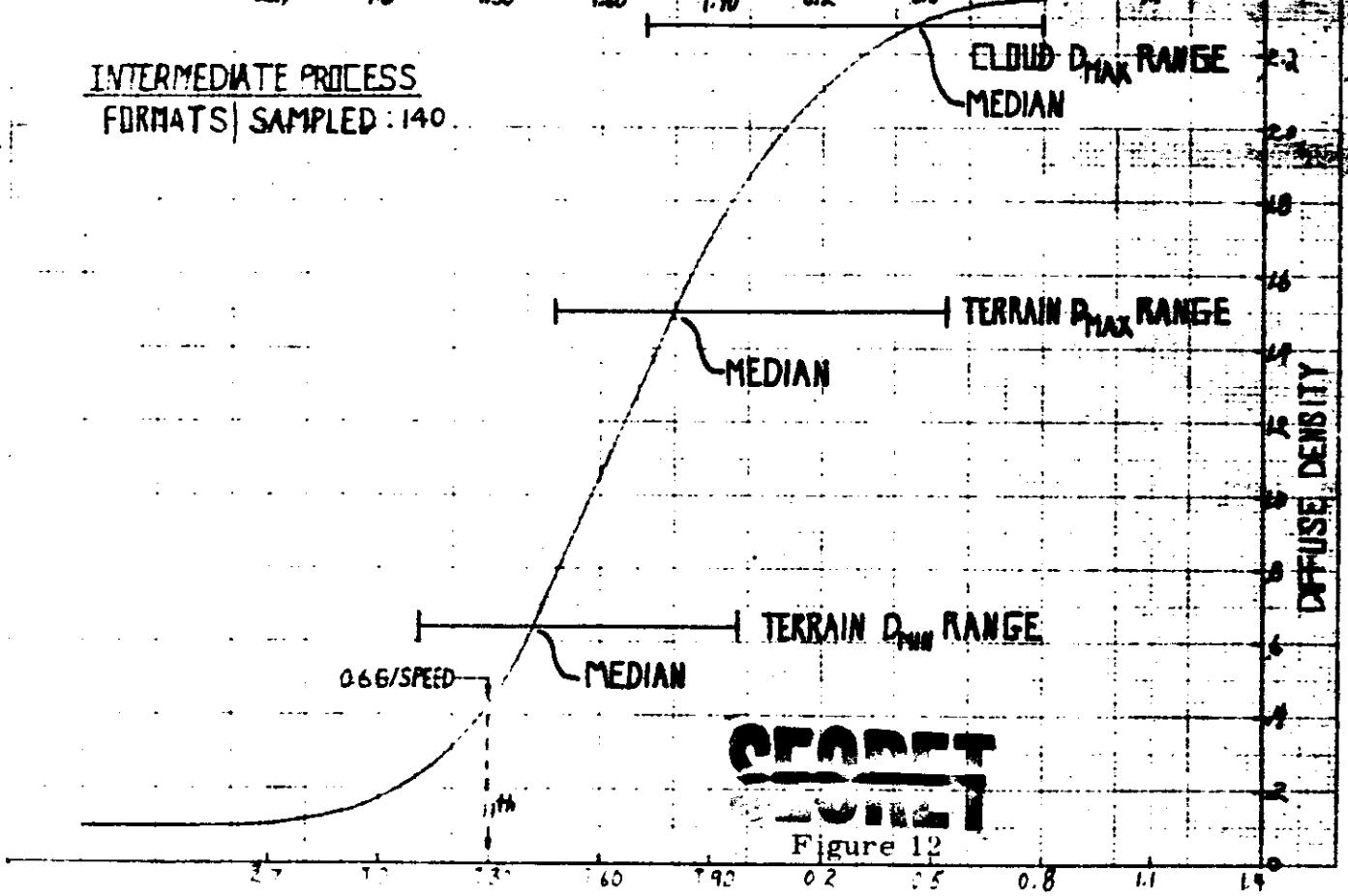
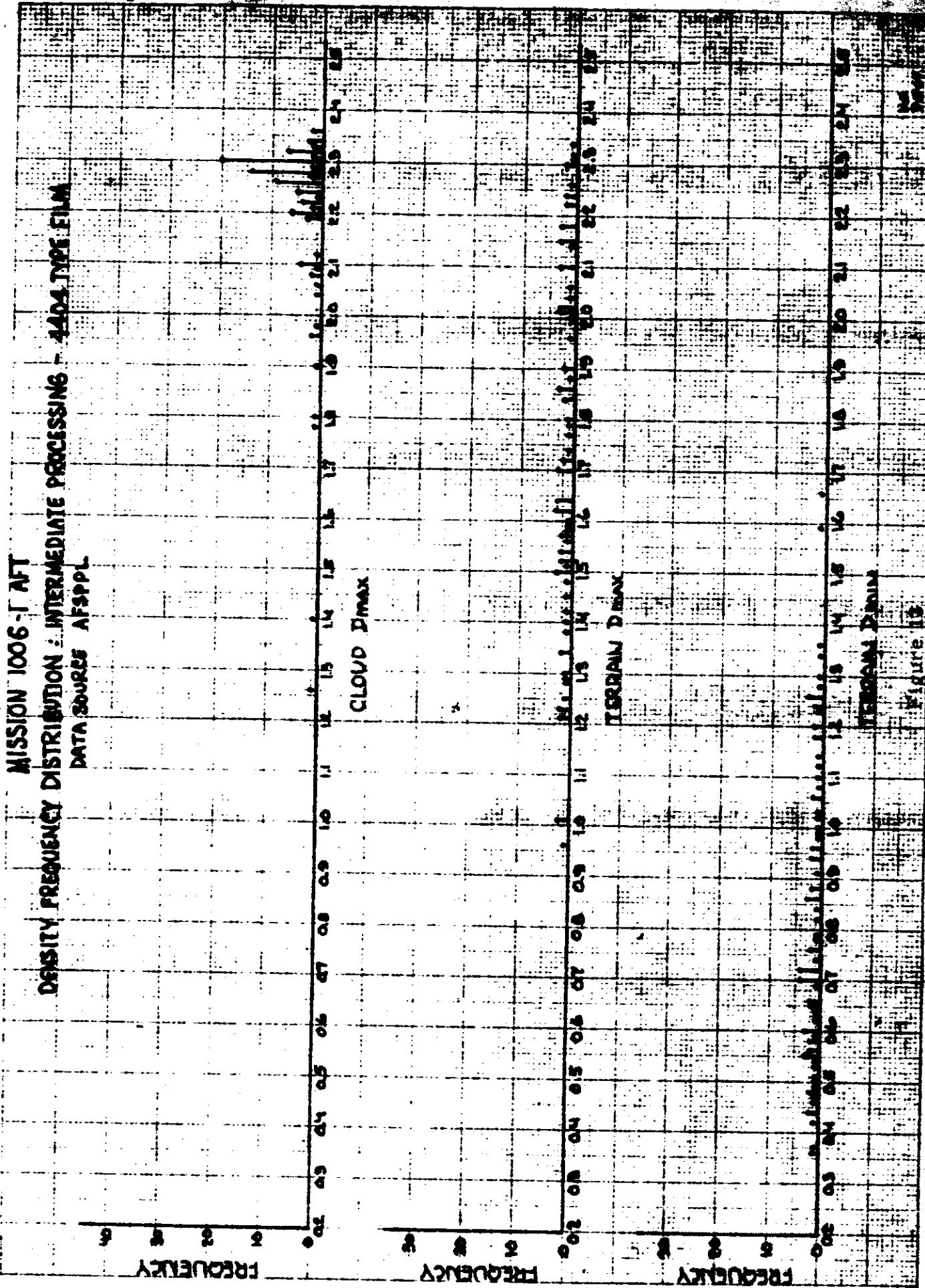
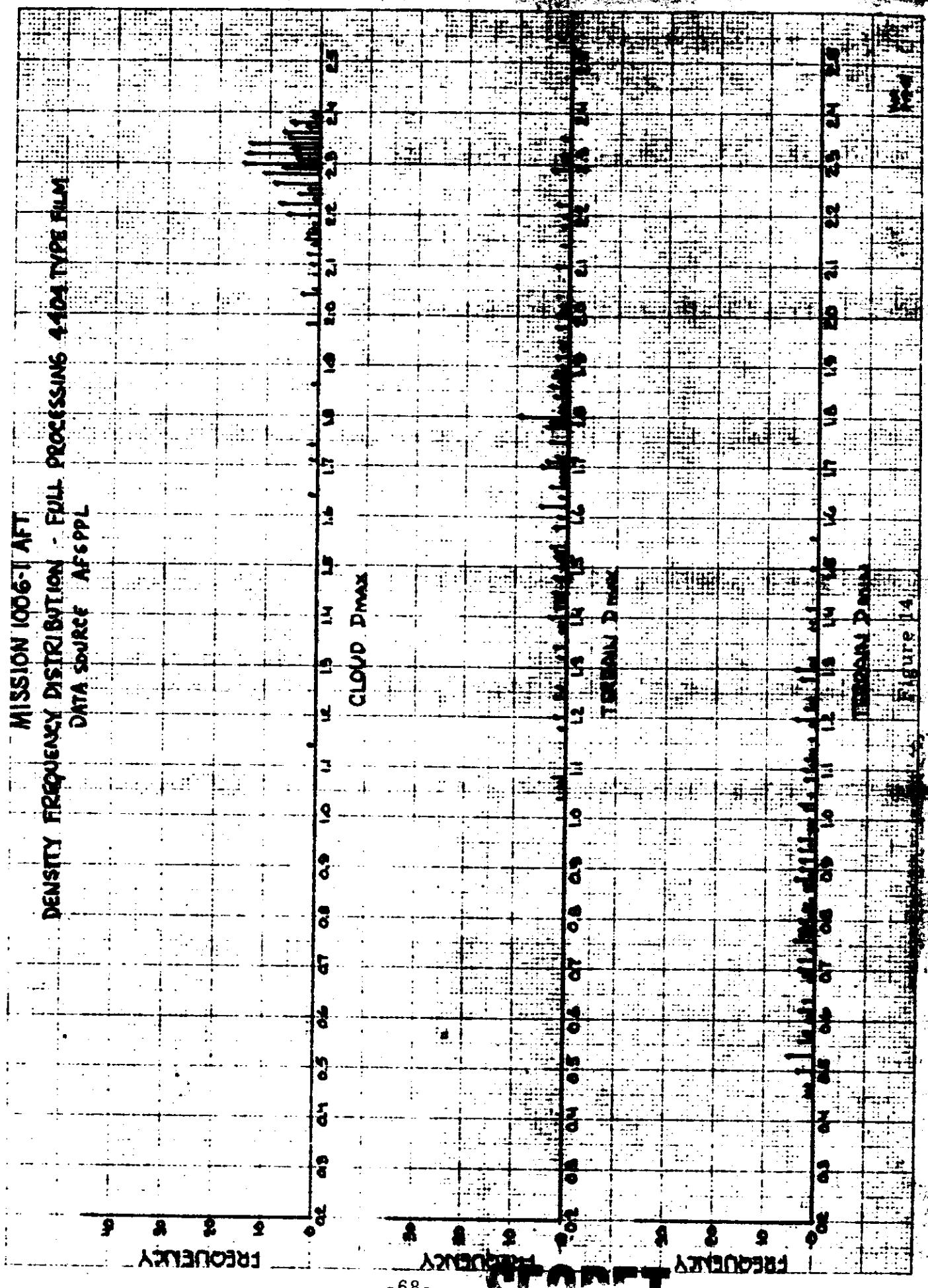


Figure 12

**CRANE**  
**GLU**



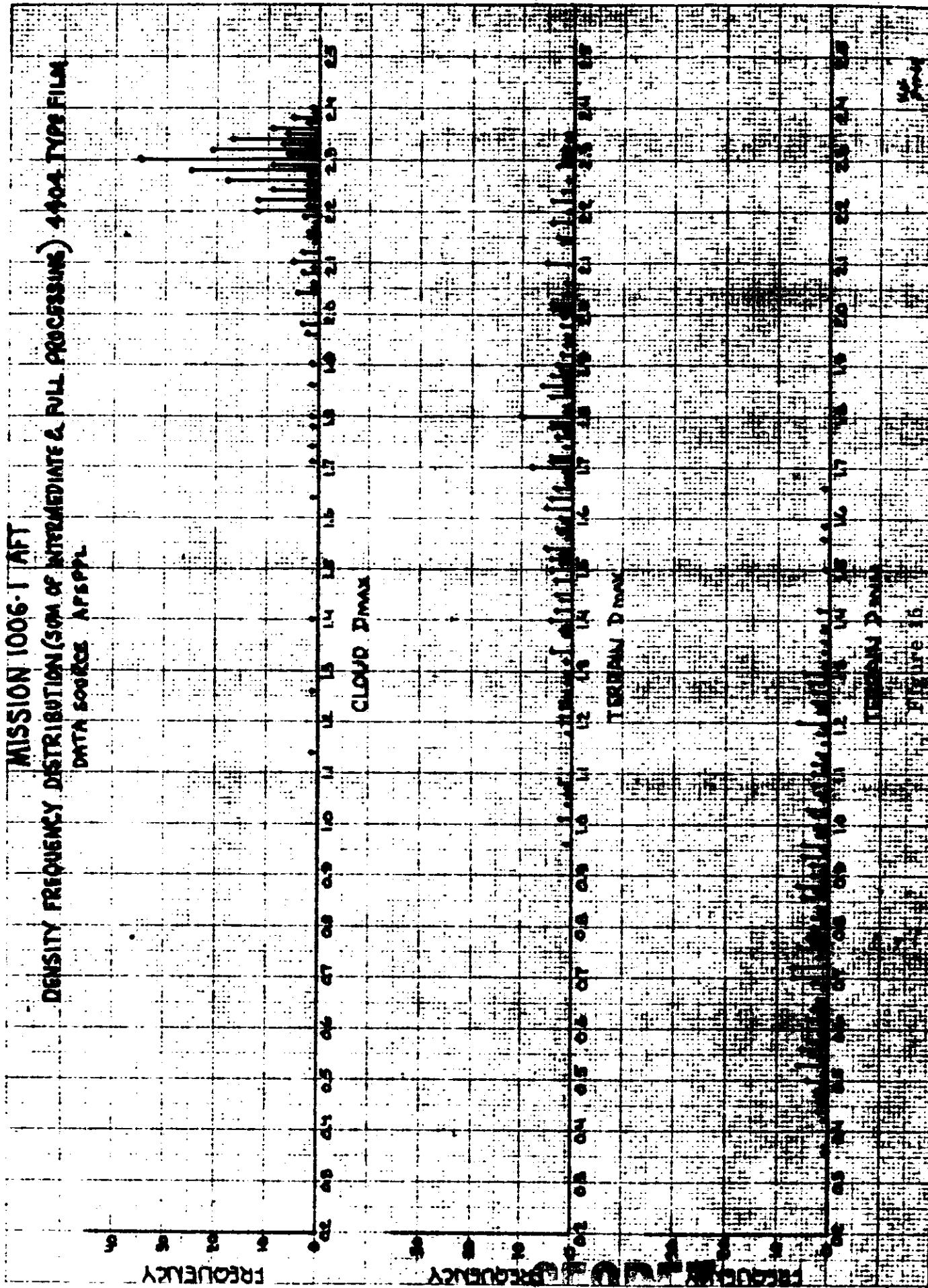
# MISSION 1



CRANE

MISSION 1006· TAFT

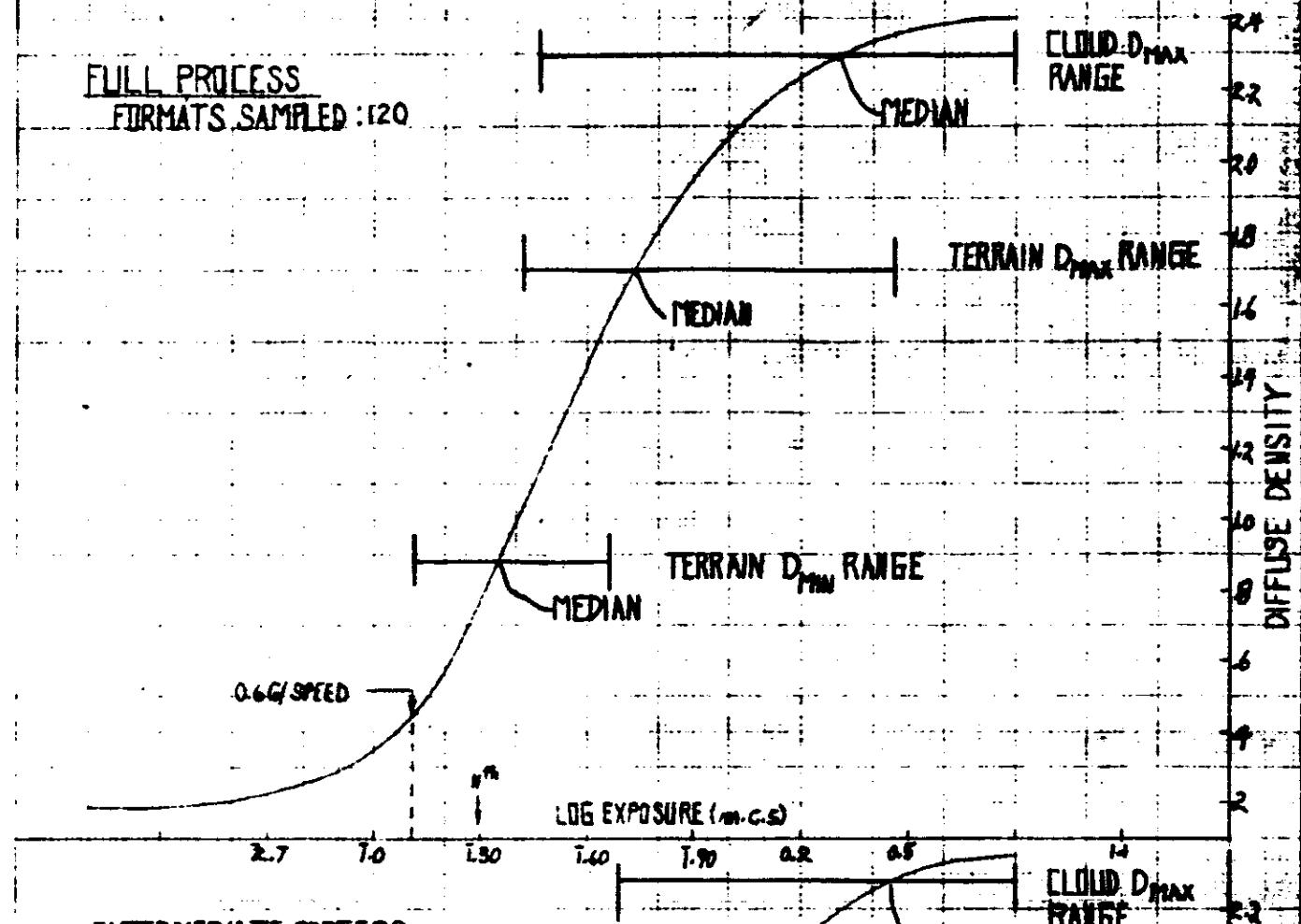
DENSITY FREQUENCY DISTRIBUTION (50% OF INTERMEDIATE & FULL PROCESSING) 4404 TYPE FILM  
DATA SOURCE AFSPPL



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MISSION 1006-L-AFT  
DENSITY DISTRIBUTION VS LOGIC RESPONSE OF 4404 TYPE FILM

FULL PROCESS

FORMATS SAMPLED : 120



INTERMEDIATE PROCESS

FORMATS SAMPLED : 85

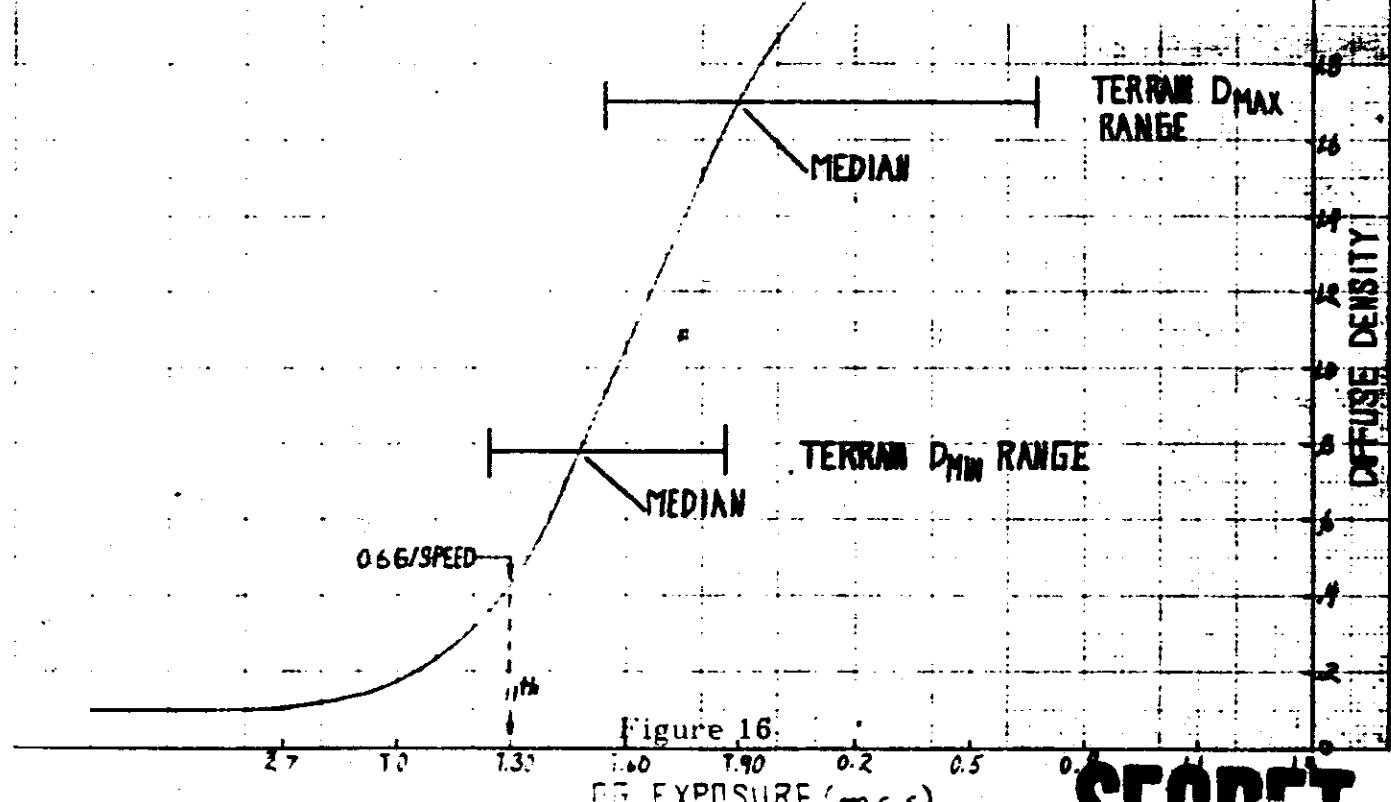


Figure 16.

LOG EXPOSURE (m.c.s.)

**CONFIDENTIAL**

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10 X 10 T-SHELF 369-14

DENSITY

FREQUENCY

CLOUD DNA

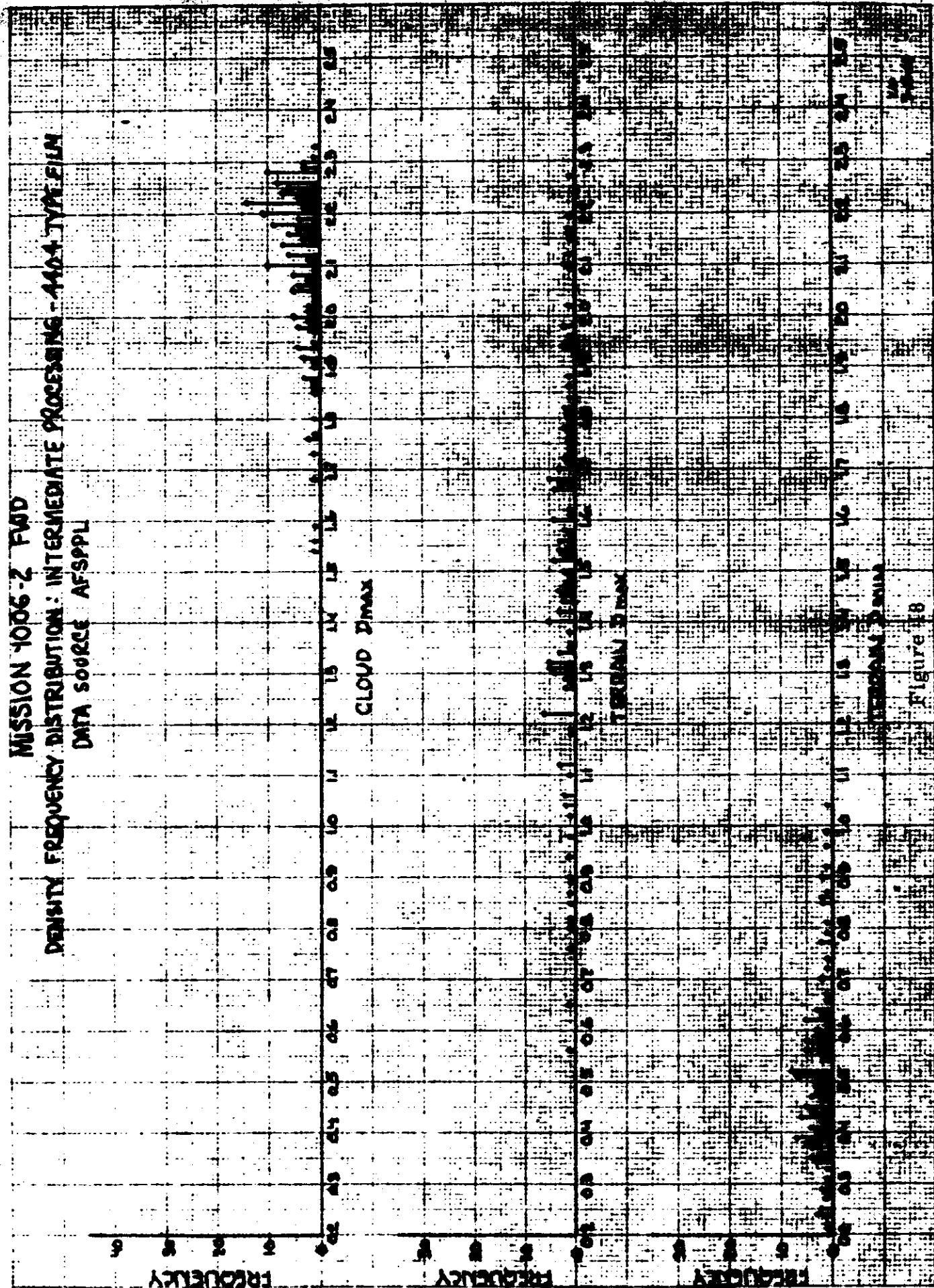
1.00  
0.80  
0.60  
0.40  
0.20  
0.00

10 9 8 7 6 5 4 3 2 1 0

10 9 8 7 6 5 4 3 2 1 0

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~~CRANE~~  
~~ULLMAN~~

NO. 2 FORM 100-14 THE COM 19014



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~~CRANE~~  
~~ULLMAN~~

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MISSION 1005-2 FWD  
DATA SOURCE ACCEPTABILITY

ACCEPTABILITY

FREQUENCY

CLOUD PMS

MISISON 1006.2 FWD

DENSITY FREQUENCY DISTRIBUTION - PRIMARY, INTERMEDIATE, FULL PROCESSING - MOST TIME

DATA SOURCE AFS PPP

FREQUENCY

10 15 20 25 30 35 40 45 50 55 60 65 70 75 80 85 90 95 100

CLOUD DMAX

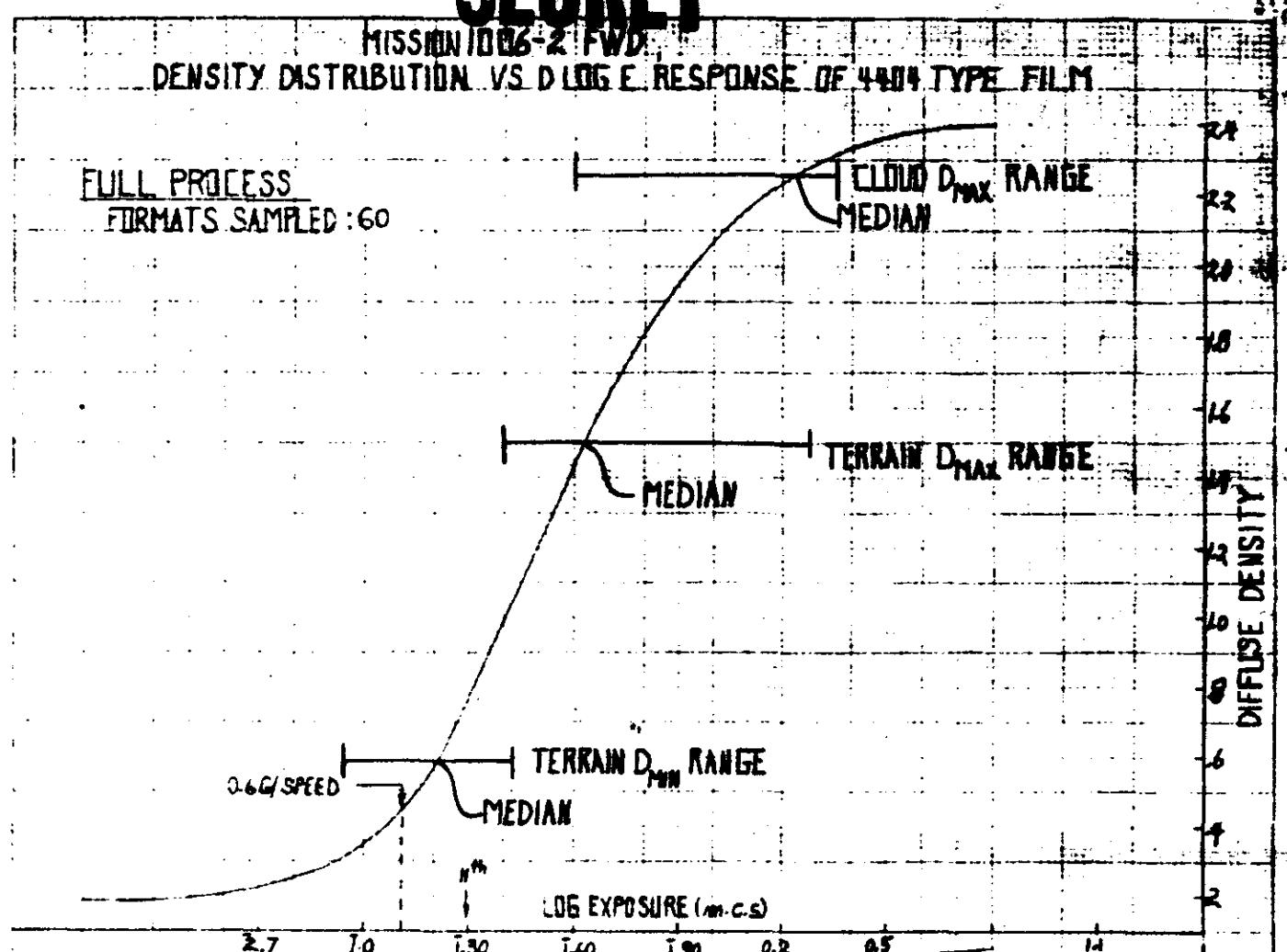
FREQUENCY

2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 27 28 29 30 31 32 33 34 35 36 37 38 39 40 41 42 43 44 45 46 47 48 49 50 51 52 53 54 55 56 57 58 59 60 61 62 63 64 65 66 67 68 69 70 71 72 73 74 75 76 77 78 79 80 81 82 83 84 85 86 87 88 89 90 91 92 93 94 95 96 97 98 99 100

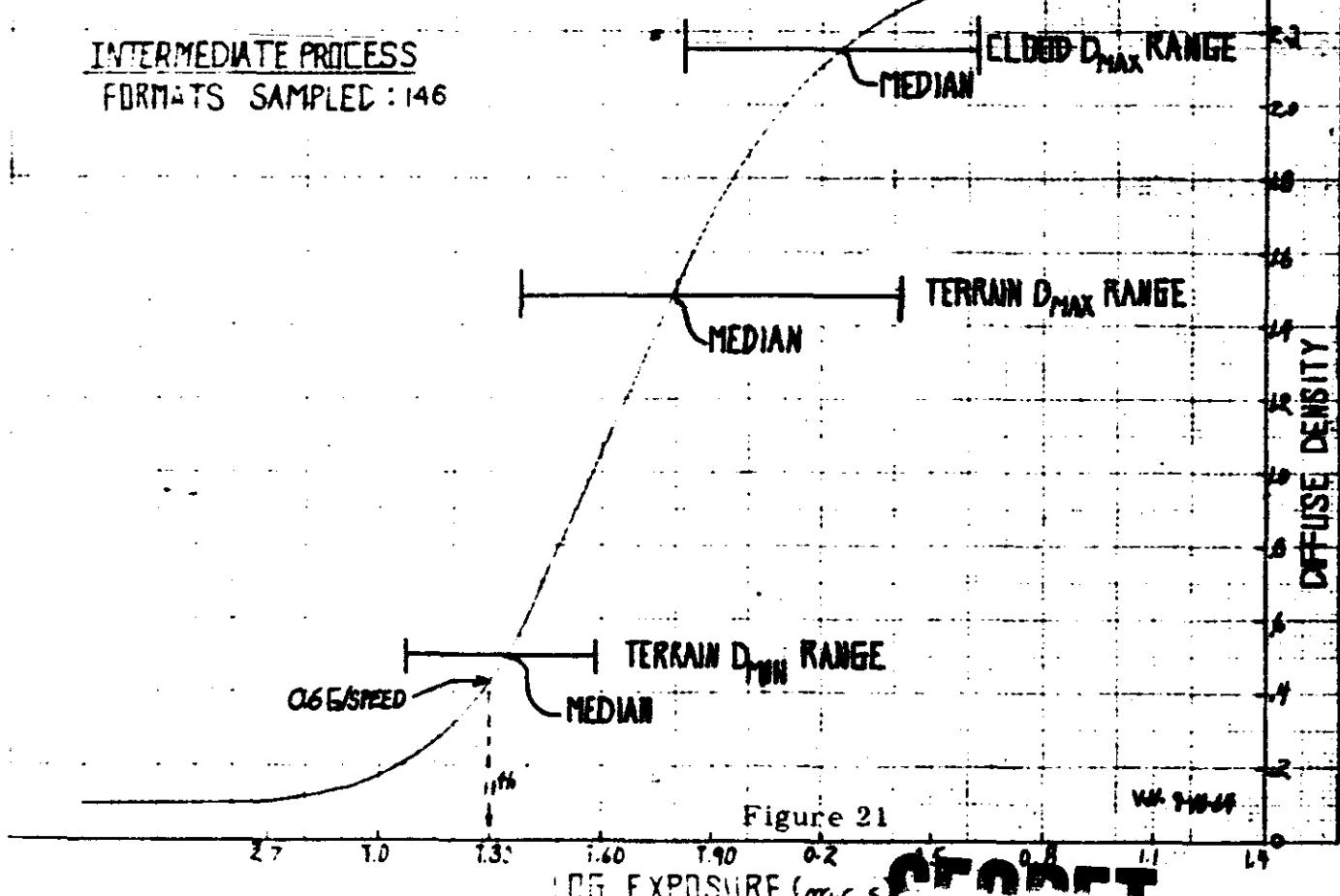
2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 27 28 29 30 31 32 33 34 35 36 37 38 39 40 41 42 43 44 45 46 47 48 49 50 51 52 53 54 55 56 57 58 59 60 61 62 63 64 65 66 67 68 69 70 71 72 73 74 75 76 77 78 79 80 81 82 83 84 85 86 87 88 89 90 91 92 93 94 95 96 97 98 99 100

MISSION 1006-2 FWD  
DENSITY DISTRIBUTION VS D LOG E. RESPONSE OF 4404 TYPE FILM

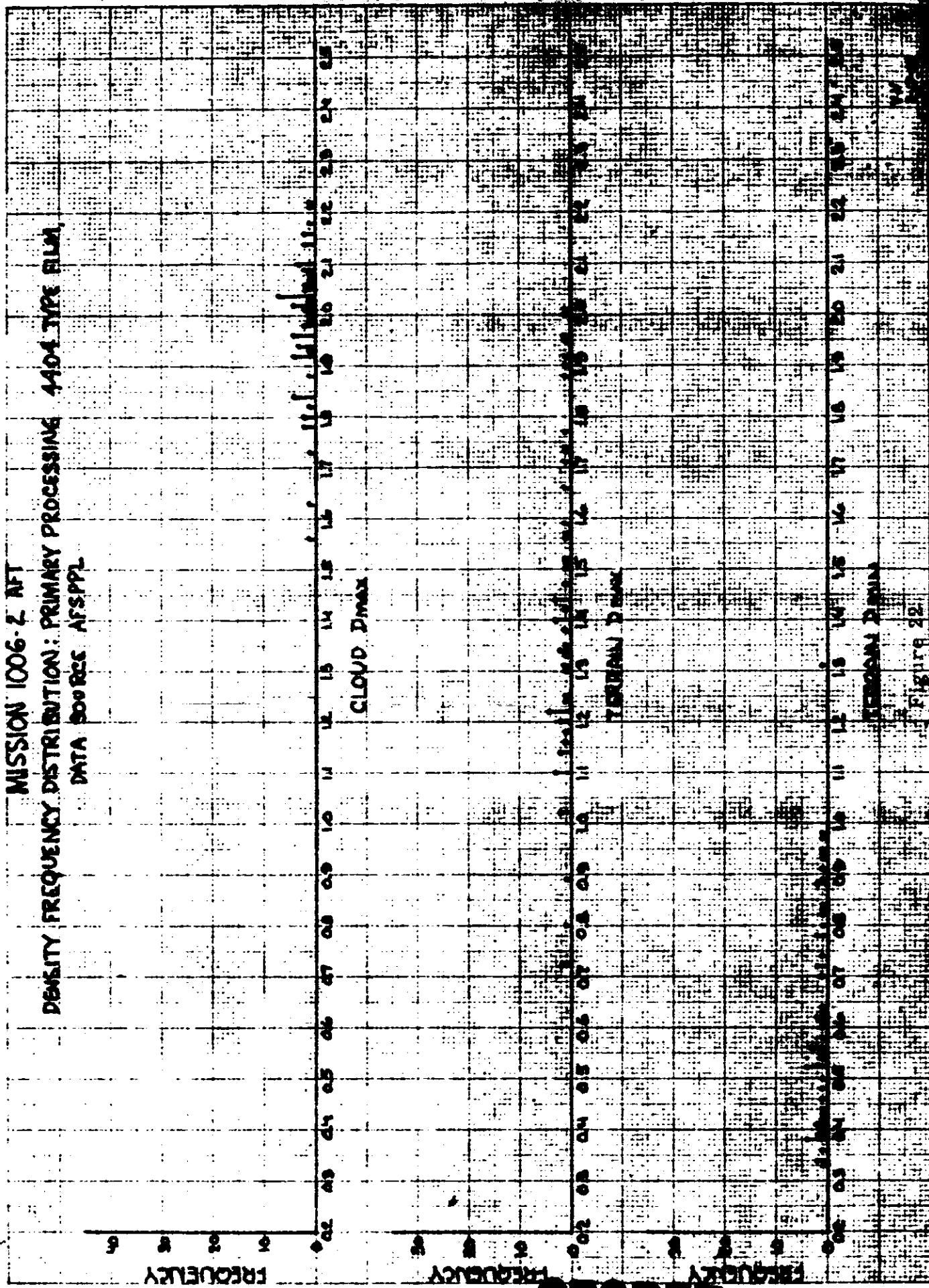
FULL PROCESS  
FORMATS SAMPLED : 60



INTERMEDIATE PROCESS  
FORMATS SAMPLED : 146

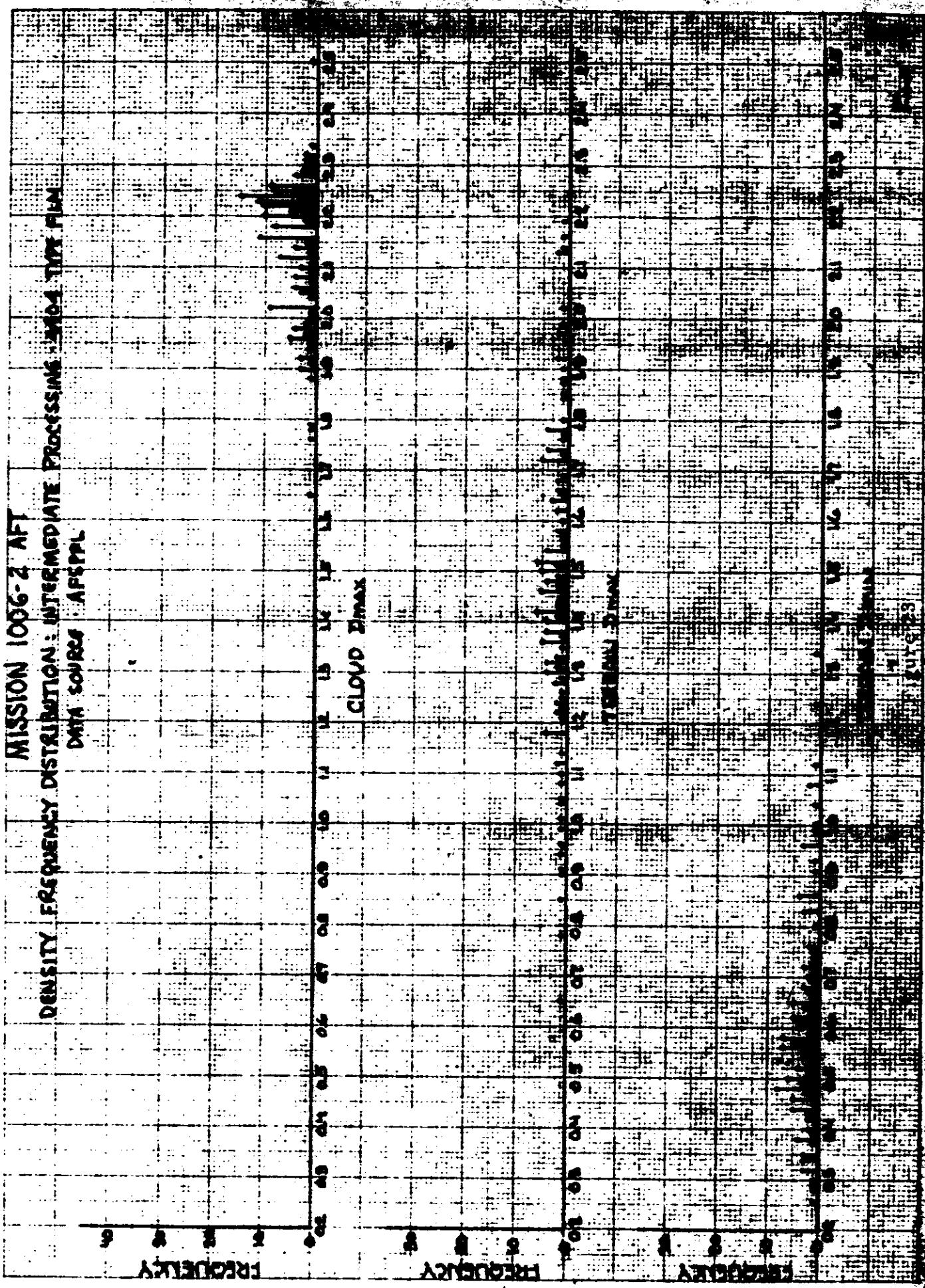


**CRIMSON**  
**SLUGLINE**



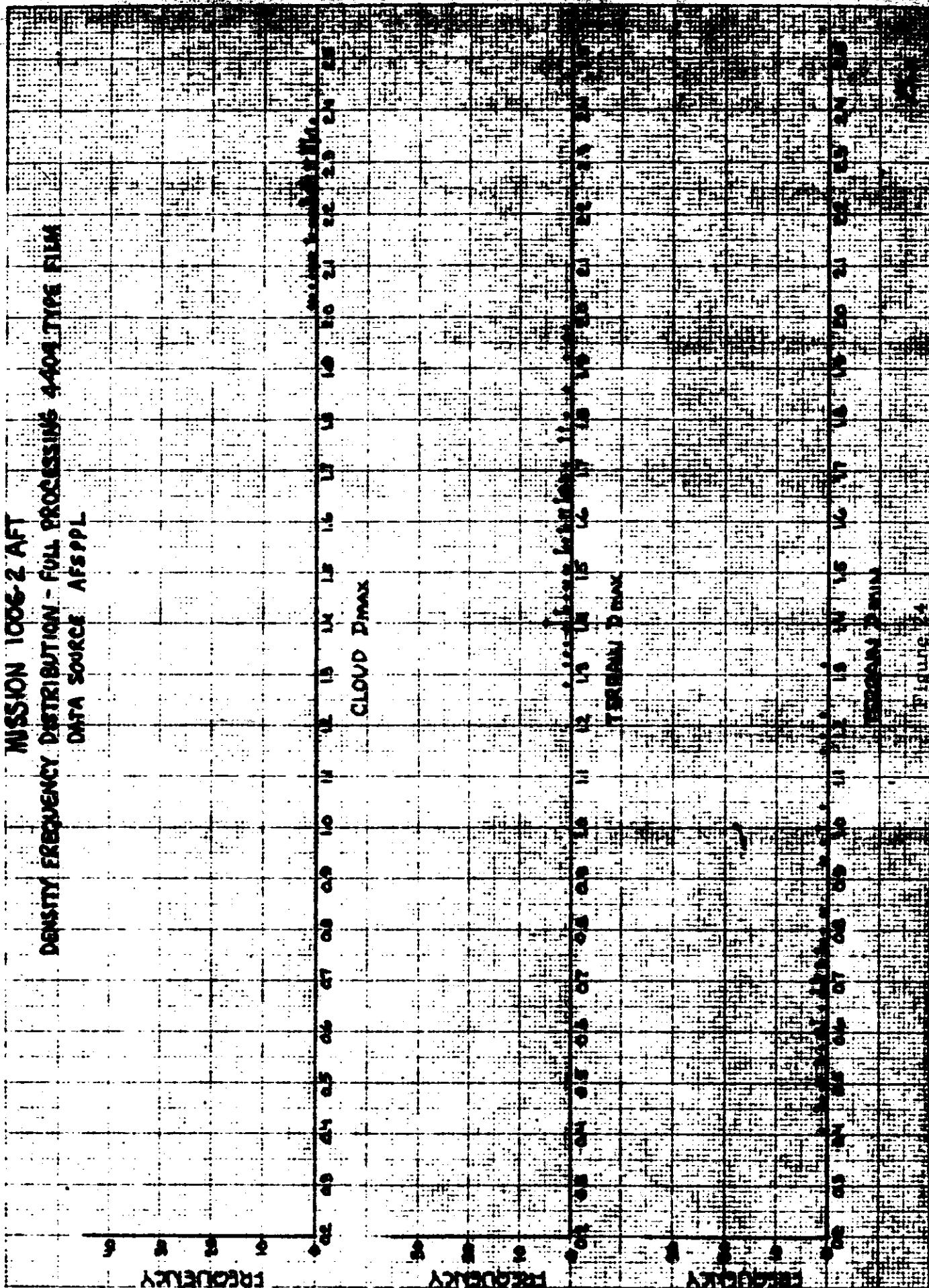
EXCELSIOR CLOTHING CO., INC. 13

DENSITY FREQUENCY DISTRIBUTION: INTERMEDIATE PRESSURE AND TURBULENCE  
MISSION 1006-2 AFT DATA SOURCE: AFSPCR



**CRANE**  
**ULTRA II**

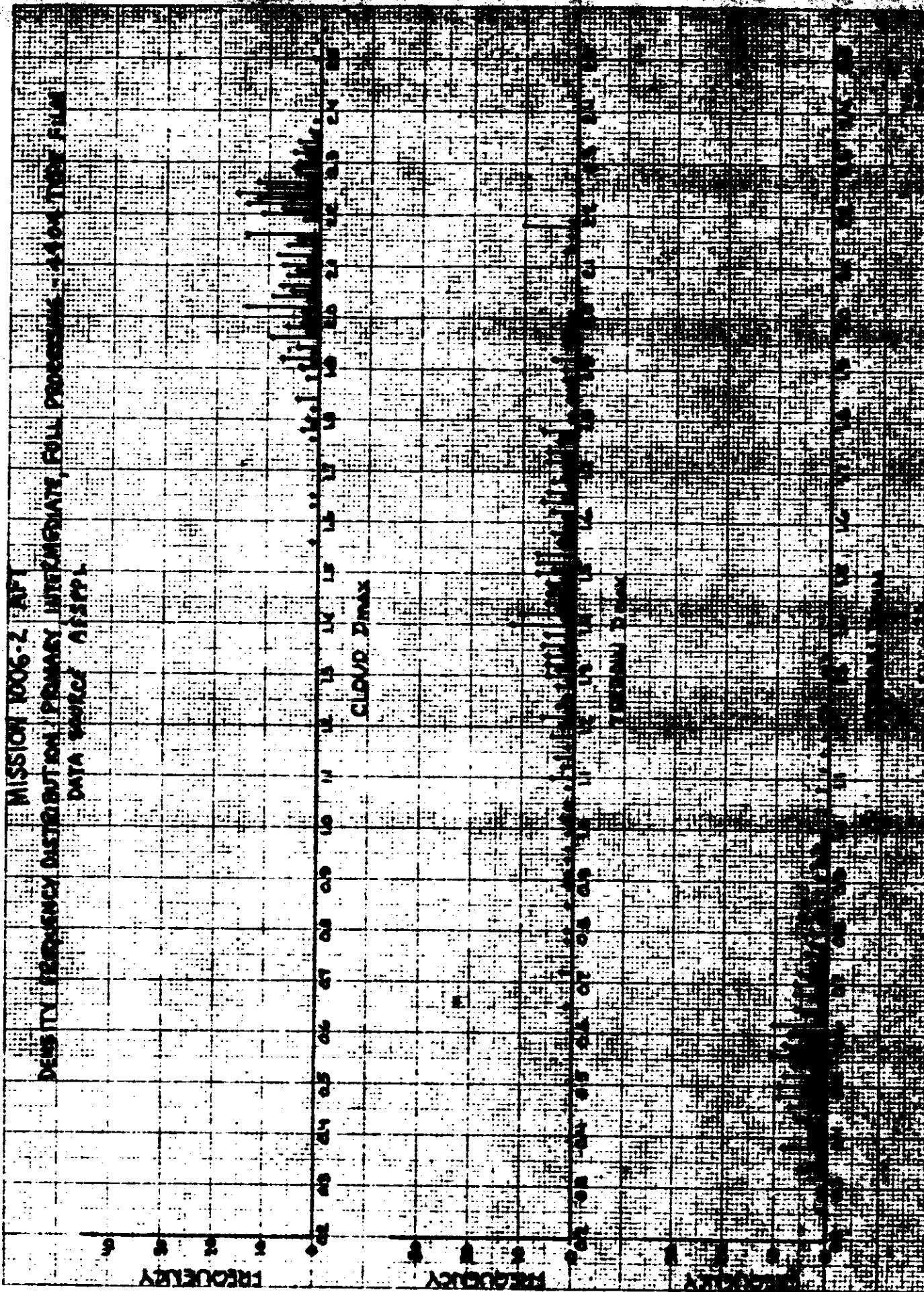
MISSION 1006-2 AFT  
OCEANIC PROCESSOR DISTRIBUTION - FULL PROCESSING 4400 LINE FILE  
DATA SOURCE: AFSPOL



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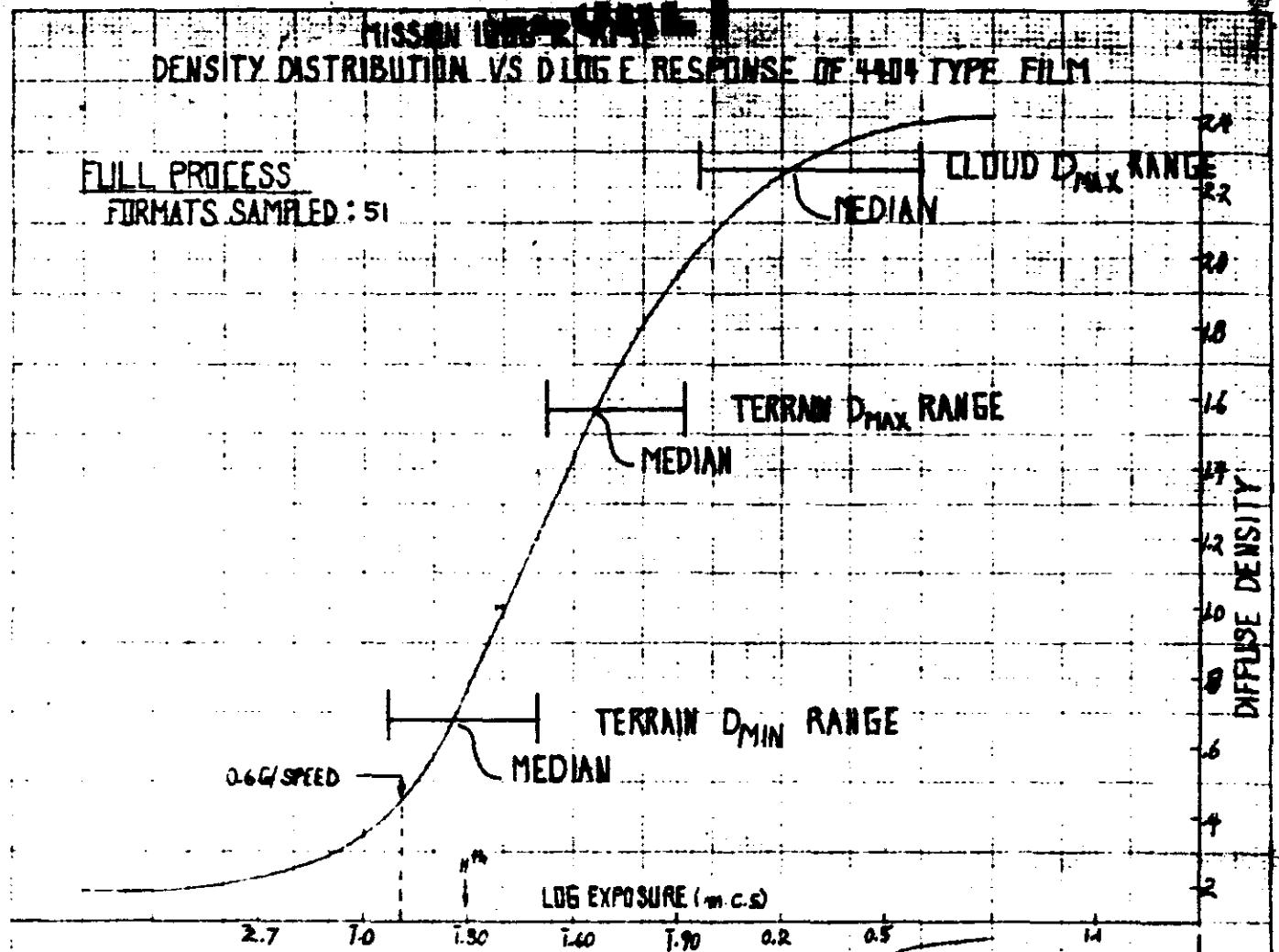
10 x 10 TO THE CM. 359-14

MISSOURI STATE LIBRARIES  
MISSOURI STATE LIBRARIES



DENSITY DISTRIBUTION VS LOG E RESPONSE OF 440 TYPE FILM

FULL PROCESS  
FORMATS SAMPLED: 5



## INTERMEDIATE PROCESS FORMATS SAMPLED: 151

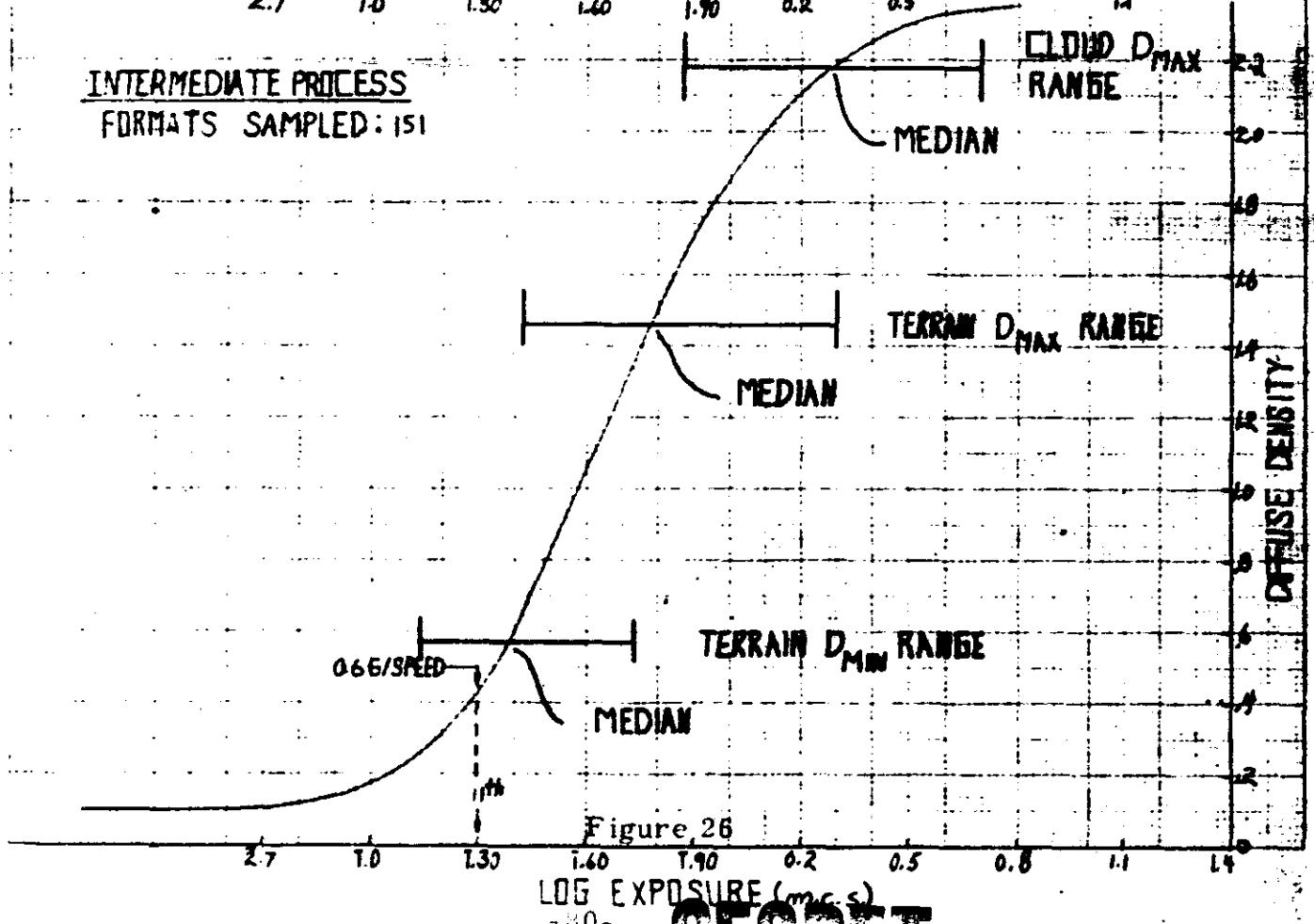


Figure 26

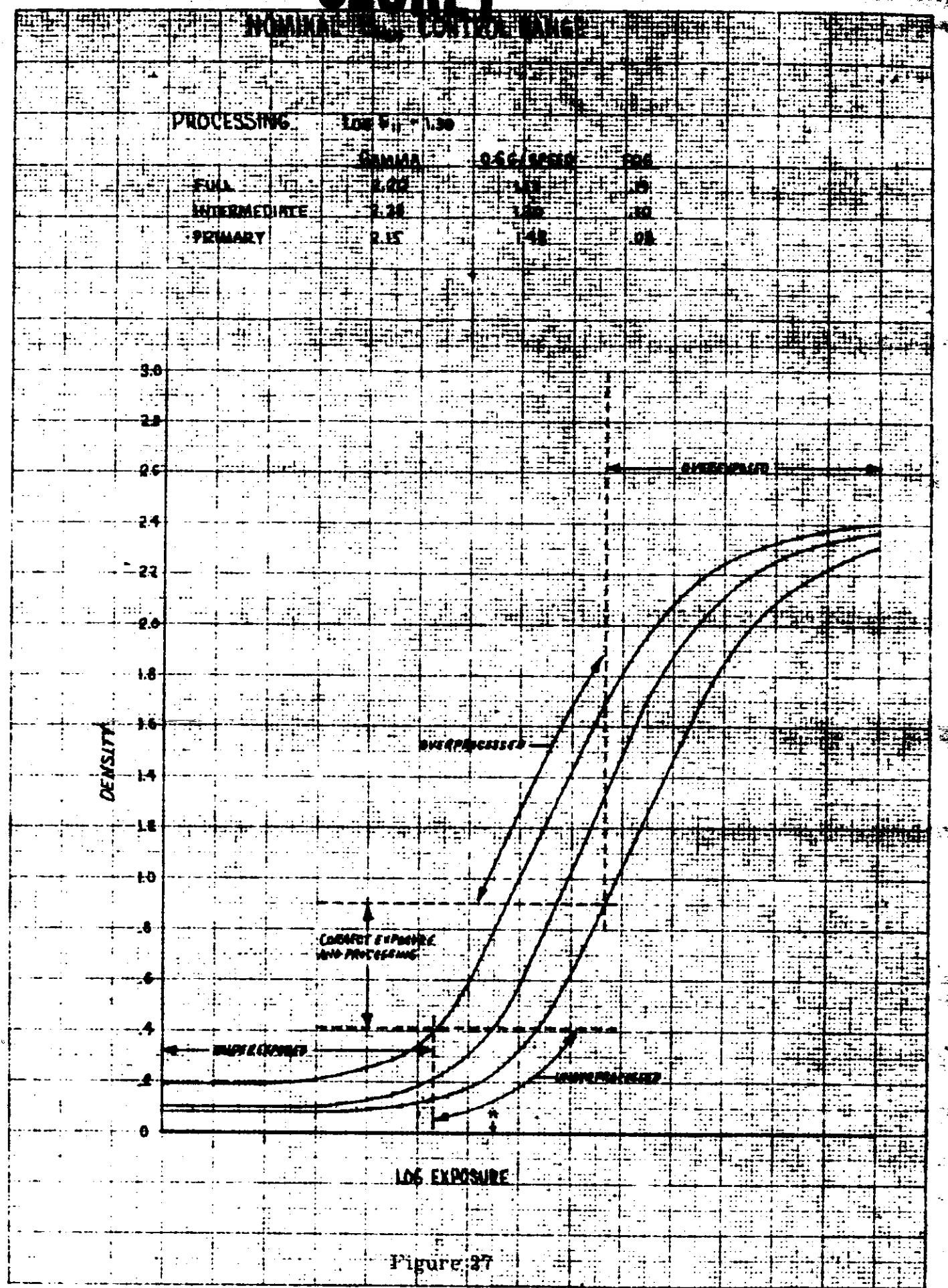


Figure 27

**TABLE 6**  
**PROCESSING AND EXPOSURE SUMMARY**  
**MISSION 1006**

PROCESS LEVEL,	SAMPLE SIZE	UNDER EXPOSED	UNDER PROCESSED	CORRECT EXPOSURE & PROCESS		OVER PROCESSED	OVER EXPOSED
				CAMERAS	CAMERAS		
1006-1 FORWARD CAMERA							
PRIMARY	0	-	-	-	-	-	-
INTERMEDIATE	140	0%	5%	50%	13%	1%	0%
FULL	68	0%	-	23%	9%	0%	0%
TOTAL	218	0%	5%	73%	22%	1%	0%
1006-1 AFT CAMERA							
PRIMARY	0	-	-	-	-	-	-
INTERMEDIATE	85	0%	1%	25%	14%	0%	0%
FULL	120	0%	-	33%	26%	0%	0%
TOTAL	205	0%	1%	58%	40%	0%	0%
1006-2 FORWARD CAMERA							
PRIMARY	25	0%	4%	6%	0	0	0
INTERMEDIATE	161	0%	17%	44%	2%	0	0
FULL	60	2%	-	22%	3%	0	0
TOTAL	231	2%	21%	72%	5%	0	0
1006-2 AFT CAMERA							
PRIMARY	56	0%	4%	14%	5%	3%	0%
INTERMEDIATE	161	0%	8%	47%	4%	0%	0%
FULL	51	0%	-	15%	4%	0%	0%
TOTAL	268	0%	12%	76%	9%	0%	3%

**DRAFT**  
**Version 1**

Table 6, shows the percentage of Mission 1006 film that was exposed and processed to produce the correct D min. value (within the 0.4 to 0.9 density range) and the percentage that was under and over exposed and/or under or overprocessed. The percentages are based upon the samples and sample sizes supplied by AFSPPL.

It is concluded that the deviations from the minimum density control range caused by exposure is insignificant. Essentially all of the values outside of this range were the result of improper processing. The difficulty in achieving proper control with continuous film motion is recognized and the future use of the frame-by-frame processor should help considerably.

The exposure criteria published by [REDACTED] and utilized to establish camera slit width appears to be sound. A comparison of the deviations from the minimum density control range approximates the deviations from the predicted processing range. It is recommended that the present study of the exposure and processing criteria be completed as soon as possible.

The frequency distribution of the maximum density values also indicate that the AFT camera film from Mission 1006-1 contained a significant amount of frames that exceeded the 1.90 density level which is considered the upper limit. The FWD camera film from this mission also had a large percentage of values beyond this desired point. The percentage of readings beyond this maximum density point of 1.90 are:

MISSION	FWD CAMERA	AFT CAMERA
1006-1	22.6	29.2
1006-2	12.4	7.8

SCANNED BY  
[REDACTED]

## SECTION VII

### PERFORMANCE MEASUREMENTS

Several techniques were used to determine the quality of the photographic imagery produced by the panoramic cameras during Mission 1006-1 and 1006-2. These techniques are the Mission Information Potential (MIP) value assigned by NPIC, the Reciprocal Edge Spread (RES) value determined by AFSPPL and NPIC.

The MIP value for both Mission 1006-1 and 1006-2 was 90 for both the Master and Slave camera. Generally, the quality of the Slave was considered slightly inferior to the Master; however, further examination concluded that the variation in quality was due to over processing.

The visual RES values furnished by AFSPPL are listed in Section VI. The average values are:

	Mission 1006-1	Mission 1006-2
Master Camera	78	85
Slave Camera	74	83

The values derived by microdensitometer traces of scene edges on the original negative by AFSPPL are shown for Mission 1006-1 in Tables 7 and 8, pages .. and .., and in Tables 9 and 10, pages 102 and 103 for Mission 1006-2. The data acquisition and reduction process is described in detail in reports published by AFSPPL. The MTF/AIM values are in lines per millimeter.

The complete reports published by [REDACTED] for the edge spread measurements made on the original negative of Missions 1006-1 and 1006-2 are included as Enclosure I, pages 87 to 101, and Enclosure II, pages 104 to 117, respectively. The image location and orientation system shown in these reports is the same system used by AFSPPL.

- The average MTF/AIM resolution in lines per millimeter values for the missions are:

REF ID: A6512

MISSION 10-1 HHS

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## MISSION 1, 10-1 AFT CHAMBER

Mission Date	Pass Nr.	Wavelength Nm.	Inchline PES	Spread Function Width 50% Amplitude			MTF/AIM			Res	Density	Subject
				Lux 350nm	Lux 434nm	Lux 434nm Micros	Lux 350nm	Lux 434nm	Lux 434nm Micros			
105	000	1	67.1	70.5	13.33	14.37	75.0	69.6	63	140	078	1.28
107	010	2	63.7	62.5	12.34	14.45	75.6	69.2	78	101	094	0.96
109	034	3	38.0	43.1	14.68	11.31	68.1	83.4	64	71	072	1.73
109	038	4	70.8	75.3	13.10	11.21	76.3	89.2	78	101	072	1.68
125	013	5	55.6	70.9	14.05	9.91	71.2	101.9	70	120	099	1.11
125	024	6	69.2	75.0	14.40	13.27	74.6	75.4	90	65	075	1.63
125	042	7	67.7	75.3	13.76	12.48	76.6	80.1	70	70	074	1.72
125	076	8	73.6	80.3	14.15	13.55	76.0	72.2	67	69	104	1.64
125	067	9	65.7	79.5	14.72	11.75	78.6	85.1	75	75	092	1.86
124	108	10	66.3	78.7	13.54	11.04	73.9	90.6	64	68	078	1.79
125	039	12	54.2	67.3	17.51	14.29	95.1	70.0	52	58	075	1.73
125	044	13	52.9	54.6	13.21	15.86	94.0	73.0	47	48	067	0.87
125	051	14	76.1	98.9	10.72	8.95	93.3	111.4	92	105	057	1.79
125	056	15	80.2	95.0	12.53	9.27	79.8	103.0	69	120	090	1.42
125	052	16	74.4	63.6	12.62	13.65	79.2	113.3	59	80	078	1.71
125	050	17	68.4	79.9	11.83	14.56	84.5	66.6	77	125	090	1.15
125	091	18	69.0	69.1	13.15	12.33	76.0	80.8	74	73	094	1.20
125	022	19	84.1	92.9	10.24	10.57	95.7	91.6	105	75	055	1.83
125	102	20	49.3	73.6	15.82	9.89	62.2	101.1	78	97	078	1.42
125	103	21	70.5	61.0	11.91	10.10	84.0	90.0	63	95	075	1.91
125	104	22	55.3	46.3	23.44	21.63	42.1	46.2	35	125	075	1.67
	61.4	22.5	12.9	12.61	70.61	82.77	73	73	90			

TABLE 6

~~APPENDIX~~

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ENCLOSURE F

July 1, 1964

TITLE:

Summary of Microdensitometer Derived Image Quality Data Collected from Mission 1006-1

SECTION I: INTRODUCTION

Microdensitometer tracing of scene edges has been used as an objective technique for evaluating photographic system performance. In this report, the evaluation data is presented as spread function width in microns and resolving power in lines per millimeter. A statistical summary of the edge data is presented in Section II, giving the arithmetic mean, standard deviation, coefficient of dispersion, and number of edges. Section III is a tabulation of the location, description, and image quality data for each edge. Frequency plots of the spread function and resolving power data are presented as Section IV, to show the distribution of values.

Section V is included to show the sensitometric data for this mission.

Appendix A is a diagram of the reference system used in describing the orientation of an edge. The edge orientation is recorded as the angular rotation of the edge in a clockwise direction away from a line parallel to the longitudinal direction of the strip of film. Appendix B is a temporary coordinate system used to locate the edges within a frame. A revision of the grid coordinate system is under study.

The image quality data was obtained from sharp scene edges in the original negative by scanning with a Kodak Model 5 microdensitometer. A 1 X 320 micron slit was used. The data reduction consisted of the following steps:

- (a) hand smoothing of the microdensitometer strip chart recording,
- (b) key punching of chart (density) values at sample distance increments of 0.277 microns,
- (c) I.B.M. 1620 computer conversion of chart values to relative exposure values, and
- (d) computer conversion of exposure data to line spread function and modulation transfer function by numerical methods.

The edge resolving power was predicted graphically as the intersection of the MTF curve and the aerial image modulation curve for 4404 film at a test object contrast of 2:1. The spread

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**VELOCITY**

function width was calculated from the first differences of relative exposure as the width at which the gradient became 50% of the maximum gradient.

**Analysis of Photographic Image to Evaluate System Performance**

**SECTION II SUMMARY SHEET**

**Mission 1006-1**

Resolution in lines/mm based on the aerial image modulation - 4404 curve from edge trace data reduced by computer techniques.

Arithmetic Mean	85.3
Standard Deviation	26.4
Coefficient of Dispersion	31.%
Number of edges	93.

Spread function width at 50% amplitude in microns from edge trace data reduced by computer techniques.

Arithmetic Mean	12.0
Standard Deviation	4.3
Coefficient of Dispersion	36.%
Number of Edges	93.

**Analysis of Photographic Image to Evaluate System Performance**

**Mission 100E-1**

**Section III**

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>50% Amplitude Spread Function Width (Microns)</u>	<u>A.I.M. Resolution</u>
1A	Aft.	D07	010	B-8	15	Airfield	15.8	50
2A	Aft.	D07	025	A-9	160	Building	13.9	61
3	Aft.	D09	060	C-5	60	Airfield	15.3	80
3A	Aft.	D06	060	C-5	60	Airfield	6.5	126
4	Aft.	D09	038	A-9	120	Airfield	9.6	64
4A	Aft.	D09	038	A-9	120	Airfield	8.4	99
5	Aft.	D09	038	A-9	120	Airfield	8.8	102
5A	Aft.	D09	038	A-9	120	Airfield	10.1	99
6	Aft.	D09	034	C-3	130	Airfield	14.2	55
6A	Aft.	D09	034	C-3	130	Airfield	18.7	49
7	Aft.	D09	034	C-3	130	Airfield	15.9	66
7A	Aft.	D09	034	C-3	130	Airfield	13.4	62
8	Aft.	D21	084	B-2	120	Airfield	17.9	40
8A	Aft.	D21	084	B-2	120	Airfield	21.1	39
9	Aft.	D21	087	B-5	90	Airfield	11.3	71
9A	Aft.	D21	087	B-5	90	Airfield	11.0	75
10	Aft.	D21	087	A-7	90	Airfield	6.6	127
10A	Aft.	D21	087	A-7	90	Airfield	6.9	120
11	Aft.	D21	088	A-8	90	Airfield	11.0	87
12	Aft.	D21	106	A-4	40	Airfield	6.6	126
12A	Aft.	D21	106	A-4	40	Airfield	9.1	93

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>50% Width (Microns)</u>	<u>A.I.M. Resolution</u>
13	Aft.	D21	125	C-4	40	Airfield	14.4	160
13A	Aft.	D21	125	C-4	40	Airfield	11.5	23
14	Aft.	D24	124	B-3	20	Airfield	14.5	63
14A	Aft.	D24	124	B-3	20	Airfield	12.8	63
15A	Aft.	D24	124	B-3	20	Airfield	10.2	77
16-1	Aft.	D37	085	C-4	100	Airfield	11.2	111
16-2	Aft.	D37	085	C-4	100	Airfield	15.4	75
16A-2	Aft.	D37	085	C-4	100	Airfield	12.8	62
17	Aft.	D52	080	C-9	170	Airfield	8.9	107
17A	Aft.	D52	080	C-9	170	Airfield	11.3	34
18	Fwd.	D09	029	B-12	140	Airfield	12.5	60
19A	Fwd.	D09	029	B-12	140	Airfield	12.1	59
20-1	Fwd.	D09	029	B-5	100	Airfield	15.0	66
20A-1	Fwd.	D09	029	B-5	100	Airfield	19.0	88
20-2	Fwd.	D09	029	B-5	100	Airfield	15.4	37
20A-2	Fwd.	D09	029	B-5	100	Airfield	26.3	39
21	Fwd.	D09	033	C-7	70	Airfield	6.7	127
21A	Fwd.	D09	033	C-7	70	Airfield	13.0	59
22	Fwd.	D09	055	C-9	80	Airfield	6.0	135
22A	Fwd.	D09	055	C-9	80	Airfield	17.1	52
23	Fwd.	D09	055	C-9	80	Airfield	14.5	57
23A	Fwd.	D09	055	C-9	80	Airfield	9.0	135

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50%  
Amplitude  
Spread  
Function  
Width

(Microns)

A.I.M.  
Resolution

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>50% Amplitude Spread Function Width (Microns)</u>	<u>A.I.M. Resolution</u>
24	Fwd.	D21	121	A-10	30	Airfield	7.6	90
24A	Fwd.	D21	121	A-10	30	Airfield	7.5	127
25	Fwd.	D21	084	C-11	80	Airfield	10.6	64
26	Fwd.	D21	083	C-8	90	Airfield	12.0	63
26A	Fwd.	D21	083	C-8	90	Airfield	6.3	90
27	Fwd.	D21	083	A-9	100	Airfield	9.0	100
28	Fwd.	D21	080	B-4	120	Airfield	12.5	53
28A	Fwd.	D21	080	B-4	120	Airfield	6.1	134
29	Fwd.	D21	158	C-12	30	Airfield	11.8	70
29A	Fwd.	D21	158	C-12	30	Airfield	9.3	82
30	Fwd.	D21	158	C-12	30	Airfield	13.5	61
30A	Fwd.	D21	158	C-12	30	Airfield	13.0	84
31	Fwd.	D21	159	C-13	30	Airfield	9.1	108
31A	Fwd.	D21	159	C-13	30	Airfield	8.3	125
32	Aft.	D25	067	C-5	130	Airfield	11.0	100
32A	Aft.	D25	067	C-5	130	Airfield	8.5	92
33	Aft.	D25	024	A-2	10	Airfield	15.6	53
33A	Aft.	D25	024	A-2	10	Airfield	10.7	73
34A	Aft.	D25	020	A-13	115	Airfield	10.9	107
35	Aft.	D25	018	A-B-10	110	Airfield	11.0	55
35A	Aft.	D25	016	A-B-10	110	Airfield	8.6	103
36	Aft.	D25	013	A-6	170	Airfield	9.0	81

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50%  
Amplitude  
Spread  
Function  
Width

(Microns)

A.I.M.  
Resolution

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>50% Amplitude Spread Function Width (Microns)</u>	<u>A.I.M. Resolution</u>
37	Aft.	D40	041	A-1	100	Airfield	9.6	90
37A	Aft.	D40	041	A-1	100	Airfield	9.6	90
38	Aft.	D38	037	B-8	45	Building	5.2	130
38A	Aft.	D38	037	B-8	45	Building	6.4	122
39	Aft.	D38	037	A-8	0	Canal Edge	14.5	90
39A	Aft.	D38	037	A-8	0	Canal Edge	10.7	60
39B	Aft.	D38	037	A-8	0	Canal Edge	6.5	123
40	Fwd.	D07	005	B-7	45	Airfield	12.2	60
40A	Fwd.	D07	005	B-7	45	Airfield	15.1	61
40B	Fwd.	D07	005	B-7	45	Airfield	15.1	65
41A	Fwd.	D07	027	A-10	120	Buildings	7.7	110
41C	Fwd.	D07	027	A-10	120	Buildings	6.5	115
41D	Fwd.	D07	027	A-10	120	Buildings	7.5	133
42	Fwd.	D24	101	C-12	25	Dam	17.3	52
42A	Fwd.	D24	101	C-12	25	Dam	14.5	61
43	Fwd.	D24	101	C-12	115	Building	-	-
44	Fwd.	D24	104	B-12	170	Airfield	14.3	55
44A	Fwd.	D24	104	B-12	170	Airfield	13.2	52
45	Fwd.	D24	120	A-12	30	Airfield	12.6	61
45A	Fwd.	D24	120	A-12	30	Airfield	13.5	53
46	Fwd.	D22	046	B-8	90	Dam	20.5	51
46A	Fwd.	D22	046	B-8	90	Dam	9.9	100
47	Fwd.	D23	001	B-8	90	Dam	30.3	50

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>50%</u> Amplitude Spread Function Width (Microns)	<u>A.I.M. Resolution</u>
49	Fwd.	D23	008	A-12	155 All Direct- ions	Airfield	14.7	61
49A	Fwd.	D23	008	A-12	155	Airfield	15.5	62
50	Aft.	D56	038	C-11	120	Airfield	10.0	63
50A	Aft.	D56	038	C-11	120	Airfield	11.3	77
51	Aft.	D56	042	B-7	80	Airfield	11.2	79
51A	Aft.	D56	042	B-7	80	Airfield	8.7	103

## SECTION IX

### OBSERVED DATA

Study of the engineering passes at A/P revealed that the performance of both panoramic cameras was excellent and stable. Differences between the master and slave camera appeared to be minor.

Study of the photographs also showed that the system was imaging small enough details of industrial complexes, oil fields, airfields, and transportation networks to conclude that the intelligence needs for search and surveillance would be satisfactorily met.

Detailed examination of individual frames, resulted in identification of automobiles in the parking lot of an industrial plant and identification of isolated automobiles possible. Small private aircraft were visible on air-fields but could not be identified. Military fighter, bombers, and transport aircraft could be identified. The catwalk and pipes in the primary clarifier of a large water treatment plant were visible. Compared to a known local installation, similar in configuration, these distribution arms are 60" wide. No observable smear nor focus anomalies were detected during this detailed analysis.

In order to evaluate the actual resolution capability of the panoramic cameras on Mission 1006 (J9), all engineering passes were searched to find targets under the following conditions:

1. No clouds
2. Minimum haze
3. Low IMC error
4. Near center of format
5. Objects of known characteristics with dimensions near the limit of system resolution, with discernability the criterion.

The target area selected was Carswell AFB near Fort Worth, Texas, where all of the above conditions were met on descending pass #78, frames 19 Aft, 13 Fwd, 2-1/2" West of the center fiducial mark.

**PROD-2**

**JL UNIT**

The specific conditions at this point were:

Location	32° 50'N, 97° 12'W
Solar Elevation	58.80
Solar Azimuth	92.2°
Slant Range	661911'
Photo Scale	1/332,000
Exposure Time	1/330 sec.
Resolution Limit:	
Along Track	3.9'
Cross Track	7.4'

Three B-52 bombers were parked on a concrete apron facing southeast. Across the taxi-strip, facing northwest were five additional B-52's also on a concrete pad. Due to the different orientation of the aircraft on opposite sides of the taxi strip with respect to sun position, those facing southeast cast a shadow on the ground against which the engine nacelles and fuel pods were clearly discernable in the aft-looking format. The forward looking format showed no such shadow because of the different "look-angle" and here the difference in contrast between the aircraft and the sunlit concrete was so low that the fuel pods could not be seen, though a slight vestige of the engine nacelles remained. (Perhaps only because they are known to be there.) In a like manner the aircraft facing northwest, being obliquely lighted from a front quarter, cast their shadows further back under the wing and these nacelles and fuel pods again were contrasted against bright concrete. The pods could not be seen in either fore or aft photography. The width across these pods is 3 - 4 feet and the engine nacelles is 6 - 8 feet as determined by scaling a silhouette from an observer's handbook.

A taxi strip running East - West with blacktop pads on both sides contained 10 C-119's. 5 were parked facing north, and 5 facing south. Against this dark surface the engine nacelles were visible on all aircraft, and the tail booms could be seen for nearly their entire length, even though the booms taper to about 2 feet at the tail. In this respect, the aft view appeared to have slightly better contrast than the forward.

**GRADY**

The C-119 tail boom dimension is considerably smaller than the calculated cross track object size limit of 3.7 feet. It must be remembered that the resolution limit is based on a standard three bar target where the bars are separated by an equal width space. If objects have adequate contrast and are widely separated, as in the case of the booms, they can still be detected. The smear will reduce the apparent object contrast but will not obliterate the object until the contrast is reduced to the limit of detection.

GRADE

**SECTION X**

**MISSION 1006-1 STELLAR-INDEX CAMERA**

**A. COMPONENT ASSIGNMENT**

<u>Component</u>	<u>Serial Number</u>
Camera	D 45
Index Reseau	47
Stellar Reseau	45

**B. CAMERA DATA AND FLIGHT SETTINGS**

**Stellar Camera:**

Lens	85mm f/1.8
Exposure Time	2 seconds
Filter Type	none
Film Type	Eastman 4401

**Index Camera:**

Lens	38mm f/4.5
Exposure Time	1/500 second
Filter Type	Wratten 21
Film Type	Eastman 4400

**C. PREFLIGHT TESTING**

Stellar-Index camera D45/47/45 operated normally throughout all test phases at A/P and VAFB. Corona discharge fogging was within specifications.

**D. INFLIGHT DATA**

No anomalies were observed on telemetry records during the mission.

## E. POSTFLIGHT EVALUATION

The Stellar camera shutter malfunctioned during a significant portion of the mission. There were 412 frames of photography acquired of which only 212 were fully acceptable. The shutter failed closed during 36 frames, failed open 37 times resulting in overexposure in 74 frames and produced double imagery on 90 frames.

The double stellar imagery observed suggested double shutter operation in a frame. The microdensitometer trace, in Figure 40, page 141, shows the density of double recorded stellar imagery obtained in stellar frame #140 and displacement between imagery. The sensitometric curve for Eastman 4401 emulsion in Figure 41, page ..., was used in conjunction with star microdensities to obtain shutter open times for the double exposure for frame #140 of 1.3 and 1.5 seconds.

The time laps between double imagery recorded in frame #140 is estimated to be approximately 4 seconds. This estimate is based on a stellar shutter failure mode observed at A/P wherein the shutter abnormally opened and closed during shutter cock followed by a time lapse proportional to the V/h setting of the master instrument before final shutter open and close.

Stellar quality was adequate to meet program objectives for approximately 357 out of a total of 412 frames exposed. Star imagery was recorded throughout 64% of the available stellar format area. Star imagery was denied in 22% of the available format area due to baffle flare light fog and in 16% of the format area due to baffle vignetting. Extension of the black paint pattern on the space structure outside the stellar camera of future flights is expected to reduce baffle flare problems.

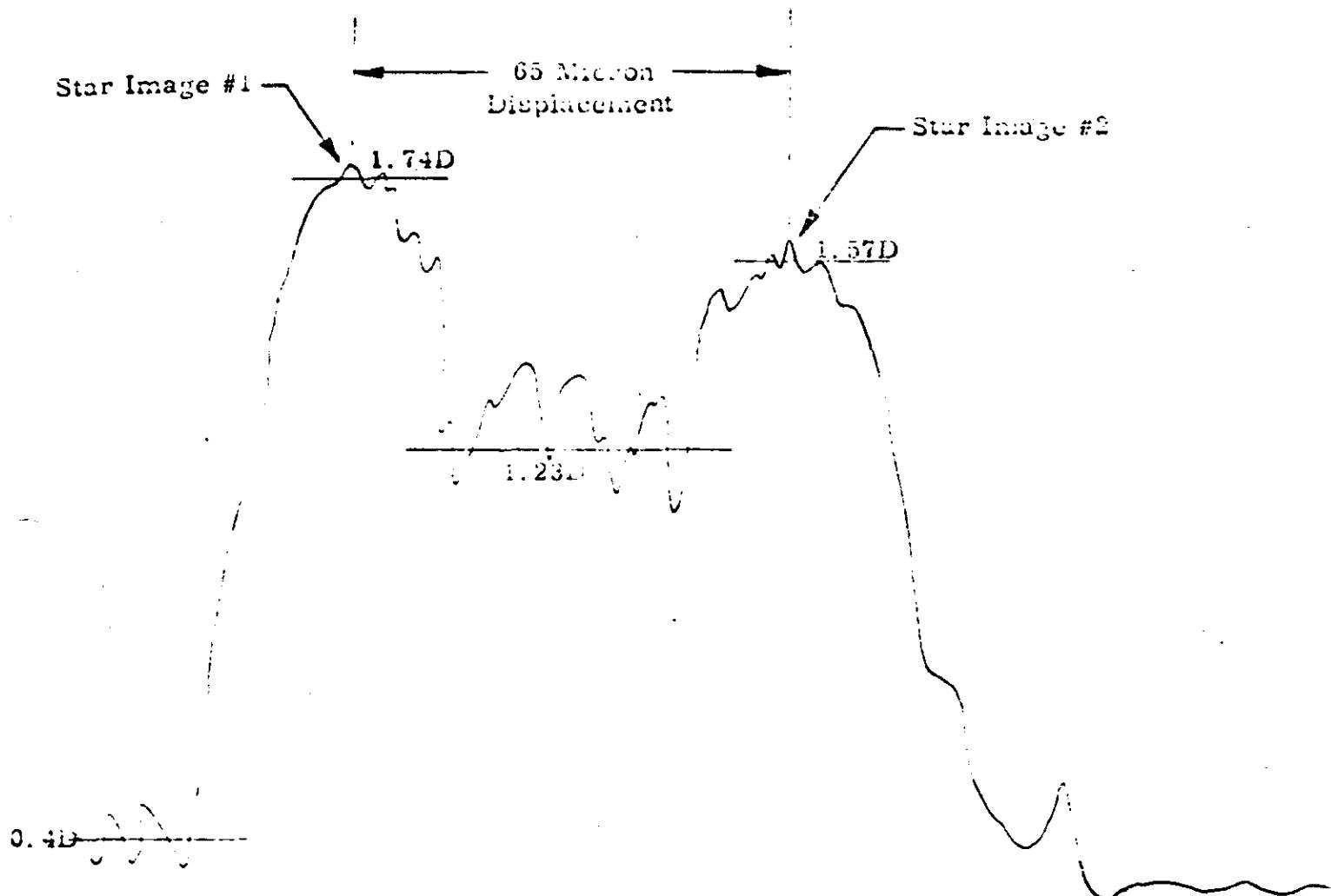
The stellar film was free of light leaks. Electrostatic corona was present along both film edges but did not degrade imagery.

The correlation lamp was operational throughout mission. Three out of four fiducial crosses were present in all flight photography. The fourth fiducial was excessively bloomed with resulting loss of the fiducial cross. The bloomed fiducial is attributed to the loss of the neutral density gelatin filter which apparently fell off the lamp housing during vehicle ascent, due to ascent vibration.

It appears that the stellar baffle was deployed into a normal flight configuration. Baffle flare and vignetting is described under "stellar quality" above. Figure 42, page ..., shows typical stellar format densities resulting from suspected baffle flare light.

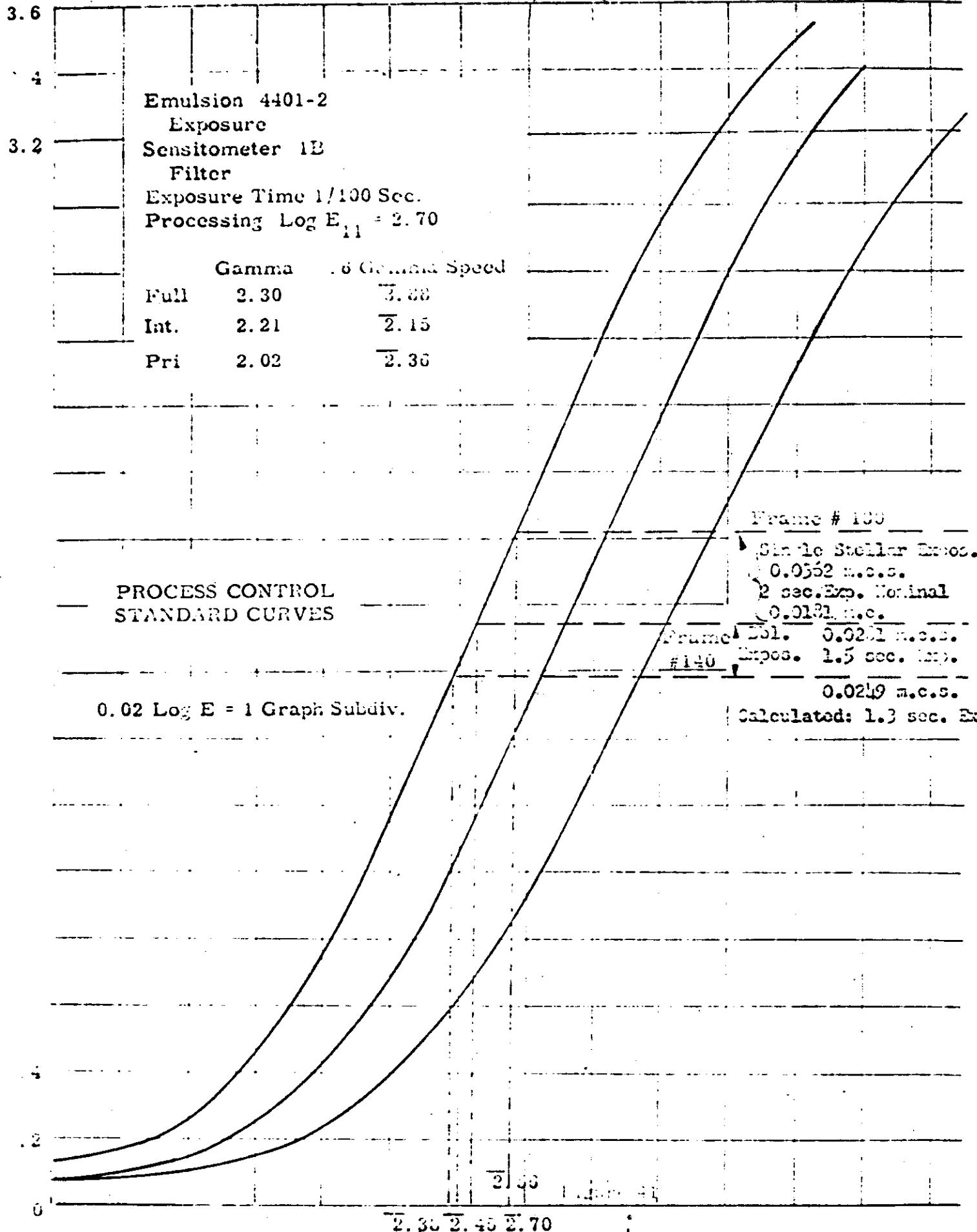
GRADE

**MICRODENSITOMETER TRACE SHOWING DOUBLE STELLAR IMAGE**  
Mission 1006-1, S/I #D-45, Frame #140



**GRADE**  
McGraw-Hill

**CLUMET**

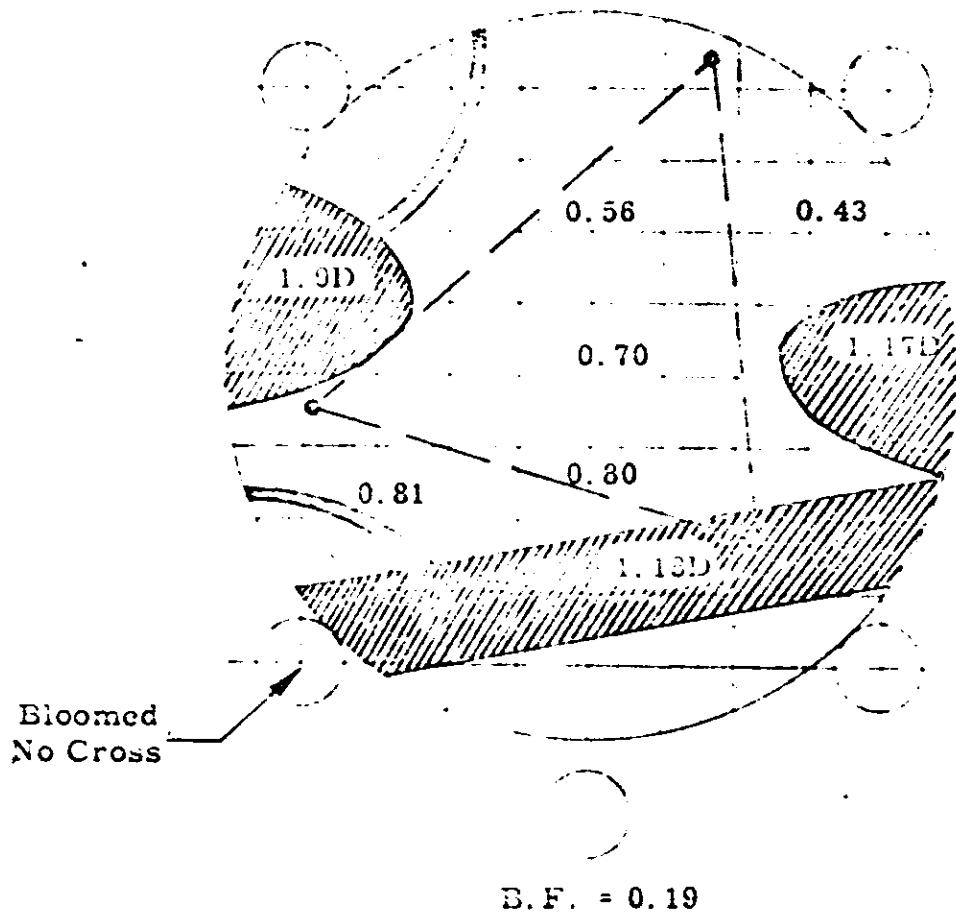


**ERADIX**

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MISSION 1006-1 (J-8)

Normal Stellar Imagery Showing Stars Recorded and Flare Density  
23 Stars Recorded at Beginning of Mission  
36 Stars Recorded at End of Mission



S/I D-45  
2 Sec. Exposure Nominal

Note: 90 Exposures Contained Double Imagery With 65 Microns Separation

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391 out of 392 frames of Index Camera photography acquired were good with no anomalies present. The first frame of the mission was double exposed.

The Index Camera ran out of film after frame 392 resulting in 20 frames of stellar photography without corresponding index frames. Post flight measurements show that a total of 82.5 feet of index film was on the supply spool which is 2.4 feet shorter than planned. An additional 4 feet was required to complete the mission program.

Evaluation of duplicate positive film at A/P showed that cloud cover ranged from 10% to 20% on some engineering passes to about 80% on others. Large geographic features of the South Platt River Basin and Western Montana mountains were easily discernible even though most frames were of low contrast. Visual observation at 10x showed grain.

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**SECTION XI**

**MISSION 1006-2 STELLAR-INDEX CAMERA**

**A. COMPONENT ASSIGNMENT**

<u>Component</u>	<u>Serial Number</u>
Camera	D 49
Index Reseau	53
Stellar Reseau	42

**B. CAMERA DATA AND FLIGHT SETTINGS**

**Stellar Camera:**

Lens	85mm f/1.8
Exposure Time	2 seconds
Filter Type	none
Film Type	Eastman 4401

**Index Camera:**

Lens	38mm f/4.5
Exposure Time	1/500 second
Filter Type	Wratten 21
Film Type	Eastman 4400

**C. PREFLIGHT TESTING**

Stellar-Index camera D49/53/42 functioned properly through all testing plans at A/P and VAFB. During environmental testing the film take-up failed resulting in film wrapping around the camera metering roller. Examination showed no damage to the camera. The take-up unit was replaced.

**D. INFLIGHT DATA**

The telemetry records showed the intermittent failure of the Master Camera center-of-format switch which resulted in the loss of function switching to the S/I Camera stepper switch.

**GRADY**

## **E. POSTFLIGHT EVALUATION**

The Stellar and Index Cameras functioned properly during the mission until the failure of the Master Camera center-of-format switch at the start of pass D103. Five frames were produced after the switch failure.

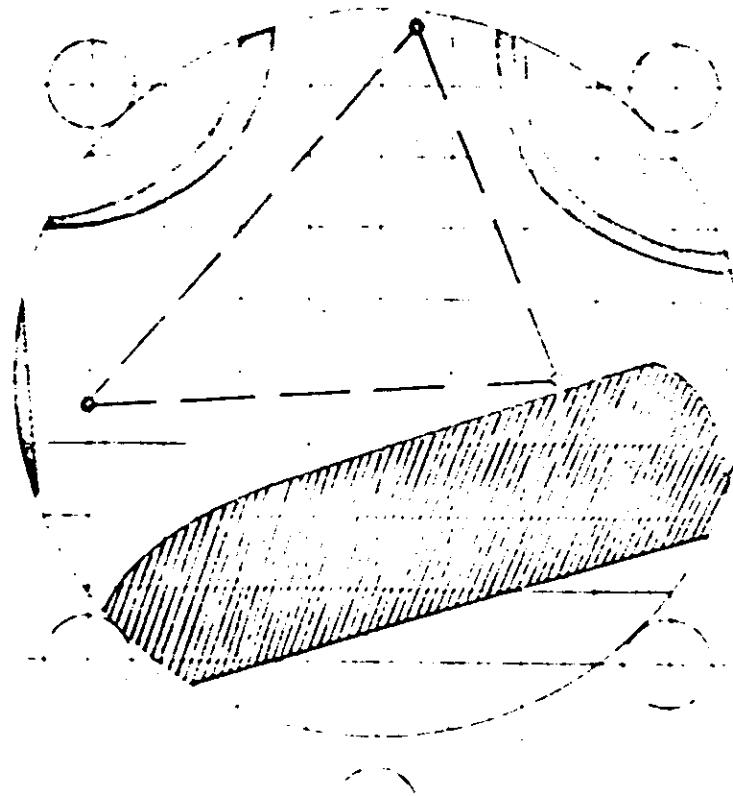
Minor light streaking was observed in Stellar Camera frames 268, 269, 279 and 273. The typical format is shown in Figure 43, page 131.

Evaluation of the Index Camera duplicate positive engineering film at A/P showed that cloud cover ranged from about 25% to 80% toward the latter part of the mission. Geographic features of the Dallas-Fort Worth area, at low contrast, were clear to the naked eye. Runways on airfields were clear at 5x magnification, and the Denver area was visible through scattered clouds. 10x magnification revealed street and freeway patterns and other details despite slight haze.

**GRADE**

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~~REF ID: A6511~~  
MISSION 1006-2 (J-9)

Typical Stellar Imagery  
25 to 30 Stars Recorded Each Exposure



S/I D-49  
2 Sec. Exposure Nominal

~~SECRET~~

## SECTION XII

### MISSION 1006-1 RECOVERY SUBSYSTEM

#### A. COMPONENT ASSIGNMENT

<u>Component</u>	<u>Serial Number</u>
Recovery Subsystem	638
Take-Up Cassette	T-21/TJ-14

#### B. PREFLIGHT TESTING

SRV 638 was received at A/P on August 23, 1963, weighing in at 155 pounds. Normal disassembly was completed prior to A/P modifications. After modifications and E. O. incorporations testing was completed for SRV 638 integration to the J9 system.

The following items are noteworthy of being above the normal testing required and the rework/mods required thus effecting additional A/P effort in rework and acceptance testing.

<u>Date</u>	<u>Item</u>	<u>Discrepancy</u>	<u>Action</u>
3-10-64	Haydon timer/ arm ckt.	Fuse blew in GSE in arm circuitry	Reworked arm ckt. ground in SRV
4-17-64	Haydon timer	Failed Post TASC	Replaced same
4-23-64	Thrust cone	Torn light leak shield	Replaced same

On April 24, 1964, all A/P testing per procedures J13xxxx, installations, modifications, alignments, and cleanup were completed. Then SRV 638 was shipped to VAFB on April 26, 1964.

VAFB pre pad installations and testing functions were completed May 19, 1964. The Mission 1006, Vehicle 1176, SRV Systems pad run was completed successfully May 22, 1964.

After the pad run additional flight preparations were completed on May 24, 1964. Then the entire J9 payload left for the launch pad #1, PALC, May 27, 1964. Launch occurred 1600 June 4, 1964.

GEORGE

### **C. INFLIGHT DATA**

Sequence of events, commands, temperatures and other payload functions are discussed in Section II.

### **D. POSTFLIGHT EVALUATION**

SRV 638 was recovered in the air June 8, 1964, after the 65th orbit. It was transported by air from Hawaii and arrived at A. P 0650, June 9, 1964 for film retrieval. It weighed in at 174.5#. Retrieval operations proceeded normally. Weight of the recovered cassette #1 plus film was 70 pounds. Weight of the cassette #2 plus film was 69 pounds. The cassettes with film were packaged for shipment and turned over to A/P security for shipment to the customer.

SRV post flight testing was per A/P procedure #5110. The beacon, recovery programmer, and telemetry were within specification. Film footage pots approximated the final flight TM readings of 4.0 volts.

Various components of this SRV could possibly be used for flight again. However, this will be at the direction of the customer.

**SECRET**

## SECTION XIII

## MISSION 1006-2 RECOVERY SUBSYSTEM

## A. COMPONENT ASSIGNMENT

<u>Component</u>	<u>Serial Number</u>
Recovery Subsystem	639
Take-Up Cassette	T-30/TJ-30

## B. PREFLIGHT TESTING

SRV 639 was received at A/P September 5, 1963, weighing in at 152 pounds. Normal disassembly was completed prior to A/P modifications. After modifications and E.O. incorporations, testing was completed for SRV 639 integration to the J9 system.

The following items are noteworthy of being above the normal testing required and the rework/mods required, thus effecting additional A/P effort in rework and acceptance testing.

<u>Date</u>	<u>Item</u>	<u>Discrepancy</u>	<u>Action</u>
4-21-64	Forebody	Outer shell had small separation from inner shell	Replaced a dowel and sealed
4-21-64	W4 Harness	Shield tie pin loose	Replaced W4 harness
5-13-64	Forebody	Shorted to thrust cone	Forebody filed to isolate contact with T

On April 21, 1964, all A/P testing per procedures J13xxxx, installations, modifications, alignments, and cleanups were completed. Then SRV 639 was shipped to VAFB on April 26, 1964.

VAFB pre pad installations and testing functions were completed May 19, 1964. The Mission 1006, Vehicle 1176, SRV Systems pad run was completed successfully on May 22, 1964.

After the pad run additional flight preparations were completed on May 24, 1964. Then the entire J9 payload left for the launch pad #1, PALC on

**SECURITY**

[REDACTED]

May 27, 1964. On June 1, 1964 at the launch pad, the beacon came on during launch checks. The payload was demated from the vehicle and shook, and retested. Interface connectors were cleaned and rechecked. Then the payload was remated and launch activities continued. Launch occurred 1600 June 4, 1964.

#### C. INFLIGHT DATA

Sequence of events, commands, temperatures and other payload functions are discussed in Section II.

#### D. POSTFLIGHT EVALUATION

SRV 639 was recovered in the air June 12, 1964, after the 128th orbit. It was transported by air from Hawaii and arrived at A/P 0500, June 13, 1964 for film retrieval. It weighed in at 176.5 pounds. Retrieval operations proceeded normally. Weight of the recovered cassette #1 plus film was 69.8 pounds. Weight of the cassette #2 plus film was 70 pounds. There was 8 feet of unwrapped slave camera film inside the capsule. It was wrinkled back four feet. The tag end was badly torn. The cassettes, with film, were packaged for shipment and turned over to A/P security for shipment to the customer.

SRV Post Flight testing was per A/P procedure #5110. The beacon, recovery programmer, and telemetry were within specifications.

Various components of this SRV would possibly be used for flight again. However, this will be at the direction of the customer.

**SECRET**

[REDACTED]

## SECTION XIV

### VEHICLE ATTITUDE CONTROL

The vehicle attitude control errors are obtained from the Horizon Camera and Stellar Camera photographs and are sent to A/P on the Frame Correlation magnetic tape. The attitude error rates are calculated on the A/P computer. Frequency distribution graphs are plotted by the computer. The graphs for Mission 1006-1 attitude control errors and rates are on Figures 44 to 47 on pages 10 to 13, and for Mission 1006-2 on Figures 50 to 53 on pages 14 to 17.

The summary below of the attitude errors and rates shows the maximum value that occurred during 90% of the photographic operations and the range of values during all photography. The predicted 90% values of attitude errors and rates are also listed.

		MISSION 1006-1		MISSION 1006-2	
	<u>Predicted</u>	<u>90%</u>	<u>Range</u>	<u>90%</u>	<u>Range</u>
<b>Errors (<sup>o</sup>)</b>					
Pitch	1.0	0.41	-0.39/+0.45	0.49	-0.75/+2.00
Roll	0.5	0.42	-0.92/+0.38	0.40	-0.80/+0.64
Yaw	1.1	1.14	-1.69/+0.27	1.08	-1.70/+0.30
<b>Rates (<sup>o</sup>/hr)</b>					
Pitch	30	27	-97 +86	31	-100/+100
Roll	30	28	-53 +98	28	-58/+74
Yaw	30	27	-63 +57	30	-42/+78

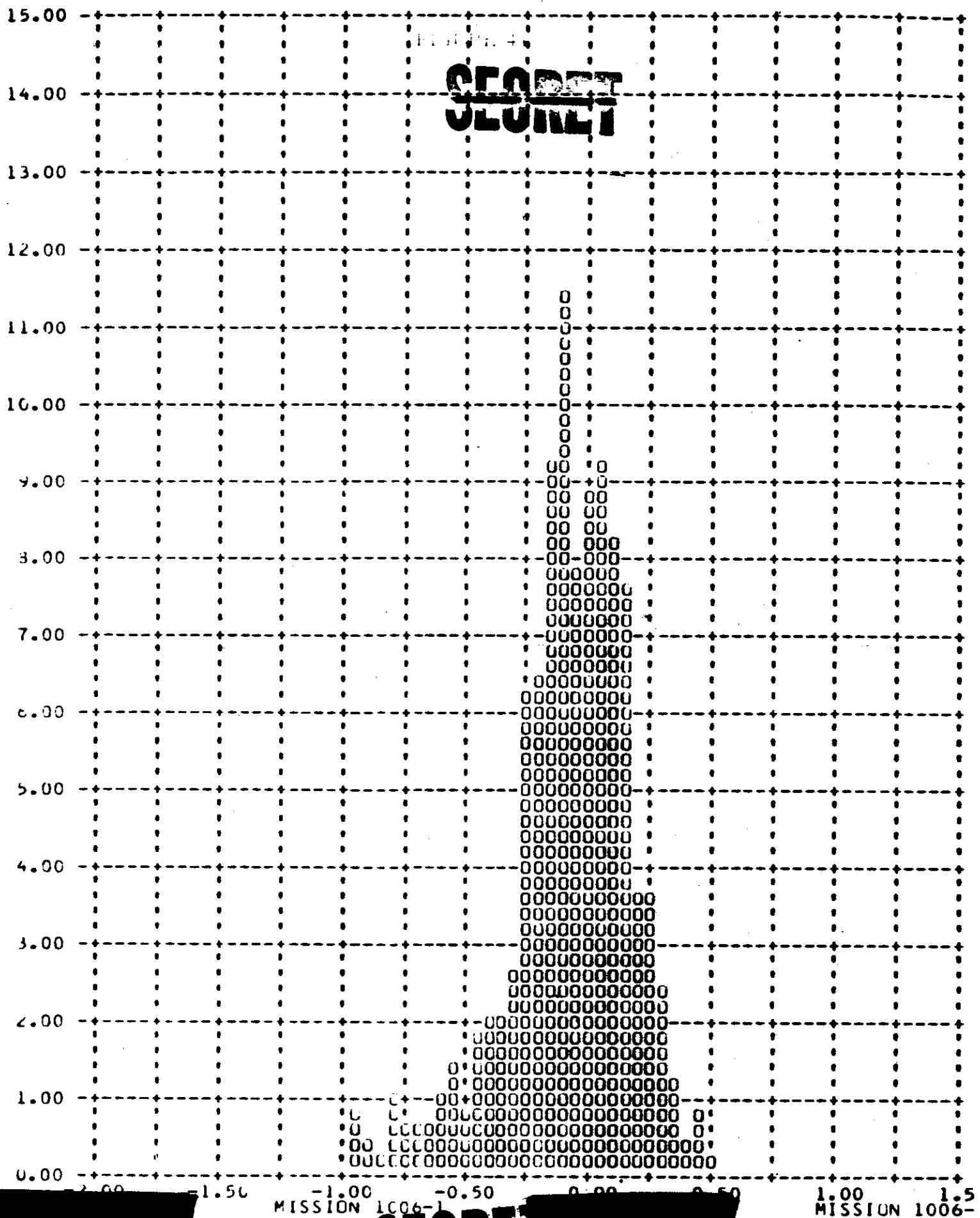
The vehicle attitude control performed as predicted and the resulting image motion was below the detectable limit of the Panoramic Camera Subsystem.

Attitude control ranges obtained from inflight telemetry data was tabulated for passes 9, 31, 47, 56, 103 and 126. The values from the attitude control horizon sensor and gyro are compared with photographic values in Table II, page .

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J-09 A BUCKET 9-14-64 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 0.41

Y PITCH ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (Y)



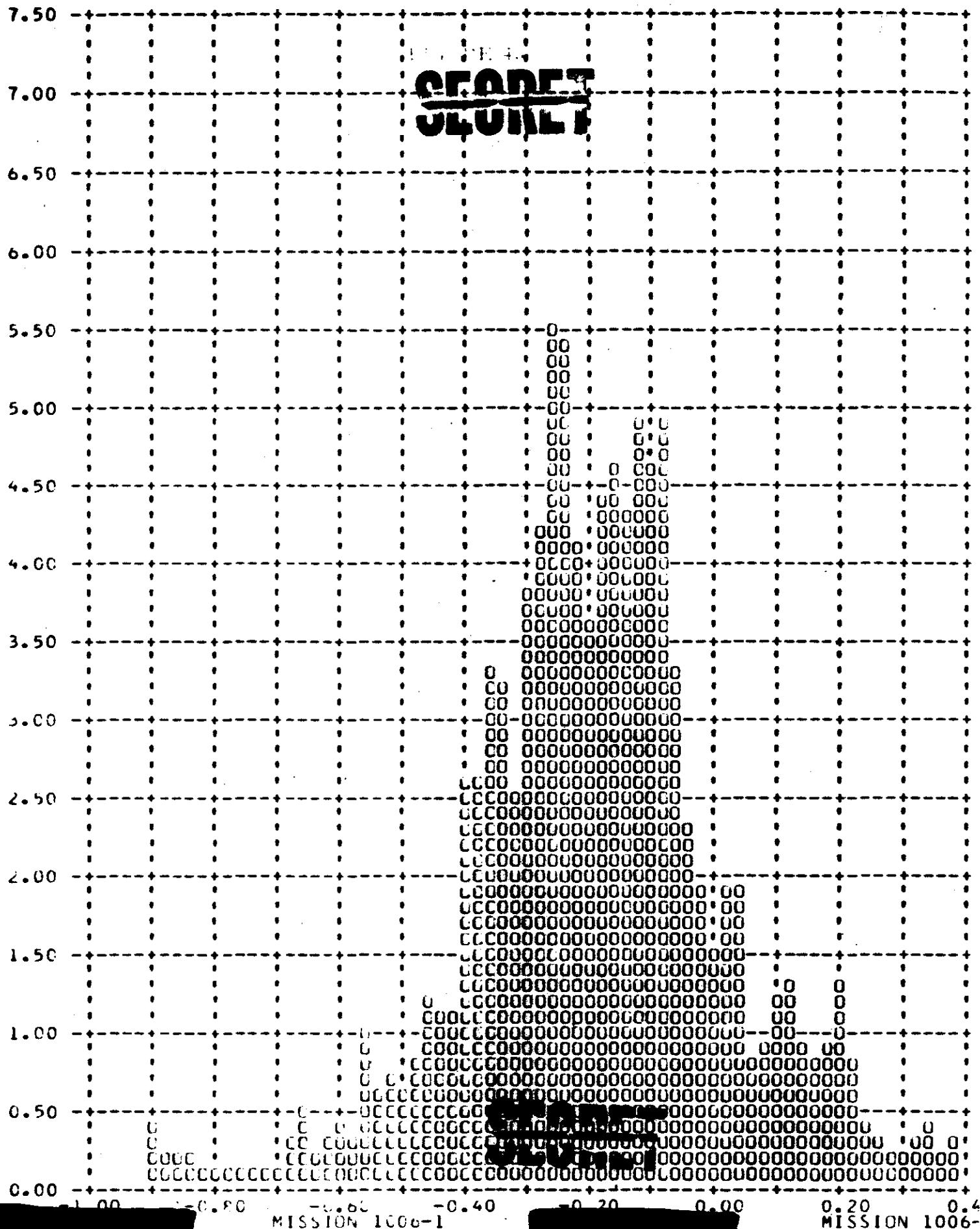
J-09 A BUCKET 9-22-64

MISSION 1006-1

MISSION 1006-

FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 0.42

Y ROLL ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (Y)

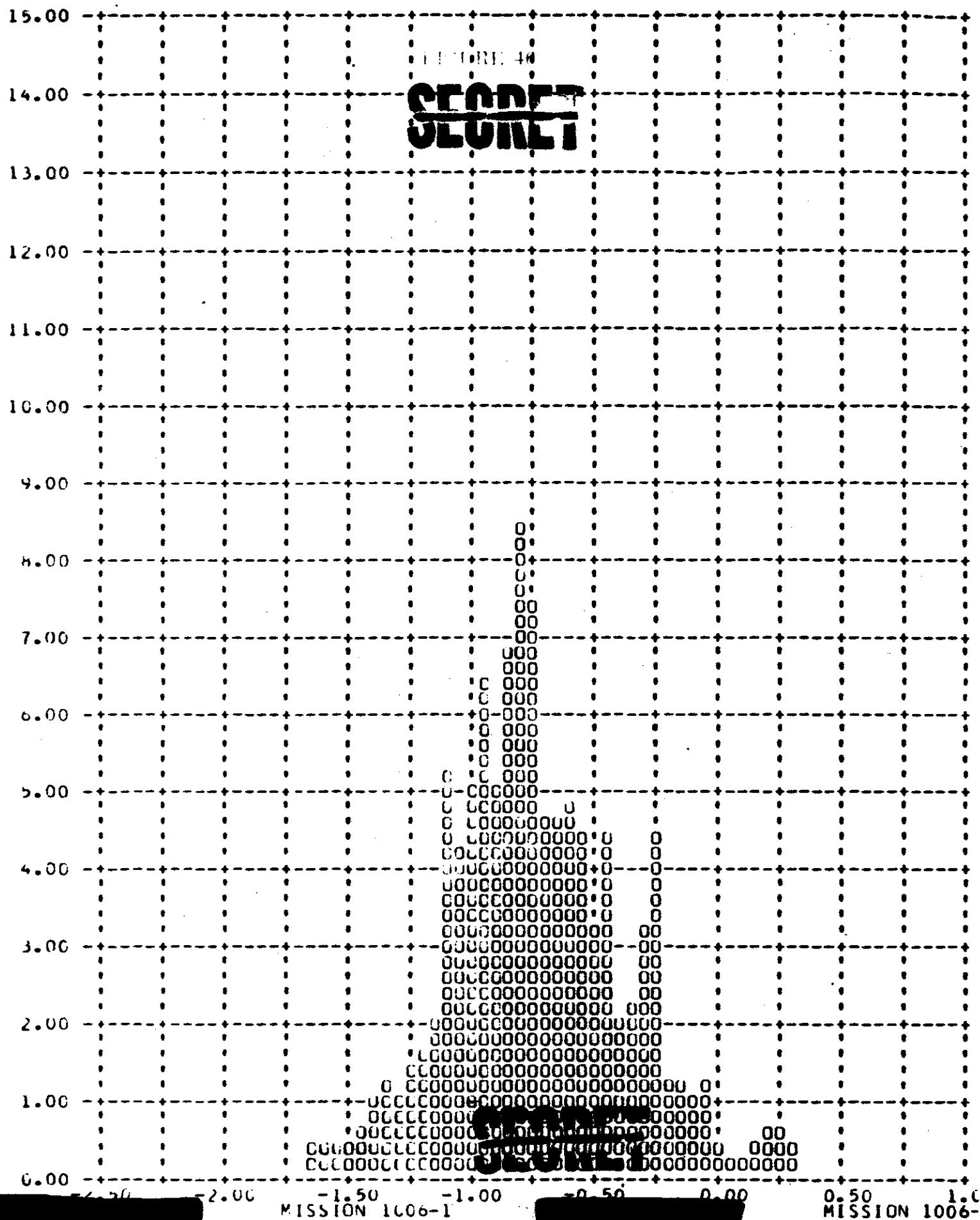


MISSION 1006-1

MISSION 1006-

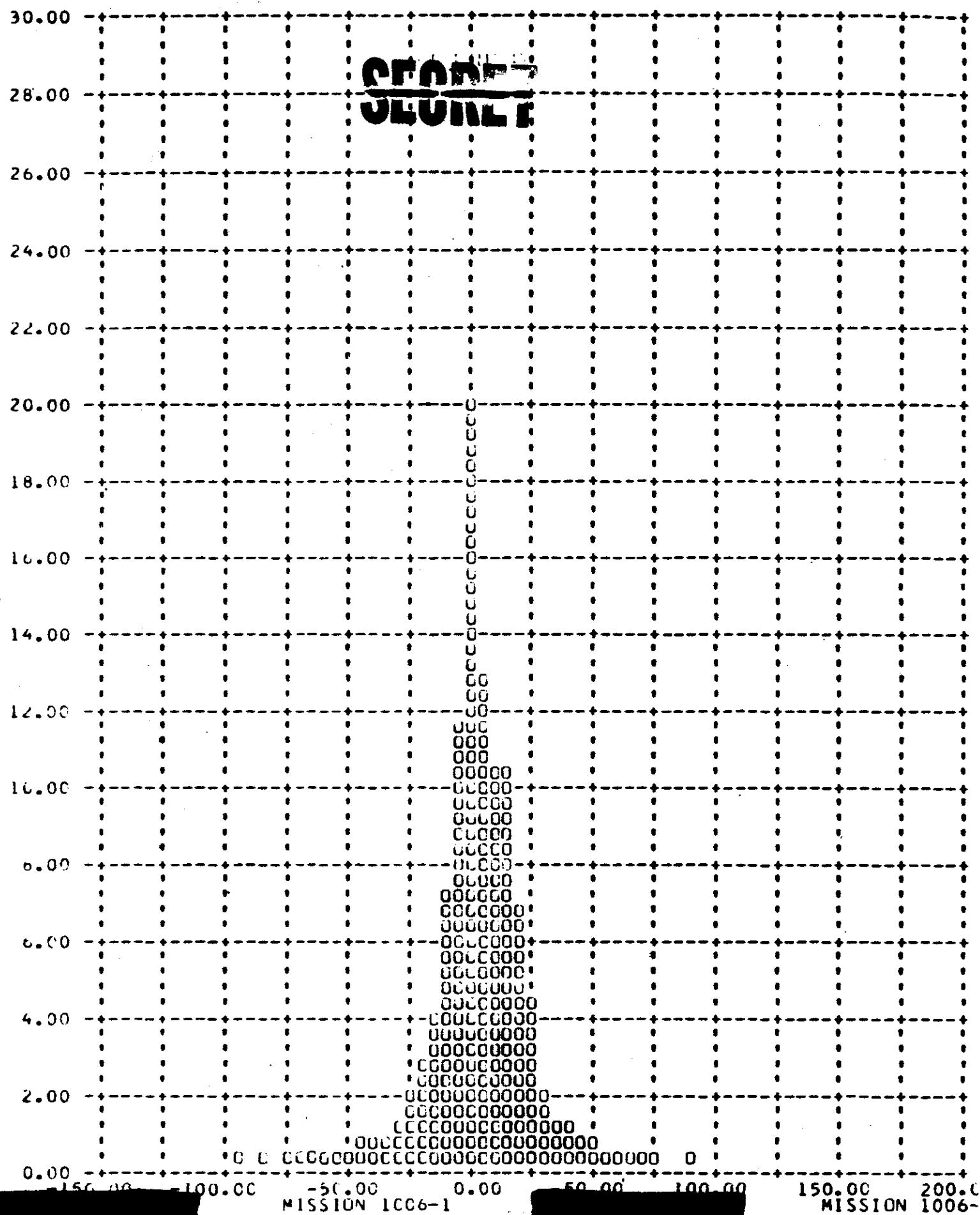
J-09 A BUCKET 9-14-64 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 1.14

Y YAW ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (Y)



J-09 A BUCKET 9-14-64 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 26.78

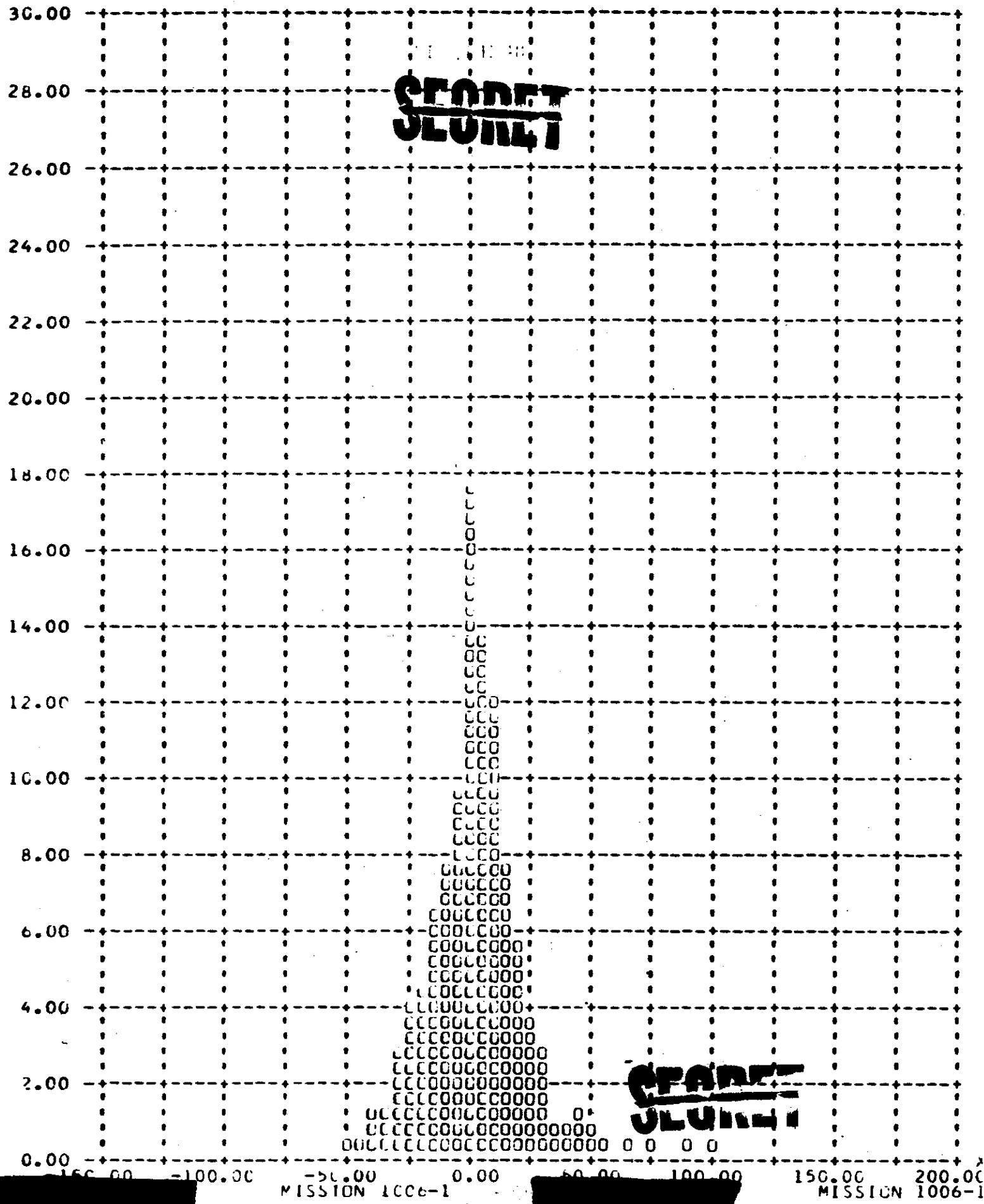
Y PITCH RATE ERRCR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)



J-09 A BUCKET 9-22-64

MISSION 1006 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 28.4%

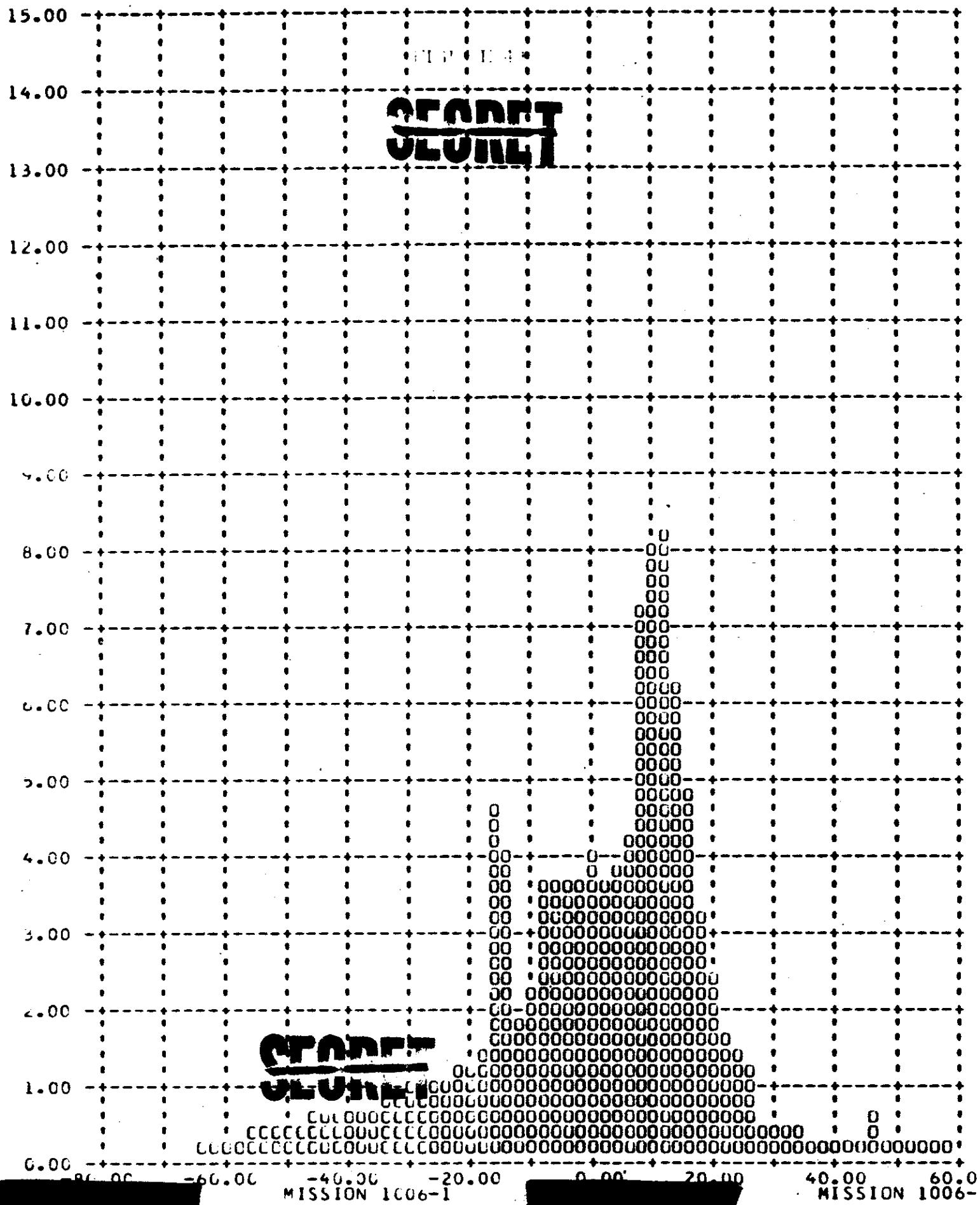
Y ROLL RATE ERRCR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)



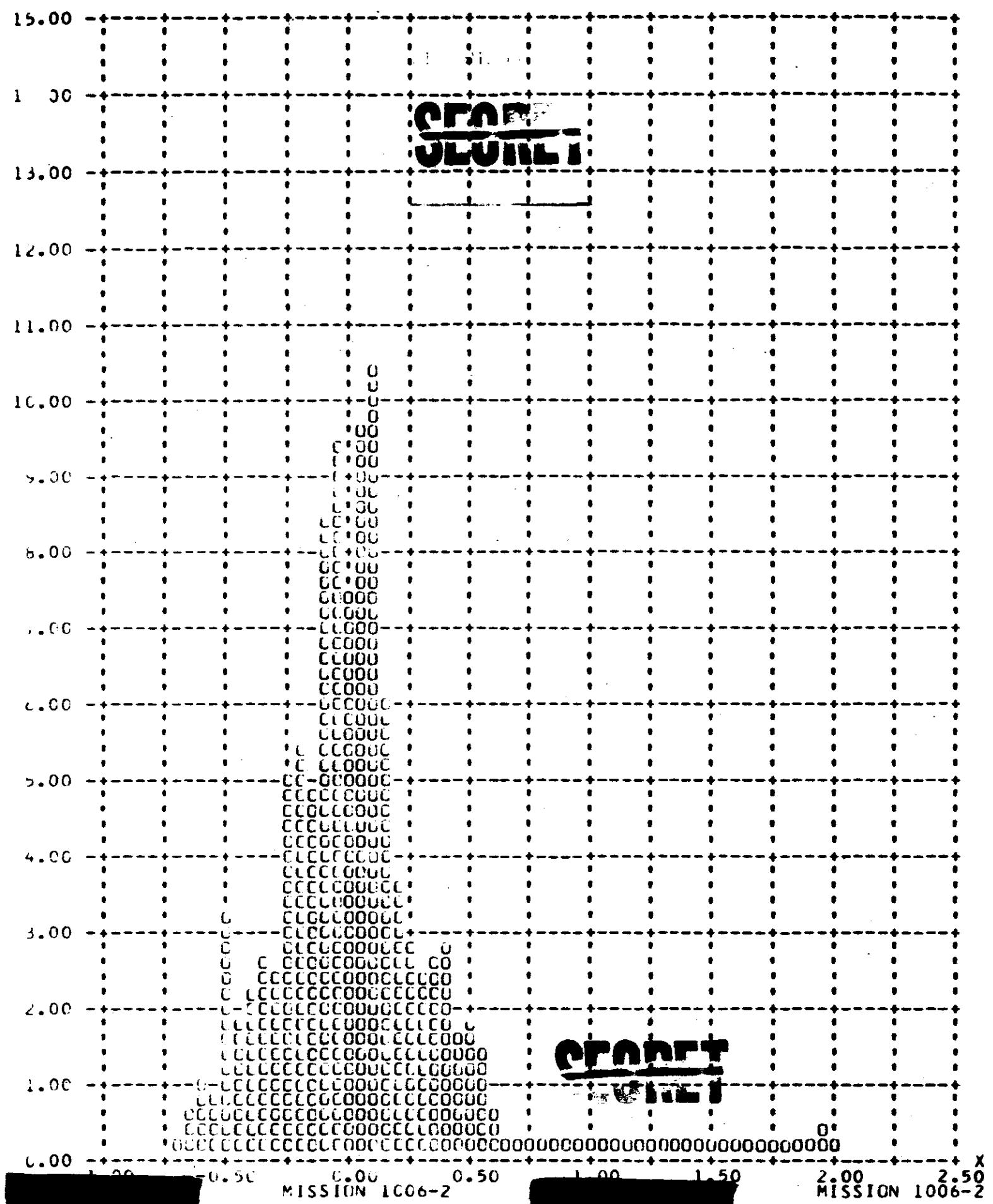
J-09 A BUCKET 9-1

~~FRAMES 1-6 OF EACH OP OMITTED~~ 90 PERCENT = 27.82

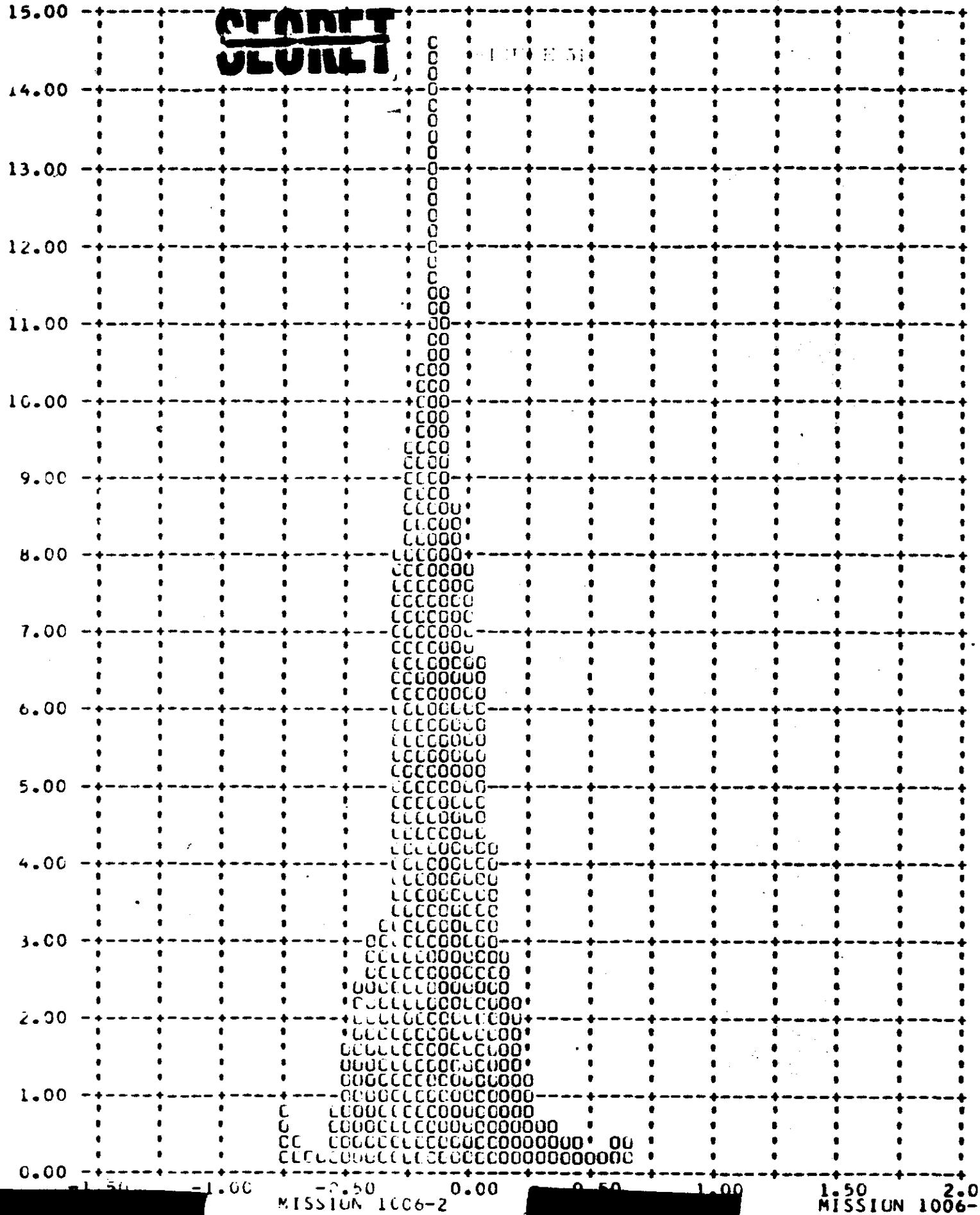
Y YAW RATE ERROR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)



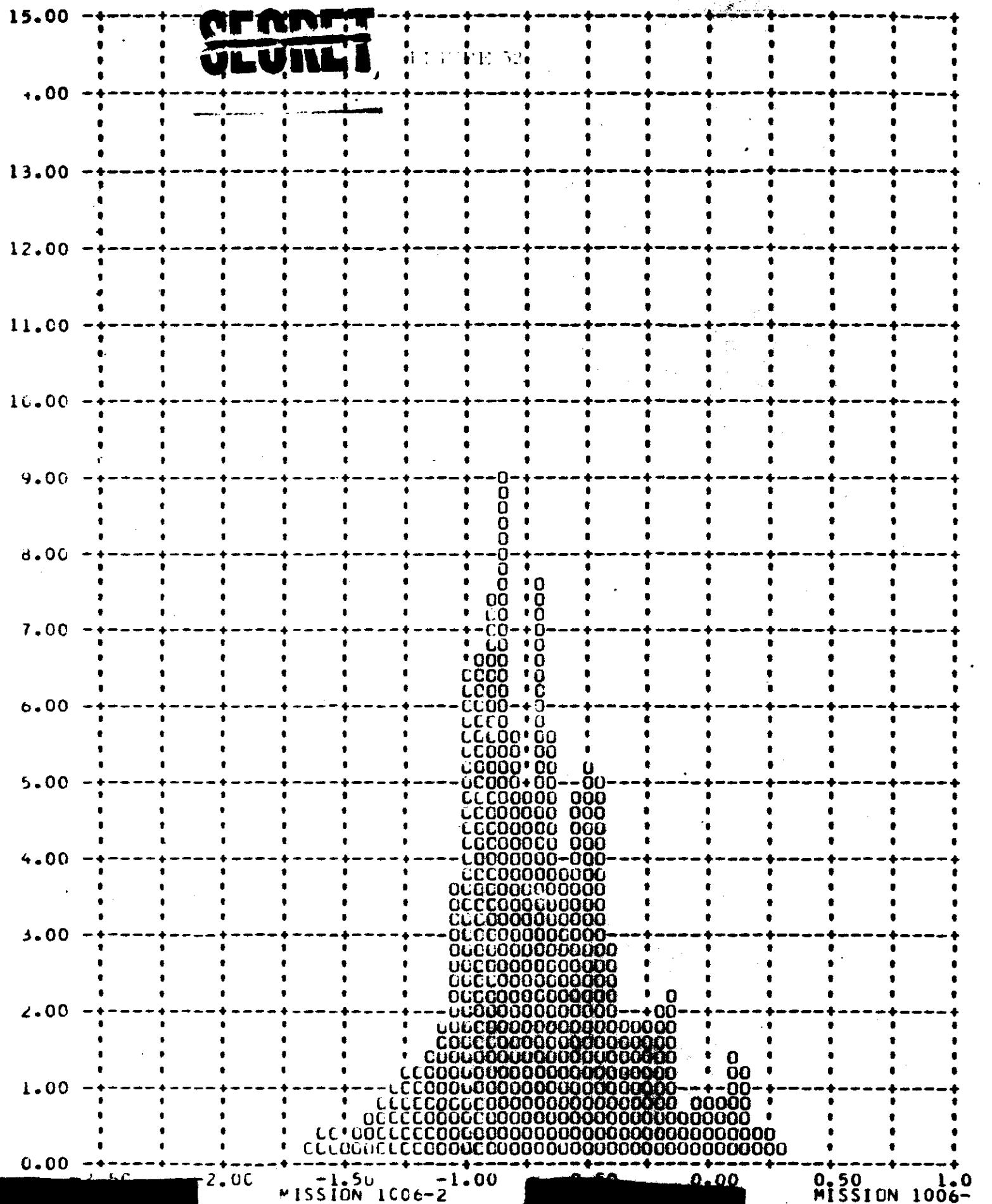
## Y PITCH ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (%)



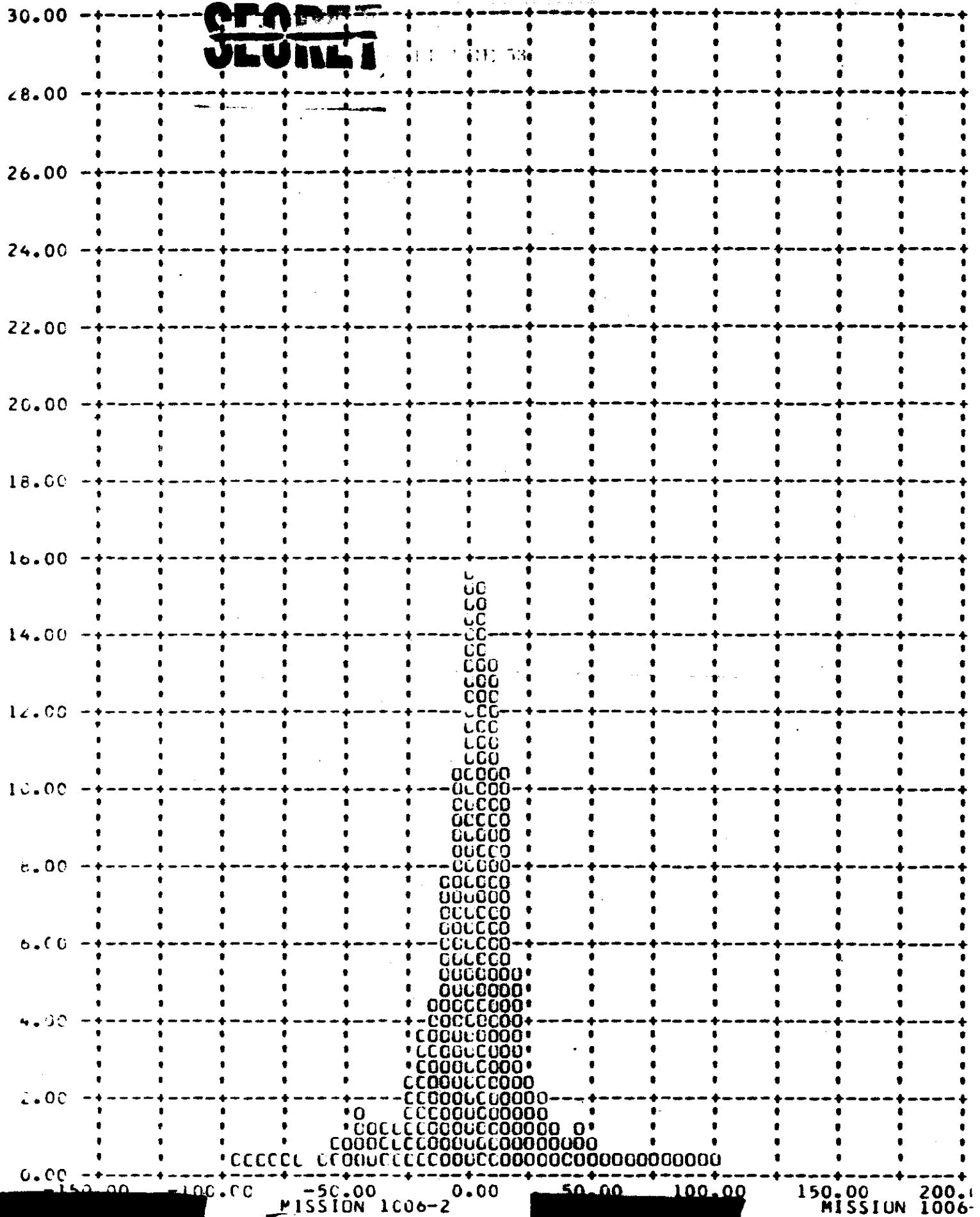
## Y ROLL ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (Y)

**GRADE**

## Y YAW ANGLE ERROR - DEGREES (X) VERSUS FREQUENCY - PERCENT (Y)

**GRANT**

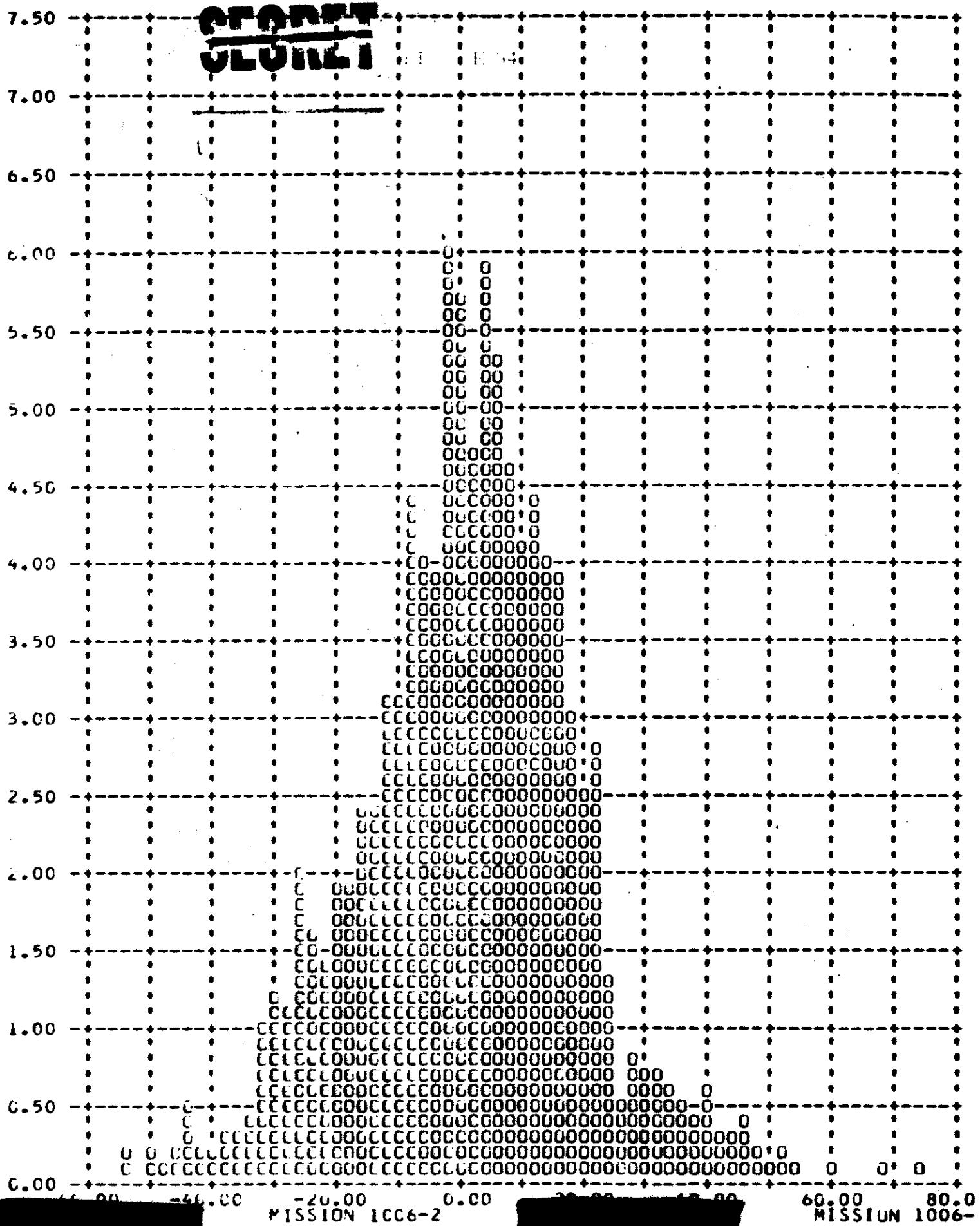
## Y - PITCH RATE ERROR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)



MISSION 1006-2

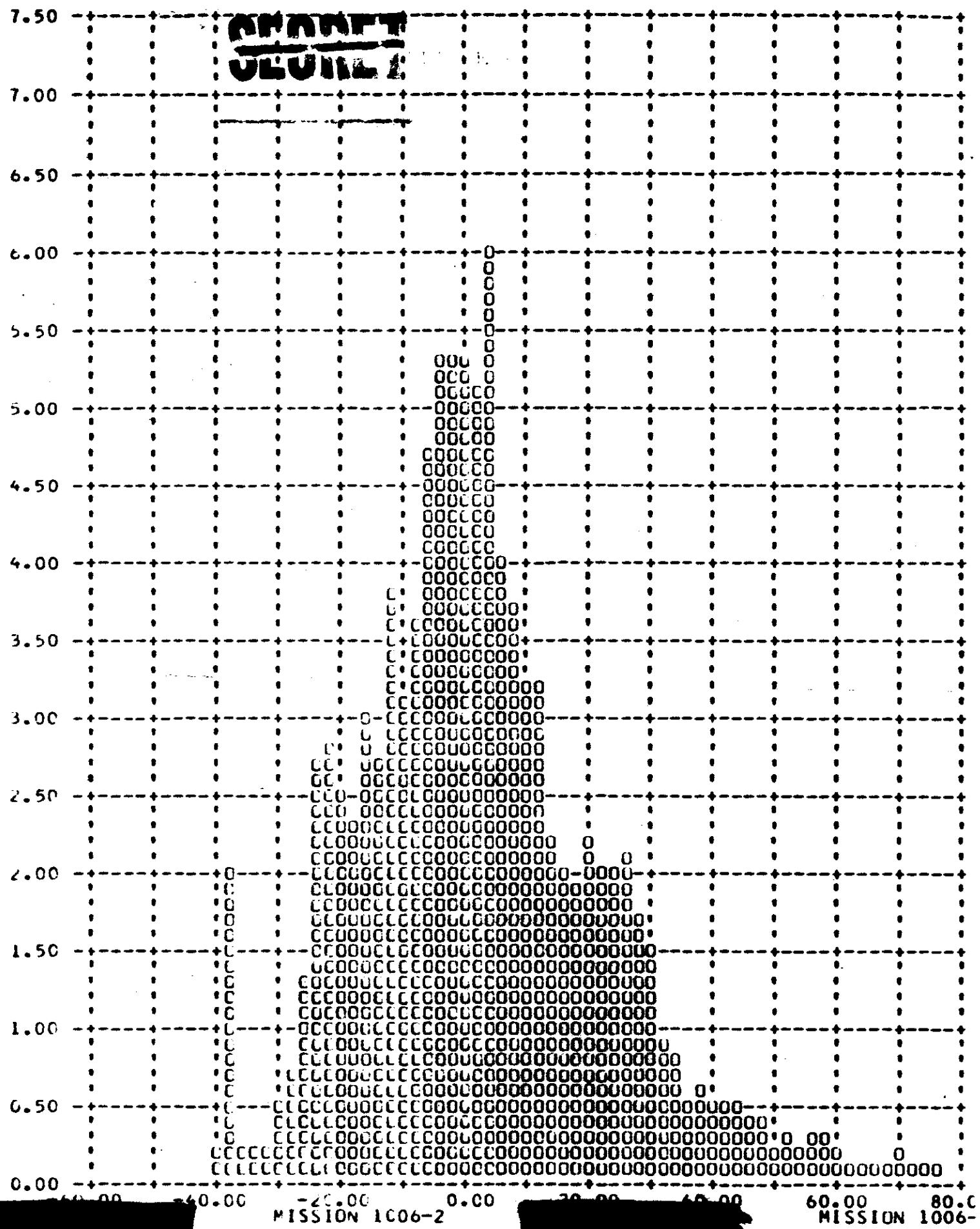
**GRANDE**

**Y ROLL RATE ERROR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)**



**GRADE**

Y YAW RATE ERRCR - DEG/HOUR (X) VERSUS FREQUENCY - PERCENT (Y)



MISSION 1C06-2



TABLE II  
ATTITUDE CONTROL DATA COMPARISON

Pass Number	9	31	47	56	103
Errors ( $^{\circ}$ )					
Pitch Sensor	+0.1/-0.5	+0.3/-0.8	-0.1/1.0	0/-0.6	0/-0.9
Gyro	+0.3/-0.1	+0.4/-0.6	+0.4/-0.6	+0.3/-0.7	+0.4/-0.5
Photo	+0.1/-0.2	-0.9/-1.0	-	-0.3/+0.2	+0.1/-0.1
Roll Sensor	+0.1/-0.3	+0.3/-0.5	+0.3/-0.2	+0.4/-0.5	+0.3/-0.4
Gyro	+0.3/-0.5	0/-0.2	0/-0.2	+0.3/-0.4	+0.2/-0.4
Photo	-0.4/-0.2	0/-0.2	-	-0.1	-0.1/-0.5
Yaw Gyro	+0.2/-0.3	+0.4/-0.6	-0.5/-0.7	+0.4/-0.4	+0.4/-0.6
Photo	-0.9	No Data	-	No Data	-1.3/-0.7
No Data					
Rates ( $^{\circ}/hr$ )					
Pitch Sensor	43	48	34	38	40
Gyro	40	44	41	32	48/40
Photo	22	45	-	35	17
Roll Sensor	41/60	42	49/74	74/42	43/50
Gyro	103/36	19	10	44	33
Photo	14	27	-	9	27
Yaw Gyro	37	61	26	22	36
Photo	2	No Data	-	No Data	30

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## SECTION XV

### IMAGE SMEAR ANALYSIS

The V/h error encountered during both missions was much larger than usual due to the low perigee. The frequency distribution of the errors for Mission 1006-1 are computer plotted in Figure 56, page 168 and for Mission 1006-2 in Figure 59, page 171. These V/h values are derived from the time word in the binary data block and compared against the ephemeral data which defines the proper V/h value.

A computer program at A/P has been developed to calculate the total image motion resulting from the various attitude errors and rates, earth rotation and the V/h errors. The resulting image motion is then used to calculate a theoretical ground resolution limit along the flight path and across the flight path, based on a 70% allowable smear factor. The resolution limit values would apply to any camera system attached to the vehicle having the same errors and using the same exposure time. The resolution limit value indicates the minimum ground resolution that can be recorded and not the actual resolution that was recorded. Normally the computed resolution limit is well below the system capability. The resolution limit frequency distribution plots for Mission 1006-1 are shown in Figures 57 and 58 on, page 172 and 173, and for Mission 1006-2 in Figures 60 and 61, page 174 and 175.

A summary of the maximum V/h errors, along track and cross track resolution limits that occurred during 90% of the photographic operations is shown below.

	1006-1	1006-2
V/h Error	15.4%	11.6%
Along Track Resolution Limit	13.8'	10.1'
Cross Track Resolution Limit	6.7'	7.0'

It is apparent that some degradation occurred in the information content of the photography however five foot objects, ten foot ground resolution, were within the system limits for the majority of Mission 1006-1 and 90% of Mission 1006-2.

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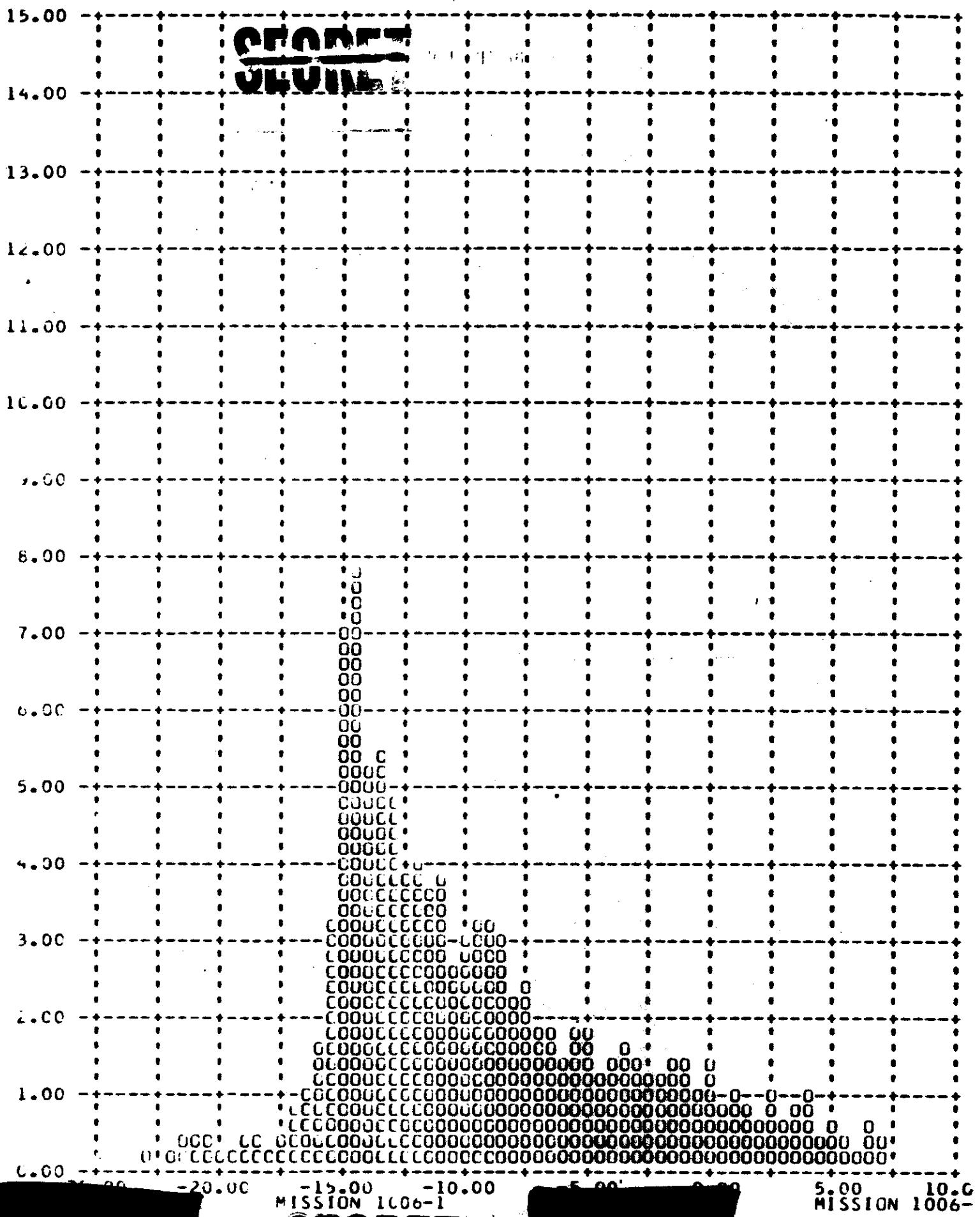
The horizon sensor and gyro data for passes 31 and 56 was obtained during camera operation whereas the remaining data was in a non-operating mode. There were no camera operations during passes 47 and 126 therefore no attitude values from photography are available.

Generally, the attitude error data from the vehicle telemetry correlates quite well with the photographic data. Far less correlation is present in the rate values particularly in the roll axis.

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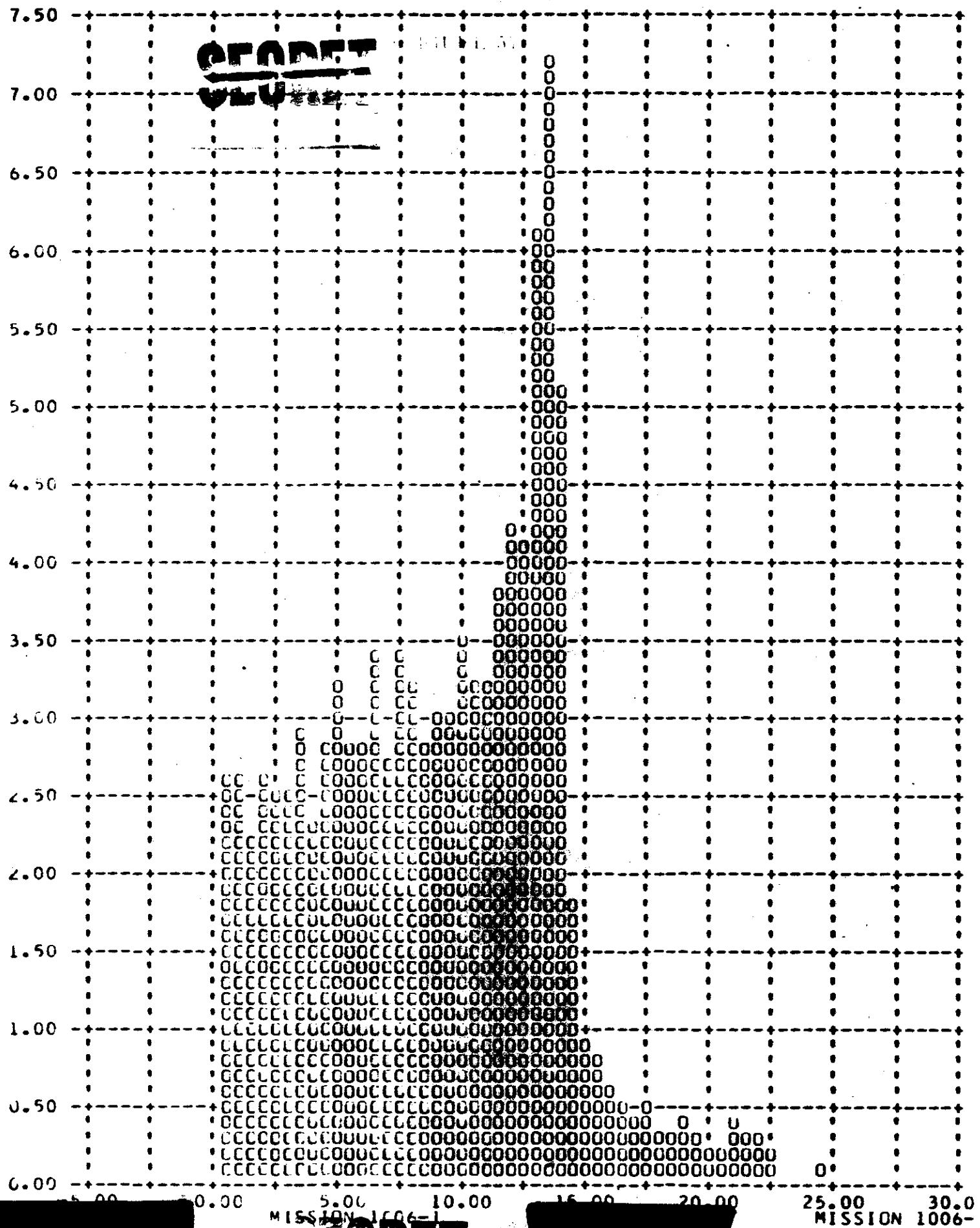
J-09 A BUCKET 9-14-64 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 15.38

Y V/H RATIO ERROR - PERCENT (X) VERSUS FREQUENCY - PERCENT (Y)



J-09 A BUCKET 9-14-64 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 13.76

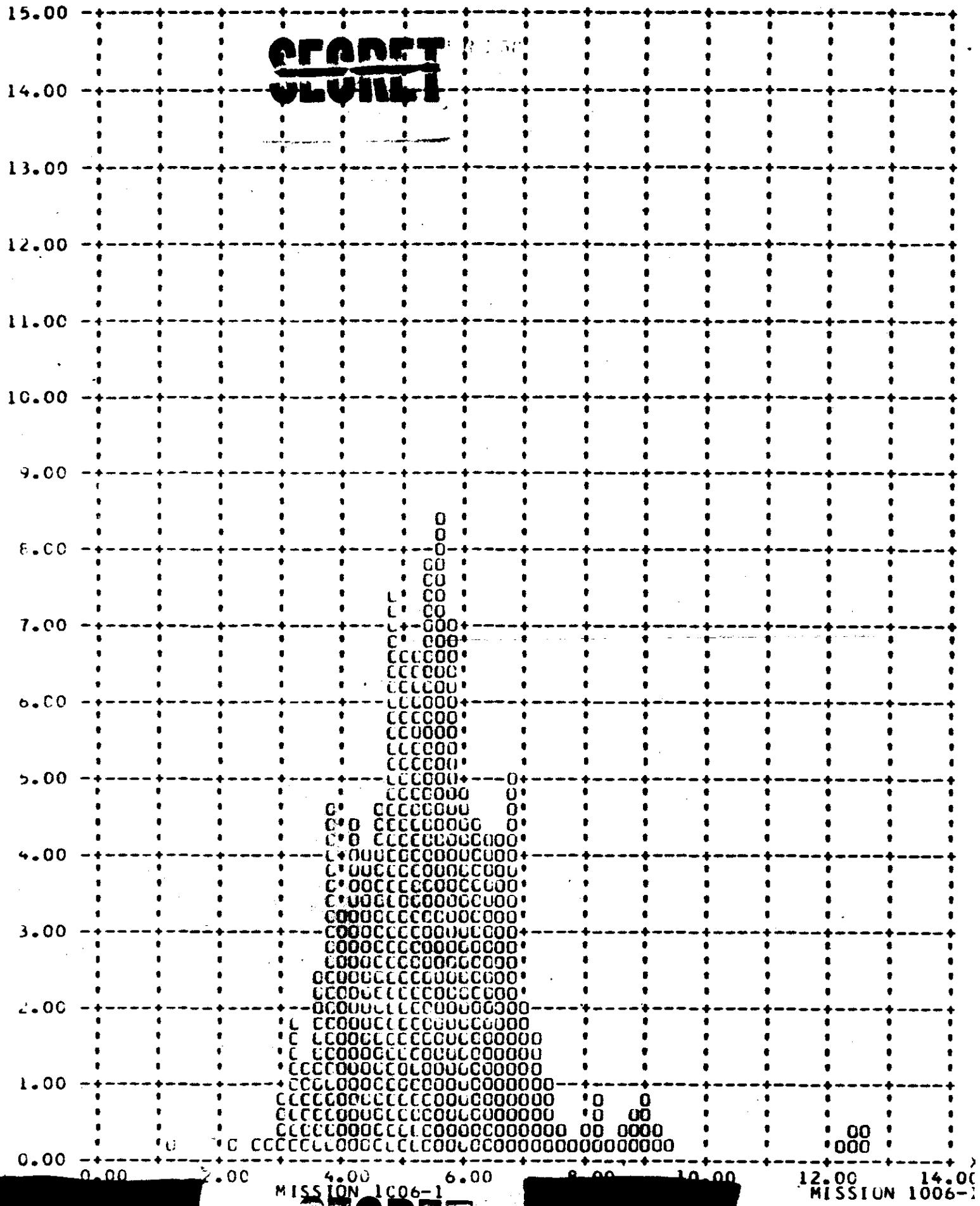
Y ALONG TRACK RESOLUTION LIMIT - FEET (X) VERSUS FREQUENCY - PERCENT (Y)



J-09 A BUCKET 9-14-64

MISSION 1006-1 FRAMES 1-6 OF EACH OP OMITTED 90 PERCENT = 6.74

Y CROSS TRACK RESOLUTION LIMIT - FEET (X) VERSUS FREQUENCY - PERCENT (Y)



## Y V/H RATIO ERROR - PERCENT (X) VERSUS FREQUENCY - PERCENT (Y)

3.75

3.50

3.25

3.00

2.75

2.50

2.25

2.00

1.75

1.50

1.25

1.00

0.75

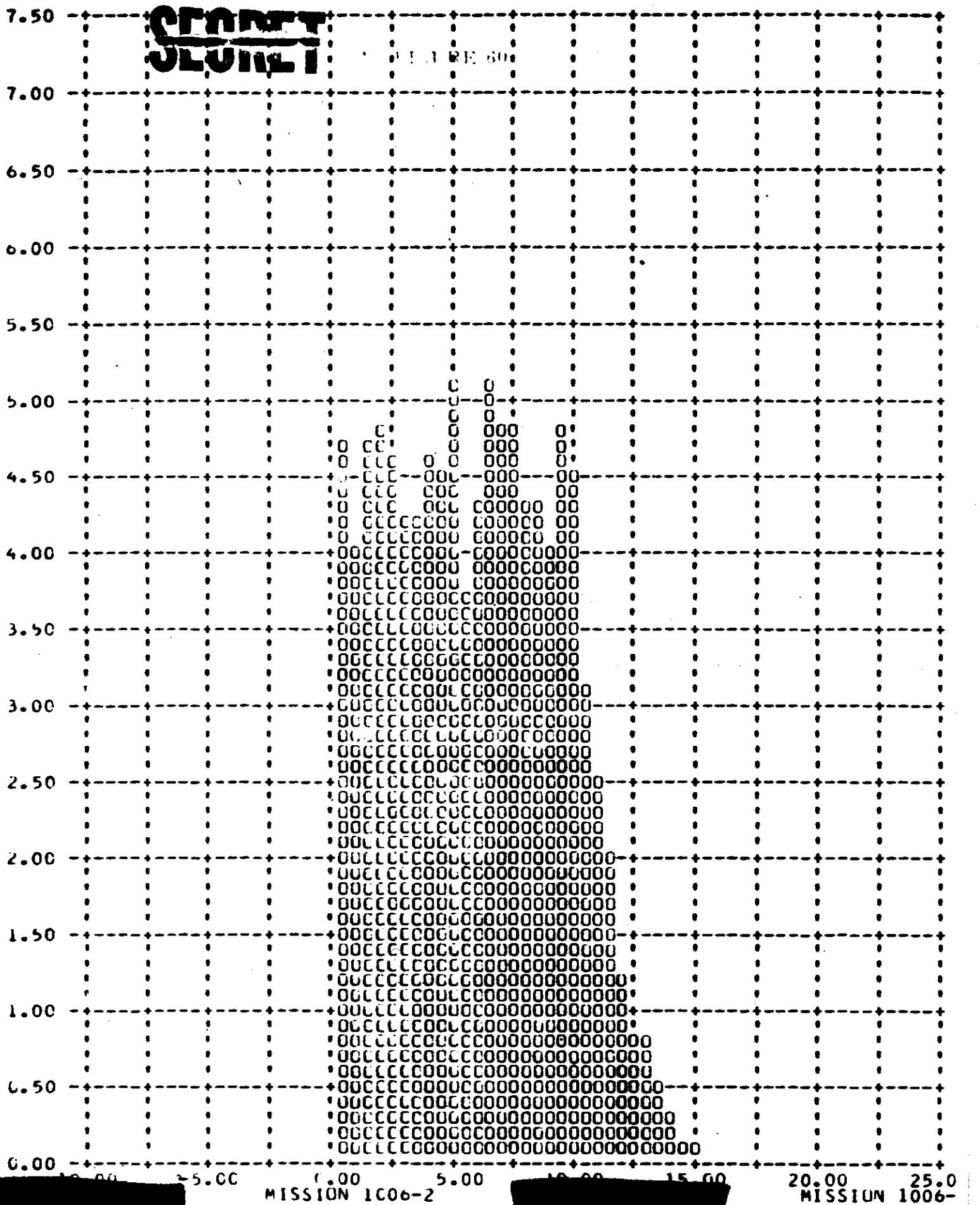
0.50

0.25

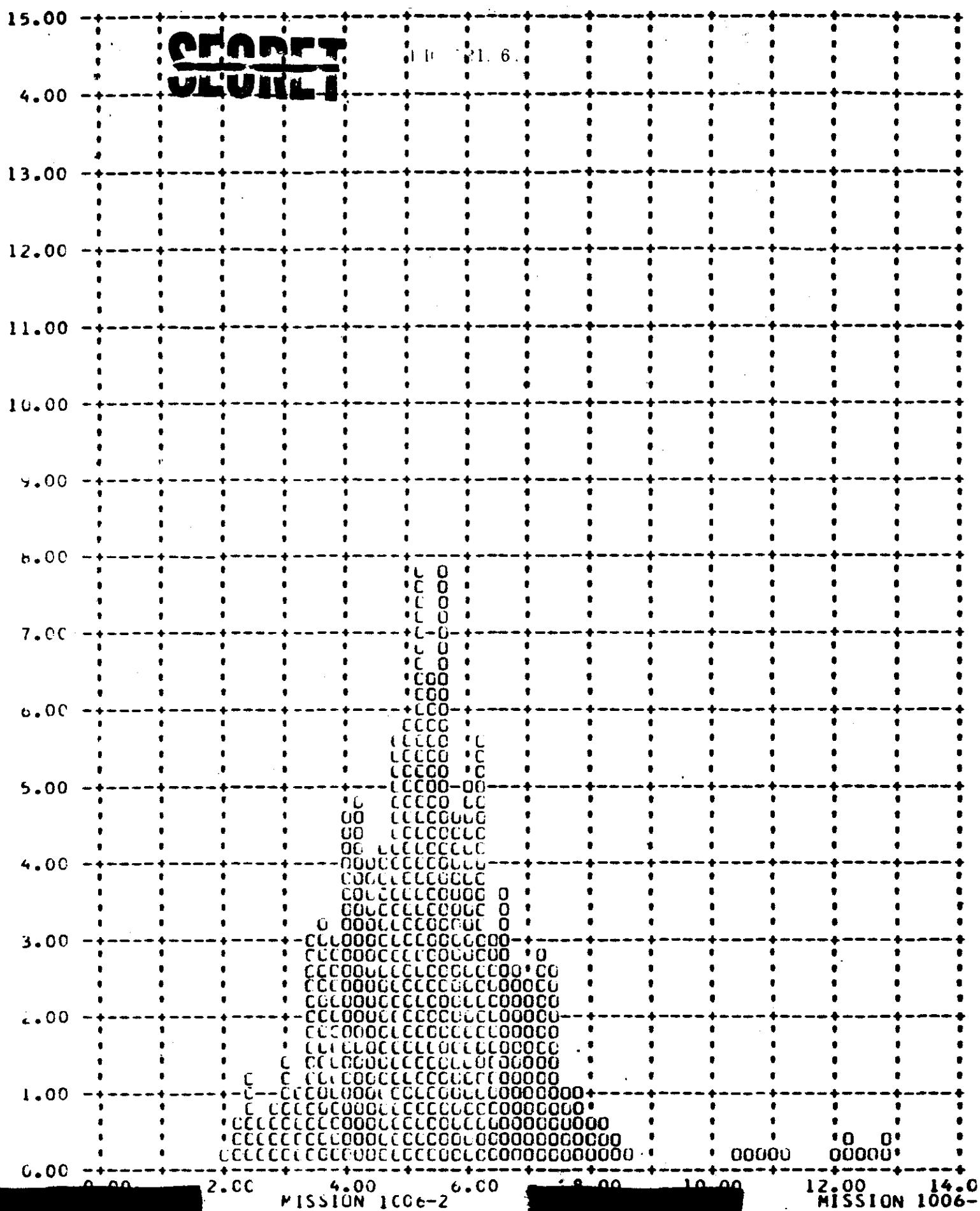
0.00

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~~REF ID: A6512~~20.00 15.00 10.00 -5.00 0.00 5.00 10.00 15.00  
MISSION 1006-2~~SECRET~~  
~~REF ID: A6512~~

## Y ALONG TRACK RESOLUTION LIMIT - FEET (X) VERSUS FREQUENCY - PERCENT (Y)



## Y CROSS TRACK RESOLUTION LIMIT - FEET (X) VERSUS FREQUENCY - PERCENT (Y)



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SECTION XVI

MISALATION DOSAGE - MISSION 1006

The dosimeter pack from Mission 1006 was recovered, processed, and evaluated for radiation dose received in orbit. Approximately one roentgen of radiation was measured by the dosimeter over a period of about eight hours. The radiation dose received by Mission 1006 was the same as that received by the dosimeter pack on the Gemini 3 flight. Film type Kodak Royal X Pan was used for processing the dosimeter film. The dose received by the dosimeter pack was approximately 1.0 roentgen. The dose received by the dosimeter pack on the Gemini 3 flight was approximately 1.0 roentgen. The dose received by the dosimeter pack on the Gemini 3 flight was approximately 1.0 roentgen. The dose received by the dosimeter pack on the Gemini 3 flight was approximately 1.0 roentgen.

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It was noted in Section IX that the tail booms of the C-119's could be detected even though they are smaller than the calculated resolution limit. As stated the booms were detectable since they were relatively high contrast and widely separated. It is believed that this explains why smear is not apparent when the image motion is known to be excessive, as during Mission 1600.

The incorporation of a yaw and roll scanner to reduce image motion will be of great benefit in the detection of small objects. This aid only in the detection of high contrast objects which will offer limited gains. On the other hand, objects of very low contrast will be more difficult to detect. However, the low contrast objects will be more prevalent in the cultural areas. The gains that can be realized by such equipment as the yaw programmer will, therefore, be much more significant.

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## SECTION XVII

### RELIABILITY

Reliability calculations for the payload are based on a sample beginning with M7. Hence both the major part of the Mural Program and the "J" Program are covered in the evaluation. For certain auxiliaries, i. e., the stellar-index camera and horizon cameras, the sample size is sufficient to recognize incorporation of modified equipment or new designs. However, reliability was one of the principal reasons for the modifications. However, for primary mission functions, the sample size is insufficient with reliability reporting for the vehicle.

The reliability estimates of this section deal exclusively with the payload. Failures to achieve orbit or vehicle induced failures are there excluded. Recoveries before a complete mission has been completed are considered as full missions providing that early termination was caused by reasons not connected with payload operation. Film quality is not considered in the reliability estimate calculation. Hence, only electrical and mechanical functioning are considered.

The reliability estimate is also divided into primary and secondary functions. The primary functions are operation of the panoramic cameras, main camera door operation, operation of the payload clock, and recovery operations. The secondary mission functions are horizon camera operation, excluding catastrophic open shutter failure mode, auxiliary data recording, and stellar-index camera operation.

#### A. Panoramic Camera Reliability

Sample size = 56 opportunities to operate

One failure - scanning shutter on slave instrument subsystem

Failures = 0.007 views per camera per mission

Estimated Reliability = 96.2% at 50% confidence level

#### B. Main Camera Door Reliability

25 vehicles = 240 = 56 opportunities to operate

1 malfunction observed - failed to eject not recovered

Estimated Reliability = 97.0% at 50% confidence level

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### C. Payload Clock Reliability

28 completed missions in sample

No failures

Estimated Reliability = 97.6% at 50% confidence level

### D. Estimated Reliability of On-Orbit functioning of payload

$$.982 \times .970 \times .976 = .923$$

### E. Recovery System Reliability

24 opportunities to recover

1 failure - improper separation due to water seal cutout activation

Estimated Reliability = 95.6% at 50% confidence

### F. Stellar-Index Camera Reliability

Sample begins with M13

Sample size = 16

Number of failures = 5

Estimated Reliability = 73.0% at 50% confidence level

### G. Horizon Camera Reliability

Sample includes M27, J5A, J5B, J7A and J8B. These are the redesigned 55mm focal length horizon cameras.

One failure has occurred - center-of-format switch

Estimated Reliability of a Single Camera = 81.9% at 50% confidence

Estimated Reliability of Four (including cameras in redundant and non-redundant system) = 80.8% at 50% confidence

### H. Horizon Camera Door Reliability

Sample Size = 22 x 4 = 88 opportunities to operate

3 failures have occurred

Estimated Reliability = 69.2% at 50% confidence

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## I. Stellar-Index Camera Door Reliability

Terrain Door, Stellar Door and Deployment of Stellar Baffle  
are functions considered

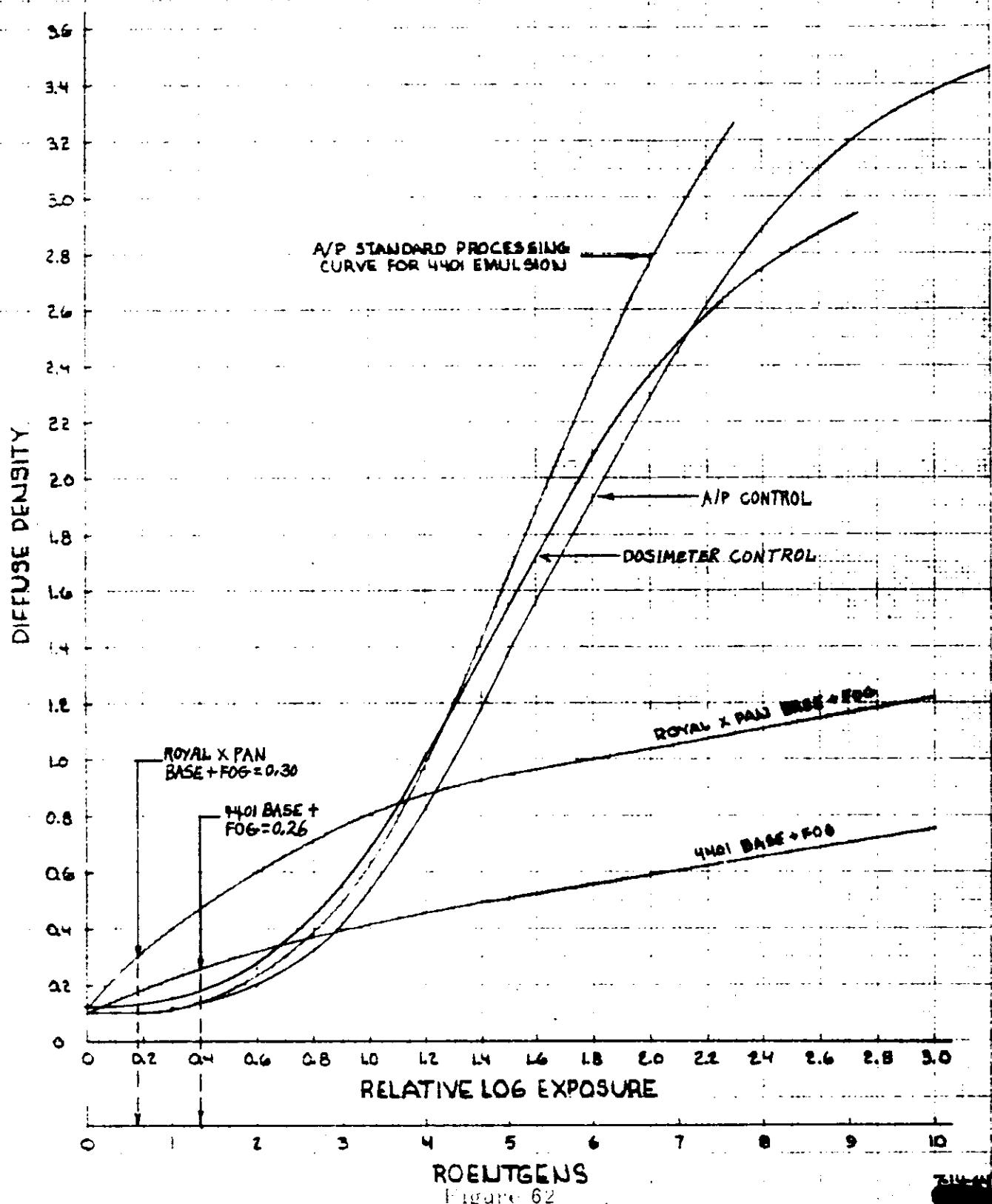
Sample Size =  $16 \times 3 = 48$  chances to operate

One failure; stellar baffle failed to deploy

Estimated Reliability = 98.7% at 60 minutes

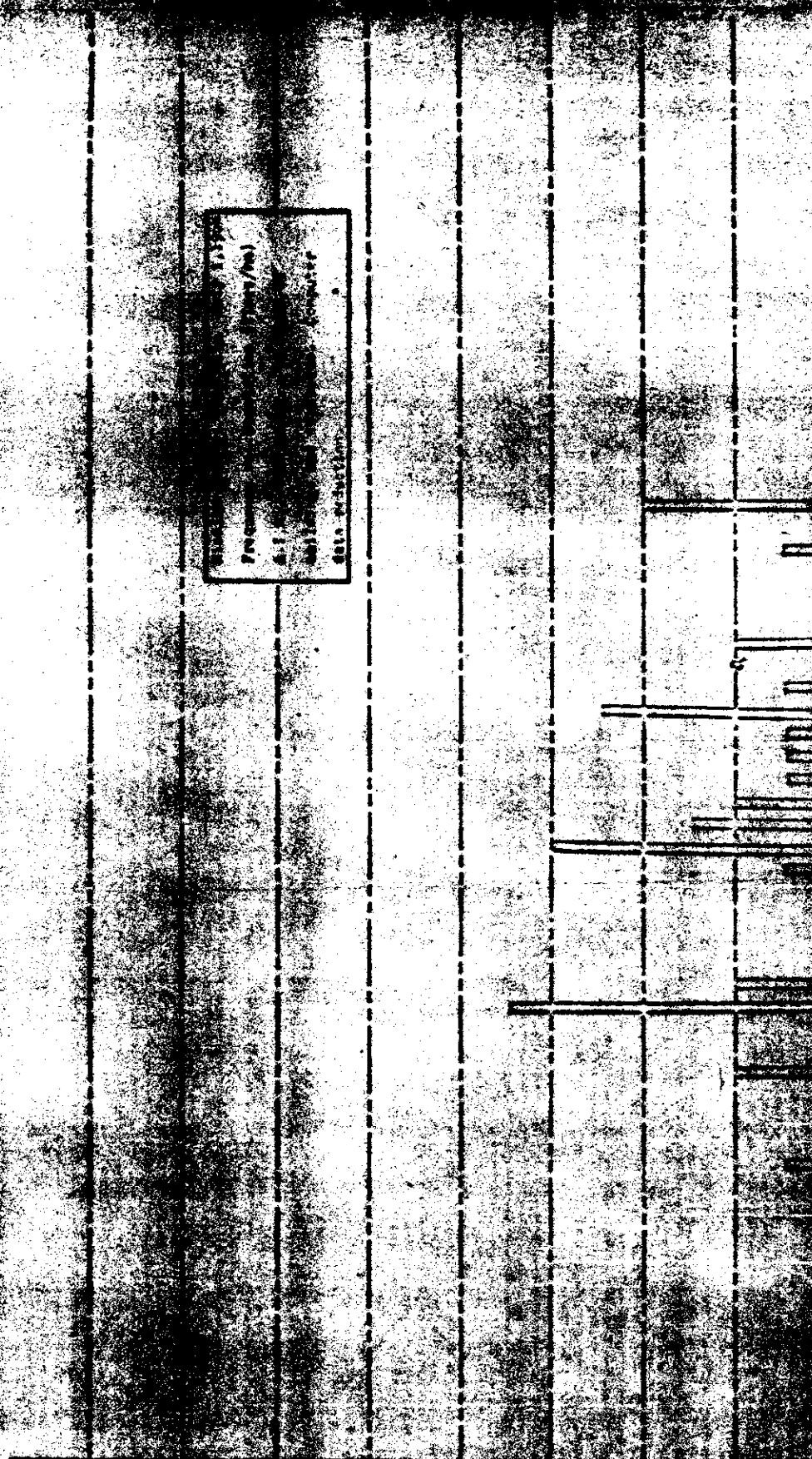
# RADIATION DOSAGE

## MISSION 1006-2



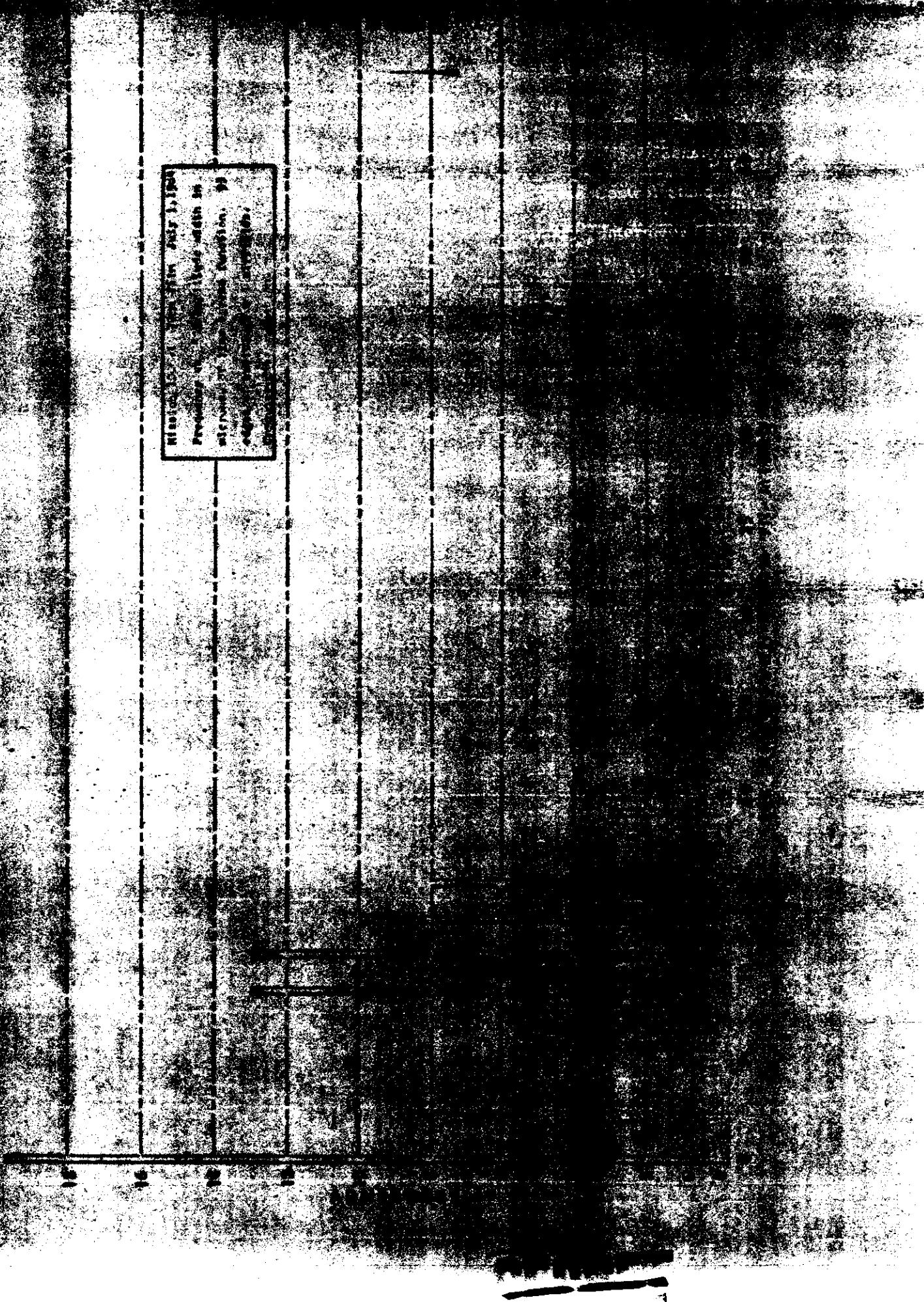
ROENTGENS

Figure 62



REVENGE





Film Manufacturers Association

**ANSWER**

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#### **Filter: 3M™**

#### **Development Committee**

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10. The following table shows the number of hours worked by each employee.

## Absolute Log E

THE  
WORLD

Fog

1  
2  
3  
4  
5  
6  
7  
8  
9  
10  
11  
12  
13  
14  
15  
16  
17  
18  
19  
20  
21  
22

Permit

-UILT

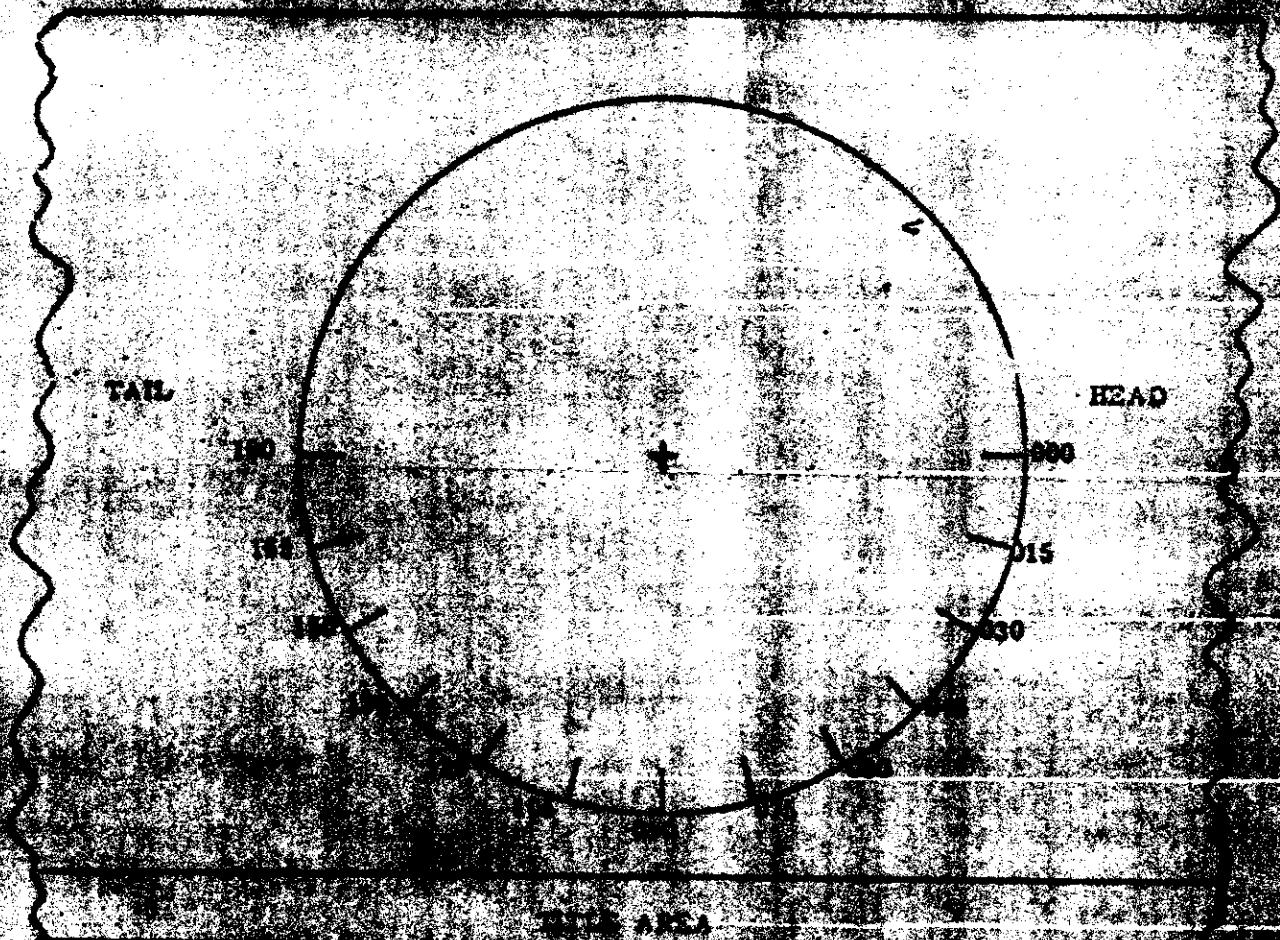
Control Standard Start Up  
Intermed. Full Primary Intermed. Full  
0.000000 1.00 1.00 1.00 1.00

CRONEX

**Section V. Page 4**

Head and Tail	Aft Camera	Mission
Head and Tail	Aft Camera	Mission
.16	.16	Pull
.18	.18	
.20	.20	
.24	.24	
.26	.26	
.27	.27	
.28	.28	
.29	.29	
.33	.33	
.34	.34	
.70	.70	
1.00	1.00	1.00
1.10	1.10	1.10
1.67	1.67	1.67
1.72	1.72	1.72
2.00	2.00	2.00
2.10	2.10	2.10
2.43	2.43	2.43
2.47	2.47	2.47
2.62	2.62	2.62
2.72	2.72	2.72

APPENDIX D  
REFERENCE SYSTEM FOR ORIENTATION OF "J" MISSION EDGES



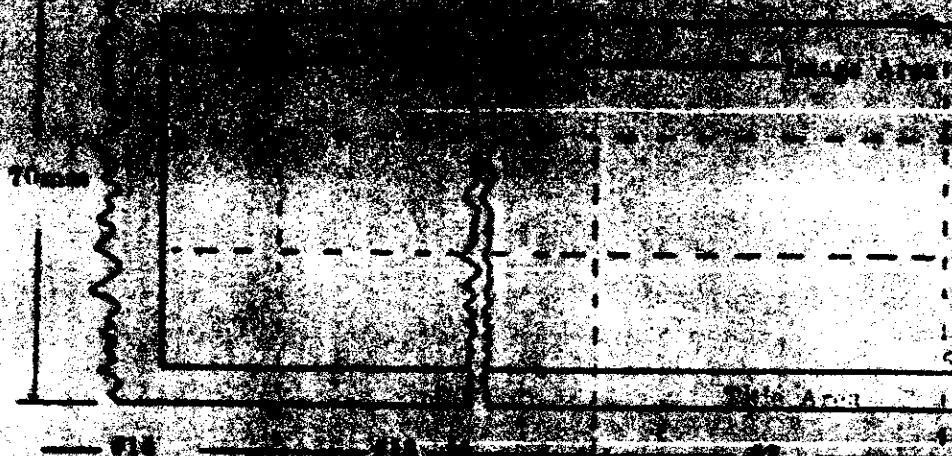
110

REF ID: A65294

**APPENDIX "P"**

**DATA FOR MANUAL IDENTIFICATION OF EDGES**

**TIN**



**PRINTED NEGATIVE IN EMULSION SIDE**

**SPAR**  
**UNIVERSITY**



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July 15, 1964

**TITLE:**

## **Summary of Microbiological Data Collected from Minnows 2004-2**

## **SECTION I: INTRODUCTION**

**Microdeformations** - The microdeformations of the edges were evaluated by the same methods as the main edge. The spread function was used to evaluate the width of the edges. A summary of the edge parameters is given in Table II. Given in Table III are the standard deviation, coefficient of dispersion, and number of edges. Section III is a tabulation of the location, description, and image quality of the main edge. Frequency plots of the spread function and resolving power data are given in Section IV, to show the distribution of values. Section V is included to show the comparison of the main and minor edges. A diagram of the reference system used in describing the edges is shown in Figure 1, and a summary of the symbols used to label the edges will be found in Appendix A.

The last time I saw him was at the airport in  
Seattle where he was flying to  
ascend with me.

- (a) Water is a homogeneous homomeric colloidal stratified medium.

(b) Water is a homogeneous homomeric colloidal stratified medium.

(c) Water is a homogeneous homomeric colloidal stratified medium.

(d) Water is a homogeneous homomeric colloidal stratified medium.

graphical representation of the data, such as bar charts or line graphs, can help in visualizing the trends and patterns.

**SLUH**

ANALYSIS OF AEROMAGNETIC IMAGE TO Evaluate System Performance

SECTION II SUMMARY SHEET

Mission 1000-2

Resolution is 1000 ft. at the center point of the 4404 curve 1000 ft. above the ground.

RESOLUTION TEST

Average	86.5
Standard Deviation	22.1
Coefficient of Dispersion	26%
Number of Observations	109

Spread function shows a slight amplitude increase from edge trace data reduced by the same techniques.

Average	11.4
Standard Deviation	3.3
Coefficient of Dispersion	29%
Number of Observations	109

CRANE

Analysis of Photographic Image to Evaluate System Performance

Mission 1006-2

Section III

Cameras	Pano	Exposure	Location	Orientation	Subject	50% Amplitude Spectral Power		A.I.M. Resolution
						Wavelength	Bandwidth	
Aft.	D73	040	B-10	180	Airfield	5.1	59	
Aft.	D73	040	B-10	180	Airfield	5.2	105	
Fwd	D73	041	B-11	005	Airfield	5.2	120	
Fwd	D73	044	B-11	005	Airfield	12.6	70	
Fwd	D73	028	A-6	175	Airfield	6.3	121	
Fwd	D73	023	E-12	148	Airfield	11.8	70	
Fwd	D73	037	E-6	000	Airfield	6.2	60	
Fwd	D73	040	E-14	168	Airfield	22.4	93	
Fwd	D73	040	E-14	163	Airfield	9.7	101	
Fwd	D73	040	A-10	068	Airfield	6.3	134	
Fwd	D73	040	A-10	063	Airfield	6.3	116	
Fwd	D73	040	C-11	137	Airfield	11.7	52	
Fwd	D73	041	C-11	137	Airfield	11.7	73	
Aft.	D73	041	B-11	140	Airfield	6.1	68	
Aft.	D73	040	A-6	180	Airfield	6.3	116	
Aft.	D73	040	A-6	180	Airfield	6.3	73	
Aft.	D73	040	A-6	090	Airfield	6.3	67	
Aft.	D73	040	C-7	090	Airfield	6.3	78	
Aft.	D73	040	E-6	160	Airfield	6.3	68	
Aft.	D73	040	C-6	160	Airfield	6.3	66	

**CRARF**  
47185-4

ITEM NO. C-100

14	AIR.	D73	613	125	111
14A	AIR.	D73	613	125	111
15	AIR.	D73	613	125	111
15A	AIR.	D73	613	125	111
16	AIR.	D73	613	125	111
16A	AIR.	D73	613	125	111
17A	AIR.	D73	613	125	111
18	AIR.	D73	613	125	111
18A	AIR.	D73	613	125	111
19	AIR.	D73	613	125	111
19A	AIR.	D73	613	125	111
20	Fwd.				
20A	Fwd.	D73	613	125	111
21	AIR.	D73	613	125	111
21A	AIR.	D73	613	125	111
22A	AIR.	D73	613	125	111
23	40				
23A	AIR.	D73	613	125	111
24	Fwd.				
24A	Fwd.	D73	613	125	111

REF  
1

<u>Edge No.</u>	<u>Camera</u>	<u>Date</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>502</u>	<u>Amplitude</u>	<u>Spread</u>	<u>Function</u>	<u>Width</u>	<u>Avg. W.</u>
							(Microns)					
25	P-6	1960	100	A-7	210	Airfield	0.8	55				
25A	P-6	1960	100	A-7	110	Airfield	0.4	55				
26	P-6	1960	100	A-7	110	Airfield	0.4	55				
26A	P-6	1960	100	A-7	110	Airfield	0.4	55				
27	P-6	1960	100	A-7	110	Airfield	0.4	55				
27A	P-6	1960	100	C-13	022	Airfield	15.0	55				
28	P-6	1960	100	A-13	022	Airfield	14.8	53				
28A	P-6	1960	100	A-13	022	Airfield	12.7	53				
29	P-6	1960	100	A-7	022	Airfield	5.1	53				
29A	P-6	1960	100	A-7	022	Airfield	6.5	53				
30	P-6	1960	100	A-7	022	Airfield	12.0	53				
30A	P-6	1960	100	A-7	022	Airfield	12.0	53				
31	P-6	1960	100	C-13	110	Airfield	10.0	53				
31A	P-6	1960	100	C-13	110	Airfield	14.0	53				
32	P-6	1960	100	A-10	022	Airfield	0.8	53				
33	P-6	1960	100	B-13	022	Airfield	7.1	53				
33A	P-6	1960	100	B-13	022	Airfield	7.1	53				
34	P-6	1960	100	C-13	022	Airfield	7.1	53				
34A	P-6	1960	100	C-13	022	Airfield	7.1	53				
35	P-6	1960	100	C-4	022	Airfield	0.8	53				
35A	P-6	1960	100	C-4	022	Airfield	0.8	53				

GRAPH

**SOURCE**  
**LEVEL**

WGS  
Altitude  
Spread  
Function  
WGS84  
(Meters)

<u>Edge No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientation</u>	<u>Subject</u>	<u>Height</u>	<u>Notes</u>
54A	TM	2111	066	A-3-11	030	Ground	11.5	
55	TM	2111	040	B-6	030	Ground	6.8	
56A	TM	2111	040	C-6	030	Ground	6.5	
57	TM	2111	040	C-6	030	Ground	6.5	
57A	All	2111	041	C-8	150	Ground	6.5	
58	All	2110	028	A-9	030	Airfield	10.5	
58A	All	2110	028	A-9	030	Airfield	7.0	112
59	All	2110	028	C-10	030	Airfield	11.0	
60	All	2110	028	B-10	030	Airfield	10.0	
60A	All	2110	028	B-10	030	Airfield	10.0	
61	All	2110	028	B-10	030	Airfield	10.0	
61A	All	2110	028	A-1	030	Airfield	7.5	
62	All	2110	028	B-7	035	Airfield	10.5	
62A	All	2110	028	B-6	035	Airfield	5.5	
64	All	2110	028	B-7	035	Airfield	11.5	75
64A	All	2110	018	B-7	035	Airfield	10.2	50
65	All	2110	028	A-6	035	Airfield	10.0	
65A	All	2110	028	A-6	035	Airfield	10.0	
66	All	2110	028	A-6	035	Airfield	10.0	
66A	All	2110	028	A-6	035	Airfield	10.0	
67	All	2110	028	A-6	035	Airfield	10.0	
67A	All	2110	028	A-6	035	Airfield	10.0	

SECRET  
REF ID: A6512

<u>Expt No.</u>	<u>Camera</u>	<u>Pass</u>	<u>Frame</u>	<u>Location</u>	<u>Orientatn</u>	<u>Subject</u>	<u>50<sub>1</sub></u> Amplitude	<u>50<sub>2</sub></u> Spiral Function	<u>Width</u> <u>(Microns)</u>	<u>Reproduction</u>
69	A.H.	D103	133	A-a	070	Settling bed	5.0	2.0		
69A	A.H.	D103	133	A-a	070	Settling bed	5.0	2.0		



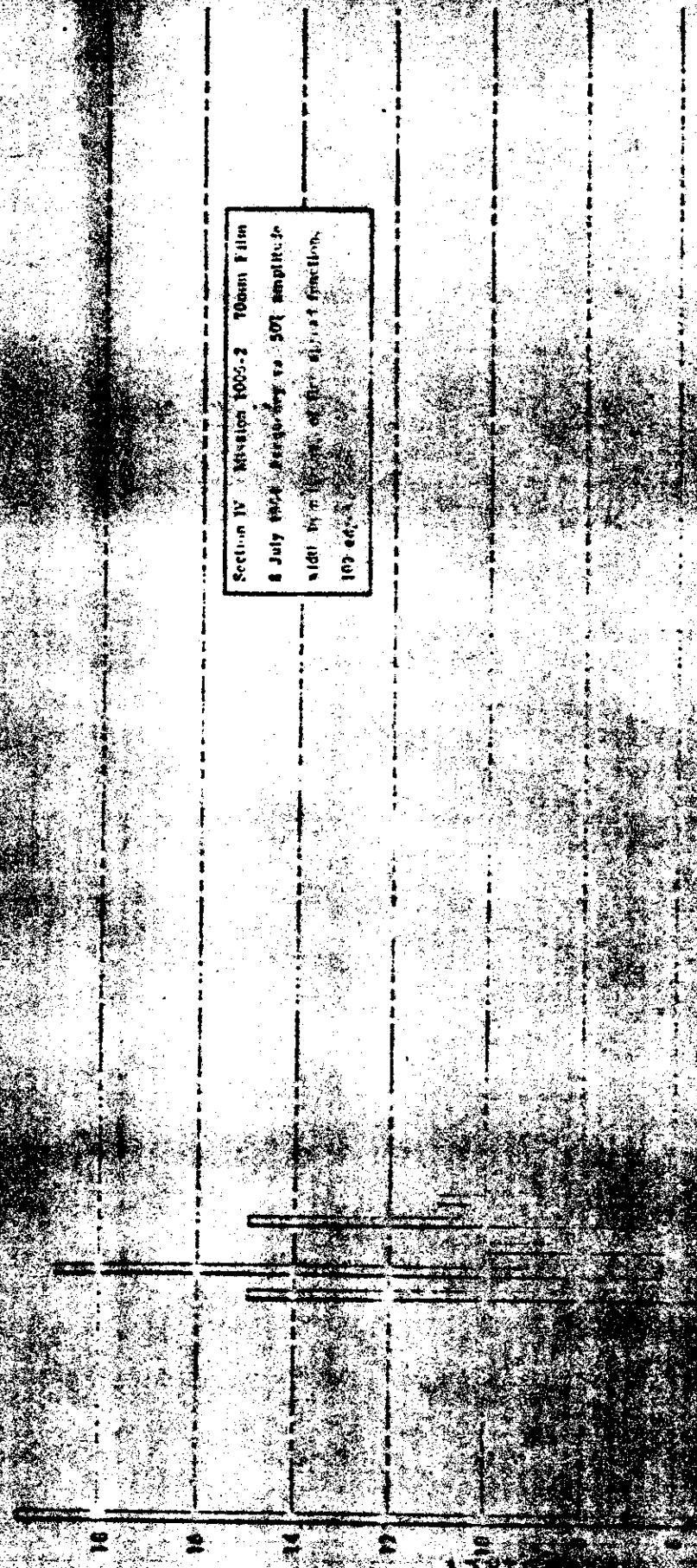
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Section IV - Mission 1005-2 70mm Film  
8 July 1969 Recording to 300' amplitude  
with 10% noise margin fractions  
10% each



**Section V Page 2**

**Sensitometric Data**

**Wavelength: 365 m $\mu$**

**Emulsion No.: 140000**

**Lamp No.: 1993**

**Exposure Time: 1/25 sec.**

**Wedge No.: VII-15**

**Filter: U.v.15**

**Development Conditions:**

**Temperature: 70° + 1° F**

**Agitation: Periodic, 100 ml. of PMS 110X10 35° C.**

**Time: 10 minutes**

**Absolute Latitude: 1.00**

SEARCH PAGE 2

SEARCHED - INDEXED - SERIALIZED - FILED

Process Control Standard

SEARCHED

Primary Response Full

Primary Response Full

Fog

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SEARCHED  
INDEXED  
SERIALIZED  
FILED

SEARCHED - INDEXED - SERIALIZED - FILED

Section V. Page 3

Front and Rear

Front and Rear

Forward Camera

All Camera

None      Tall

None      Tall

Fog

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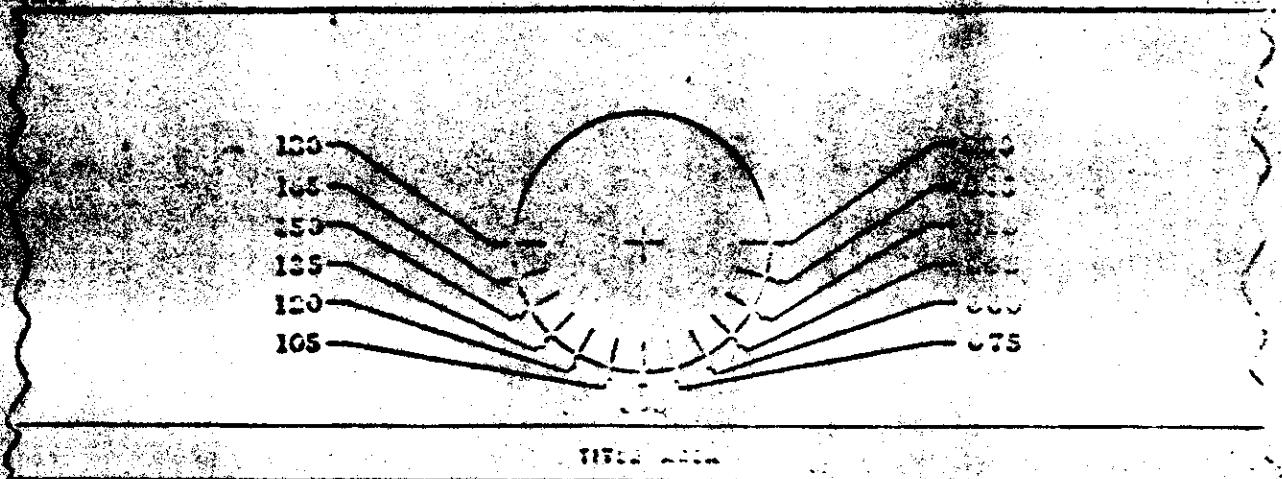
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GENERAL EQUIPMENT

**APPENDIX "A"**

**Reference System For Orientation Of C/M/J Missed Edges**

**original negative -- emulsion up**

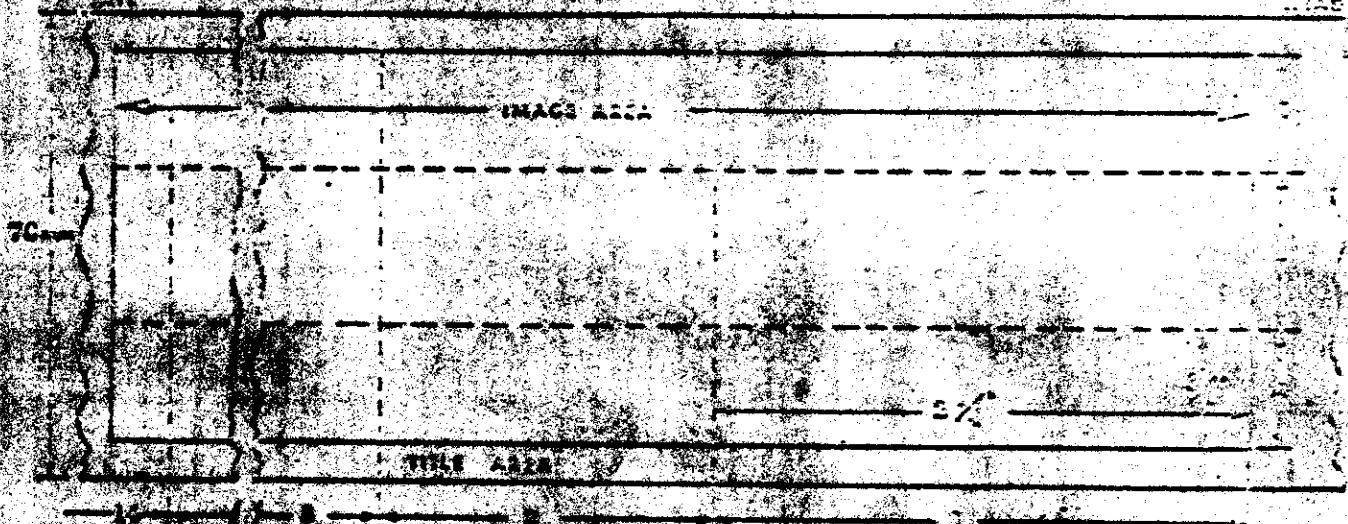
**TAR**



**Grid For Orientation Of C/M/J Missed Edges**

**original negative -- emulsion up**

**TAR**



## Mission 1006-1 Mission 1006-2

Master Camera - AFSPPPL	65/88	64/81
- [REDACTED]	84	84
- [REDACTED]	97	90
Slave Camera - AFSPPPL	71/90	72/90
- [REDACTED]	87	87
- [REDACTED]	92	94

The data from [REDACTED] shows that in most cases two traces were made over the same edge. Assuming that this repeated measurement was made by the same operation and at nearly the same time as the first measurement, it would be expected that the results should be fairly close to each other. An examination of the data, however, shows that in many cases the reduced MTF/AIM values differ by a factor greater than two. This lack of repeatability is most probably caused by minor error in aligning the slit to the edge or by positioning the slit to another portion of the imaged edge. If the former is the cause of the difference, it is admissible to argue that the lower value is an incorrect value as misalignment would tend to spread the true edge. Following this line of argument, the [REDACTED] values were re-averaged and are shown in the table as asterisked values. If the latter cause of lack of repeatability is the case, the measurements then represent two different edges of differing quality some of which may be due to changes in the actual edge itself. Under such a circumstance, averaging of all the results represents the most logical statistical treatment of the data. In actuality, the differences most probably are due to a combination of both causes. Since it is not apparent that these variables and others can be separated, some caution is therefore required in using edge traces as absolute measures of quality. However, their value as a relative quantized measure of quality is recognized, and further refinement of techniques and possibly instruments could result in a reliable and reproducible method of measuring system quality largely free of subjective judgement which is inherent in other techniques.

The AFSPPPL data is reduced for a 350 micron and 43 micron long slit. It is noted that in most cases the values obtained with the 43 micron slit are higher. This should be expected as an edge should be straighter and of more uniform reflectivity over an approximate 40 foot length than an edge approximately 350 feet in length. However, it is noted that there are cases in which the 43 micron slit shows a lower value than the 350 micron slit. From what is known of the technique and practices at AFSPPPL, no reason is apparent for this difference. It is also noted that the AFSPPPL values for the 350 micron slit

generally are lower than the values for the 320 micron slit used by [REDACTED] This difference is attributed to machine sensitivity, operator skill, selection of edges, data reduction or some combination thereof. It further suggests that standardization may be necessary before adopting the edge trace measurement as a measure of system quality. It is understood that this is a matter of primary concern for the governmental-industry team on image quality.

While difficulties such as these exist in using the edge trace data, attempts are nevertheless being made to correlate edge trace values with other known system parameters. At this time, no definite correlations are apparent. Continuing efforts are underway and will be reported at a later date if significant correlations are established.

**DRAFT**

## SECTION VIII

### SCAN RATE ANALYSIS

#### A. GENERAL

Since the first operational engineering film became available at Advanced Projects (Mission 1004), studies have been conducted of the panoramic camera scan velocity as measured by the spacing of the 200 cps timing light marks on the payload material. While the velocity for any given frame is specified to be a constant, the value of which depends on the V/H setting, actual velocity patterns with variations of more than 20% have been observed in frames of each mission. Velocity profiles for ground tests are very similar to those obtained from flight material. Further, there are strong similarities between the profiles of different cameras when operated at similar speeds.

Unfortunately, neither the causes nor the effects of the scan velocity variations are fully understood. Both the causes, which relate to camera characteristics, and the effects, which relate to image quality, are being extensively investigated by the contractor. The availability of operational payload material at contractor facilities has been an important contribution to these investigations.

#### B. EFFECTS OF SCAN HEAD VELOCITY VARIATIONS

The effects with which we are concerned are ones that produce a displacement of an image ray with respect to the film during exposure and thus result in a degradation of the image. If the deviation from a constant velocity involved only the scan head, as distinguished from the rotation of the lens cell, the effects would be only those of exposure variation and perhaps some lens distortions. These effects would clearly be very small. The exposure variation, which would appear as banding, has not been apparent in coverage of land areas. NPIC personnel have occasionally noted a slight banding in coverage during past missions over water areas which have a very constant tone value. This effect is consistent with the velocity profiles collected to date.

Although the drum and scan arm structures will allow some relative movement of the scan head, the magnitude of velocity deviations, as well as other test data, show that the disturbances affect other parts of the camera. Since the shape of the cam that compensates for forward motion is based on a constant scan velocity assumption, it follows that a deviation in FMC adjustment of the system, the scan rate deviations then would equate to a proportional FMC error and image smear.

Finally, the scan velocity data suggests a variety of noise or unresolvable motion elements. Since these elements can not be directly or analytically related to image ray film relative motions, the effects appear as residual degradations. The residuals may be sufficiently large to obscure systematic effects noted above. The geometry and dynamics of pan systems introduce particularly difficult problems in relating causes and effects. Several structural and analytical approaches are now being explored. It is expected that some significant correlations will be obtained.

### C. MEASUREMENT CONDITIONS

The 200 cps timing marks are produced by a neon bulb controlled by the system clock through an amplifying circuit. The clock signals are accurate to five parts in  $10^6$ . The bulb and circuit are known to have some self-heating effects, but these are considered to be entirely negligible in this case. The neon bulb is located in the scan head assembly approximately one inch behind the shutter slit (with respect to the direction of scan motion). The spacing of the timing marks is considered to be a direct measure of scan velocity. However, strictly speaking, it should be noted that the spacing is the resultant of scan head and film motion.

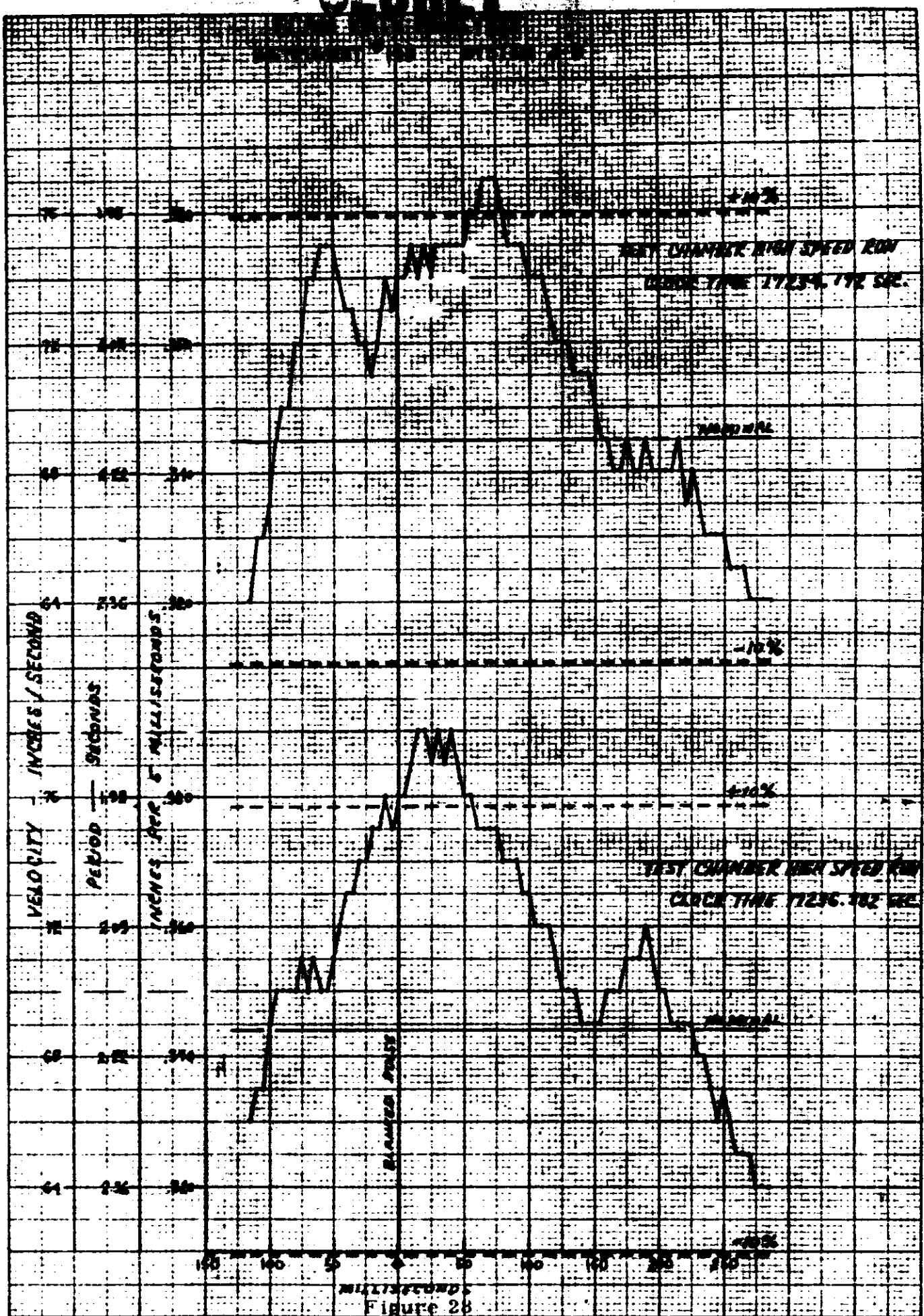
The timing marks on the film are of variable shape and density. Variations result from adjustment of the instrument, rail reflections, operating speed, and image density in the adjacent format area. While these variations lead to some variability in measurement accuracy, it has been found that with a suitable crosshair reticle the center-to-center spacing can be measured to about 0.001 inch. The measurements contained in this report as well as other made at Advanced Projects have been obtained with a 7x tube magnifier with a reticle graduated in 0.005 inch units. These data appear to have a probable error of less than 0.003 inch. The data generally has been recorded by 0.005 inch increments so that some smoothing of small fluctuations has occurred.

### D. PRESENTATION OF DATA

Because of the lack of significant differences in velocity characteristics between the two panoramic cameras or their performances on the two parts of the mission, the description and analysis of data has been combined into a single presentation. Figures 28 to 39, pages 122 to 137, show velocity profiles for the following conditions:

- a. Master camera (#148); four consecutive frames of high speed operation during HATS test, Figures 28 and 29.

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**Figure 28**

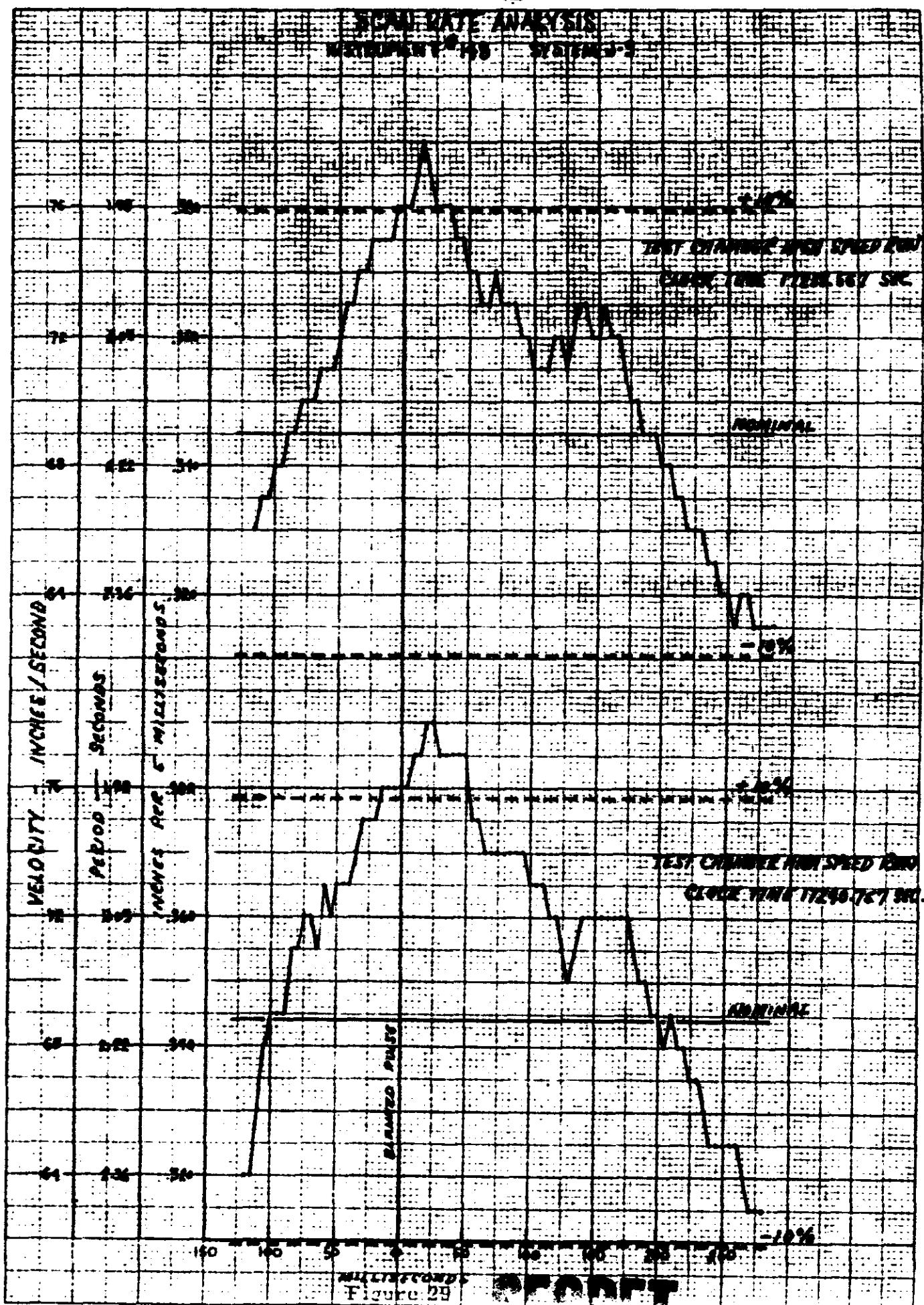
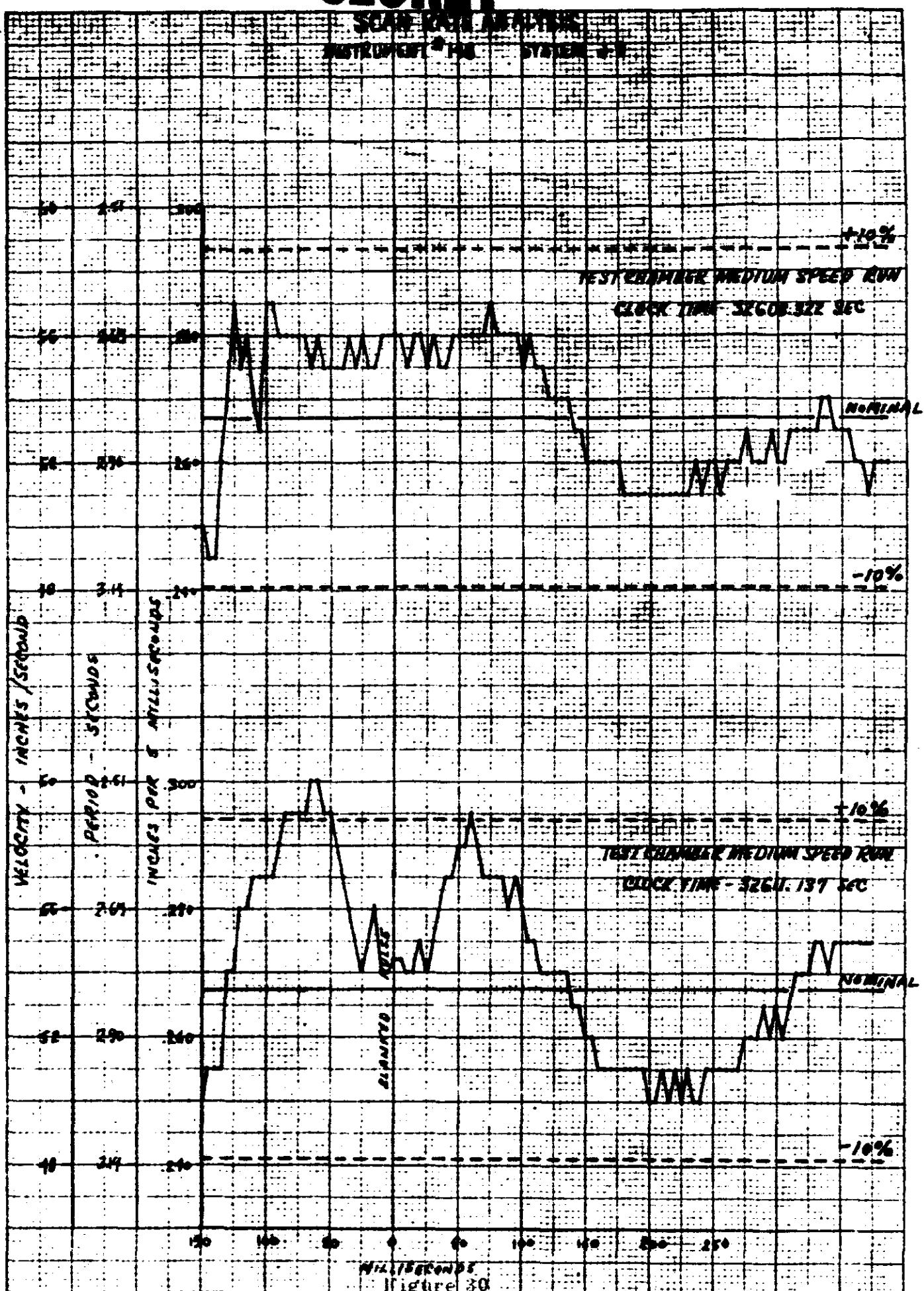


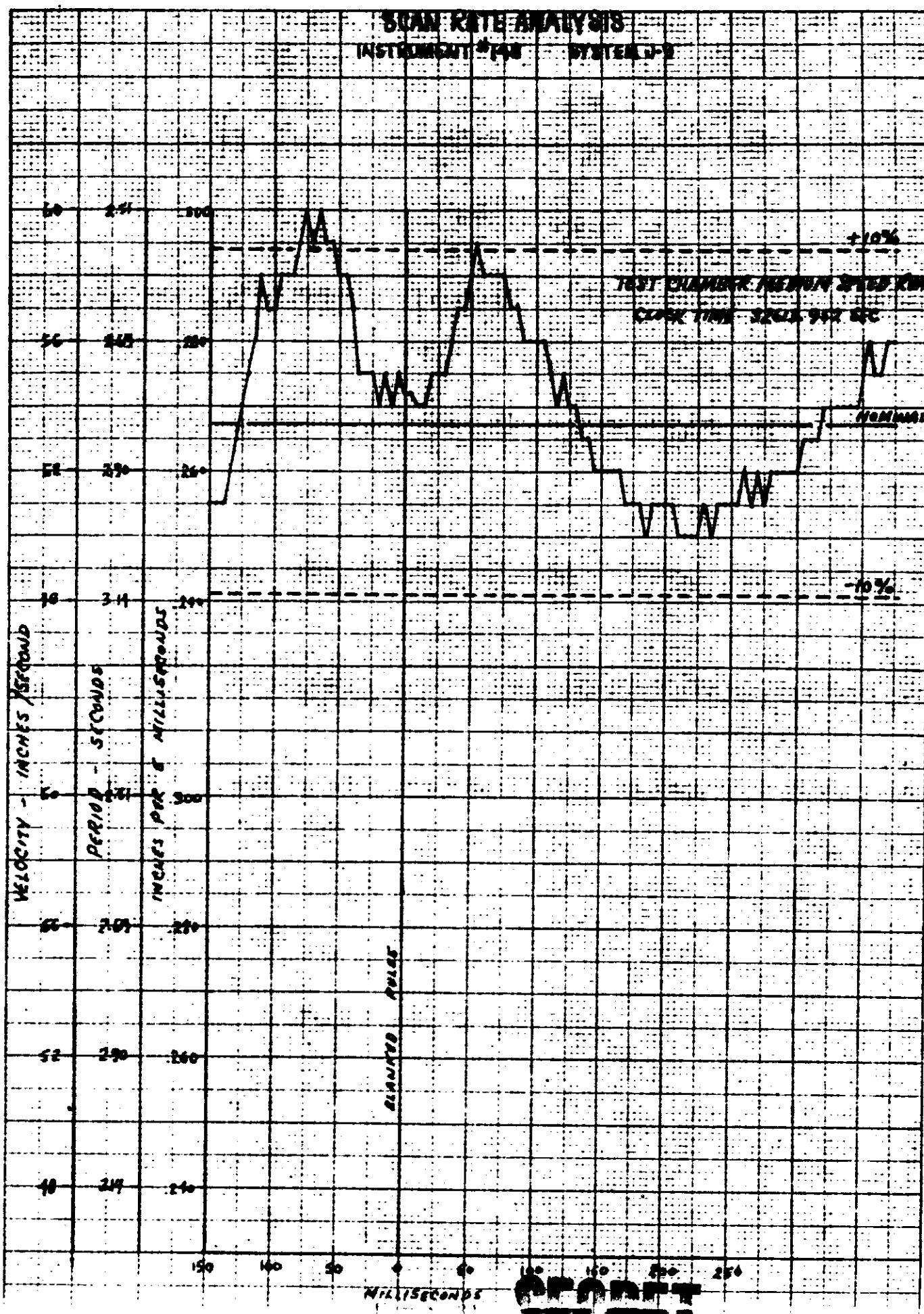
Figure 29

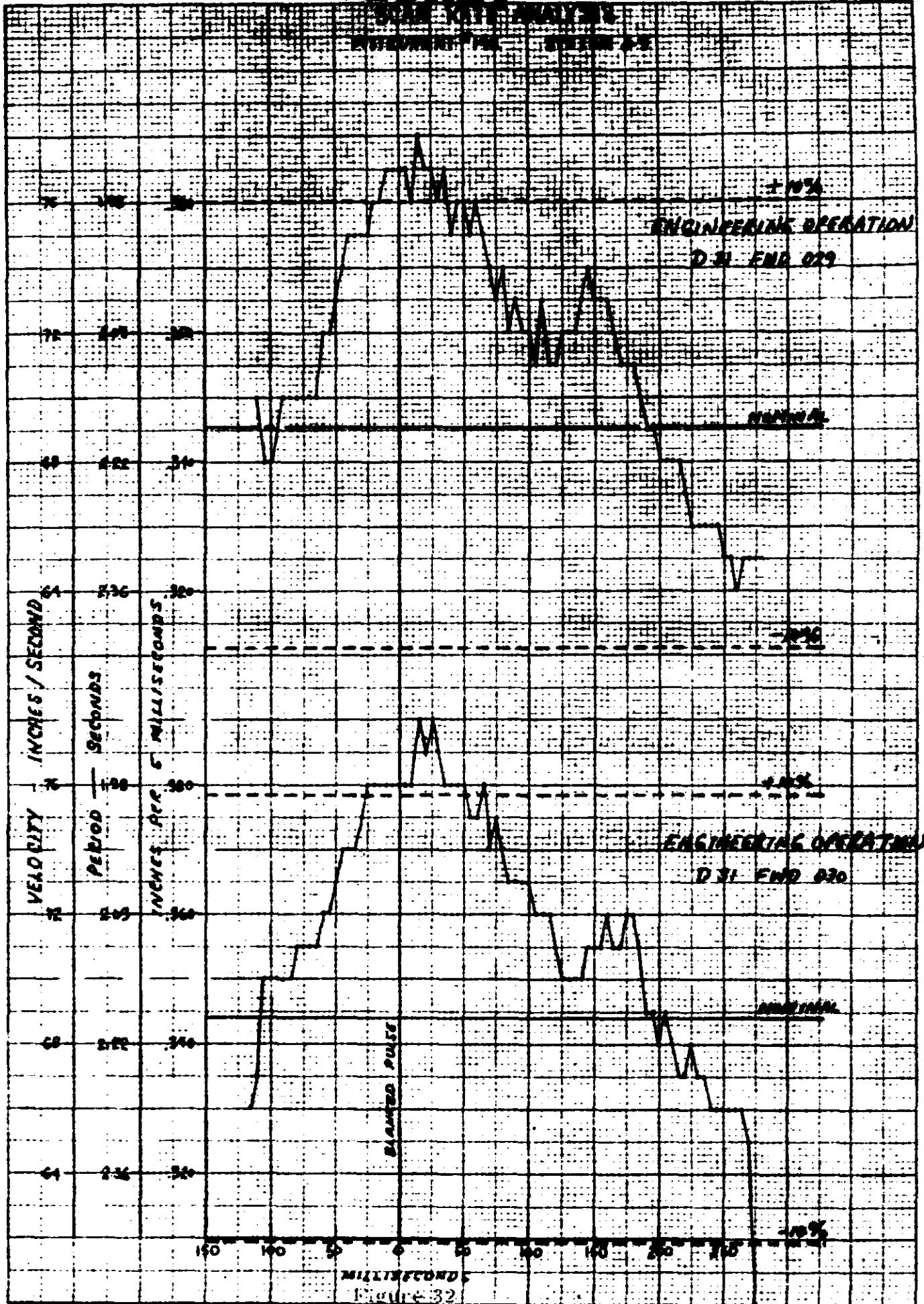


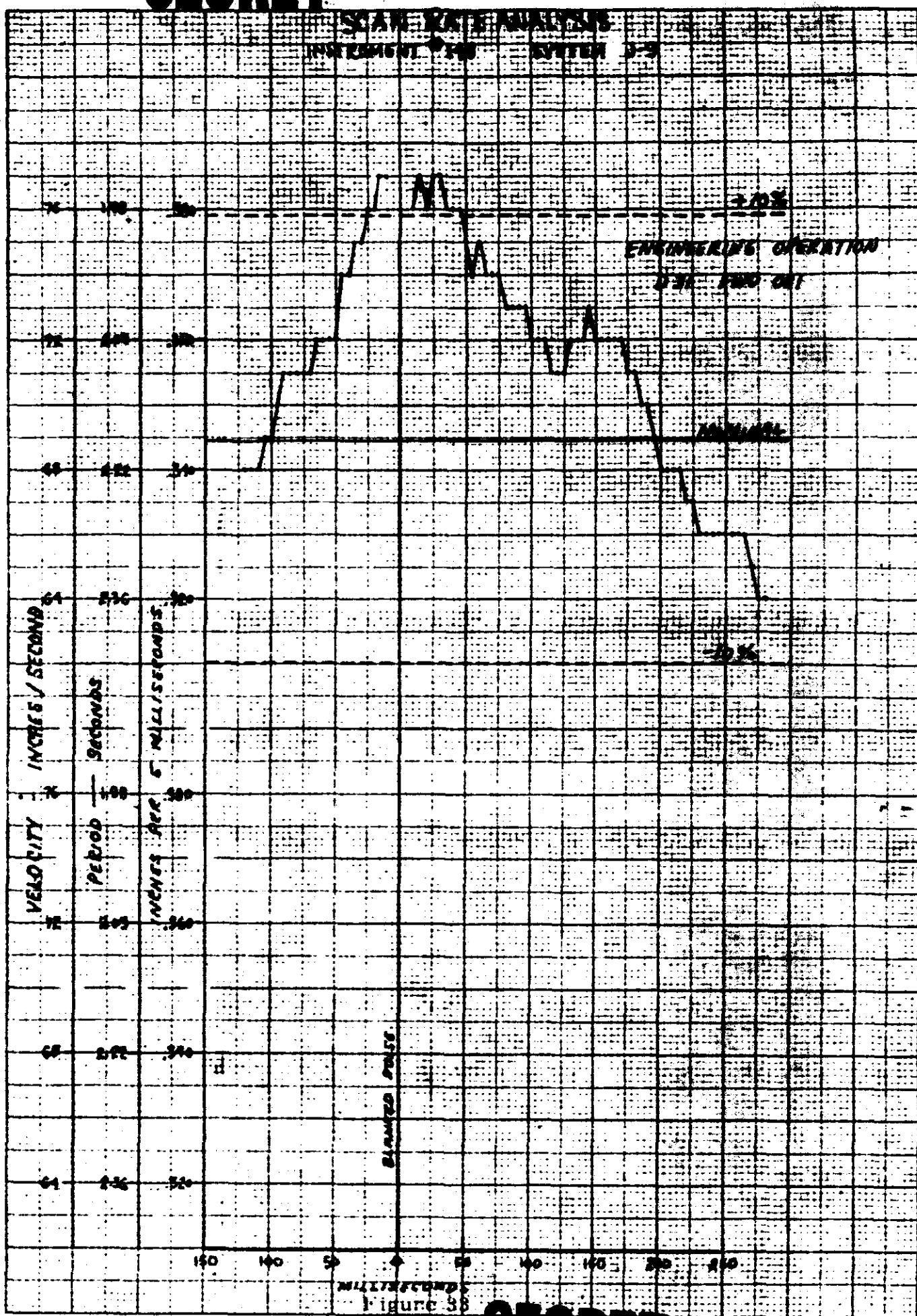
ULVAC

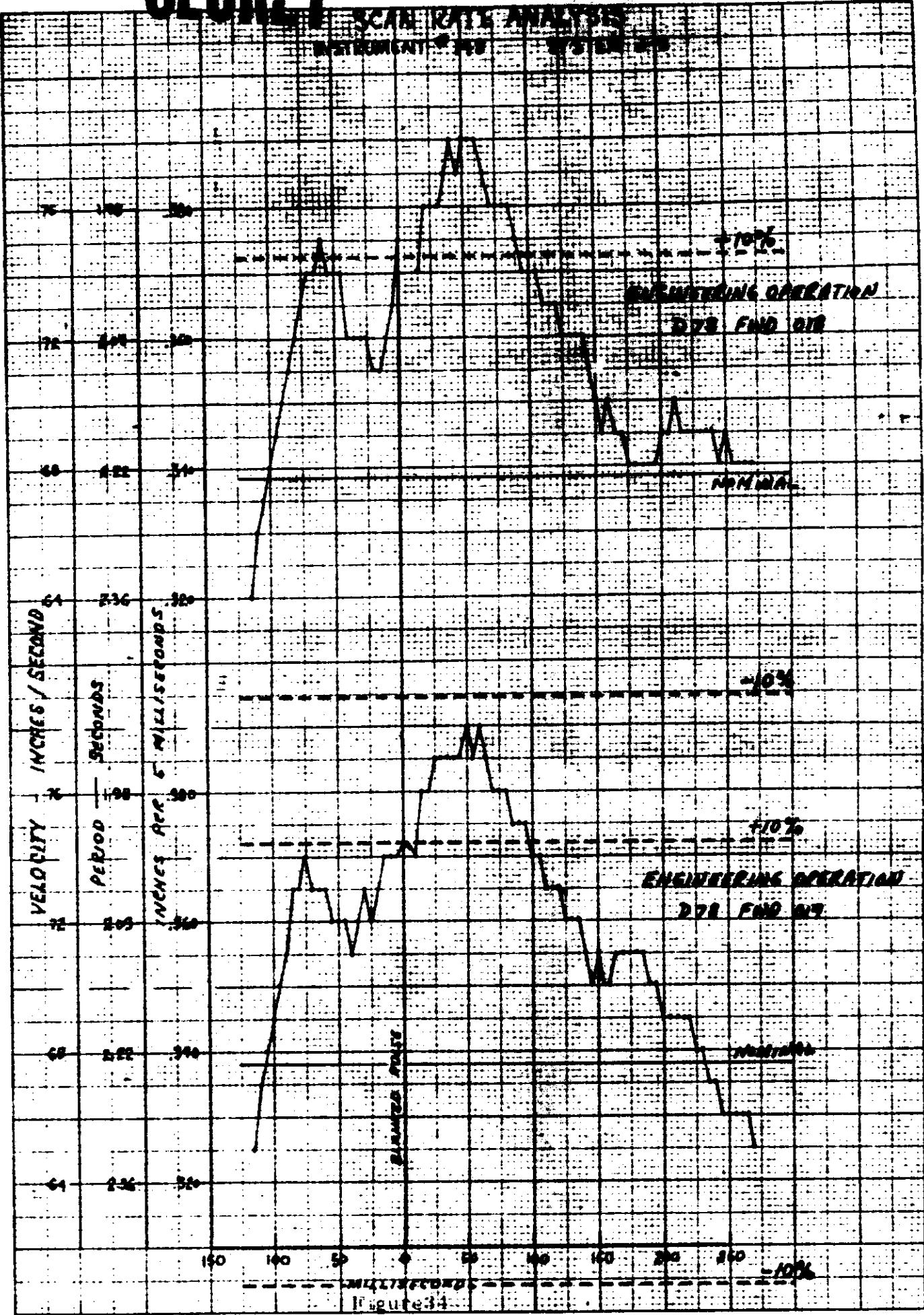
SWING RATE ANALYSIS

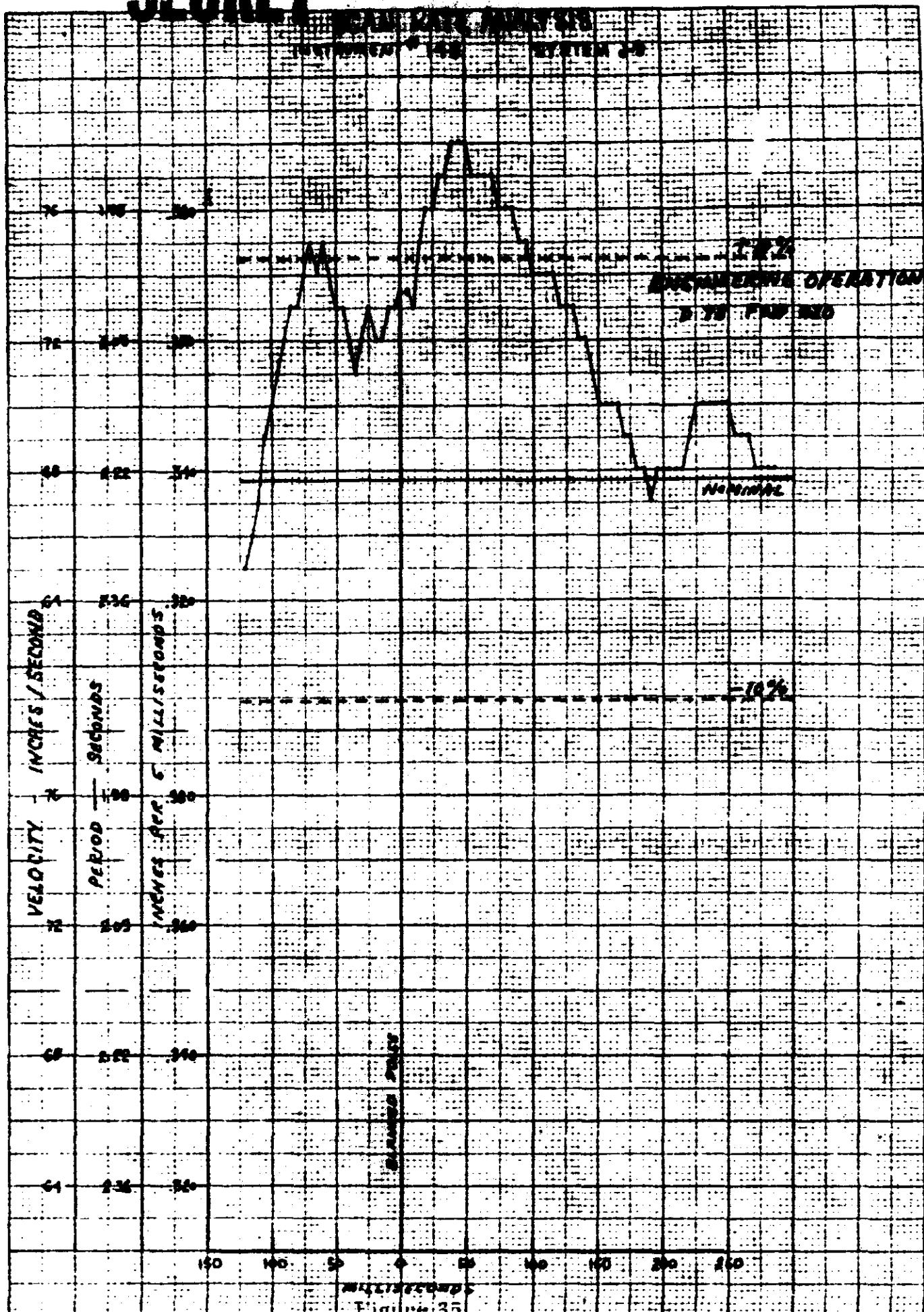
INSTRUMENTATION SYSTEM DATA

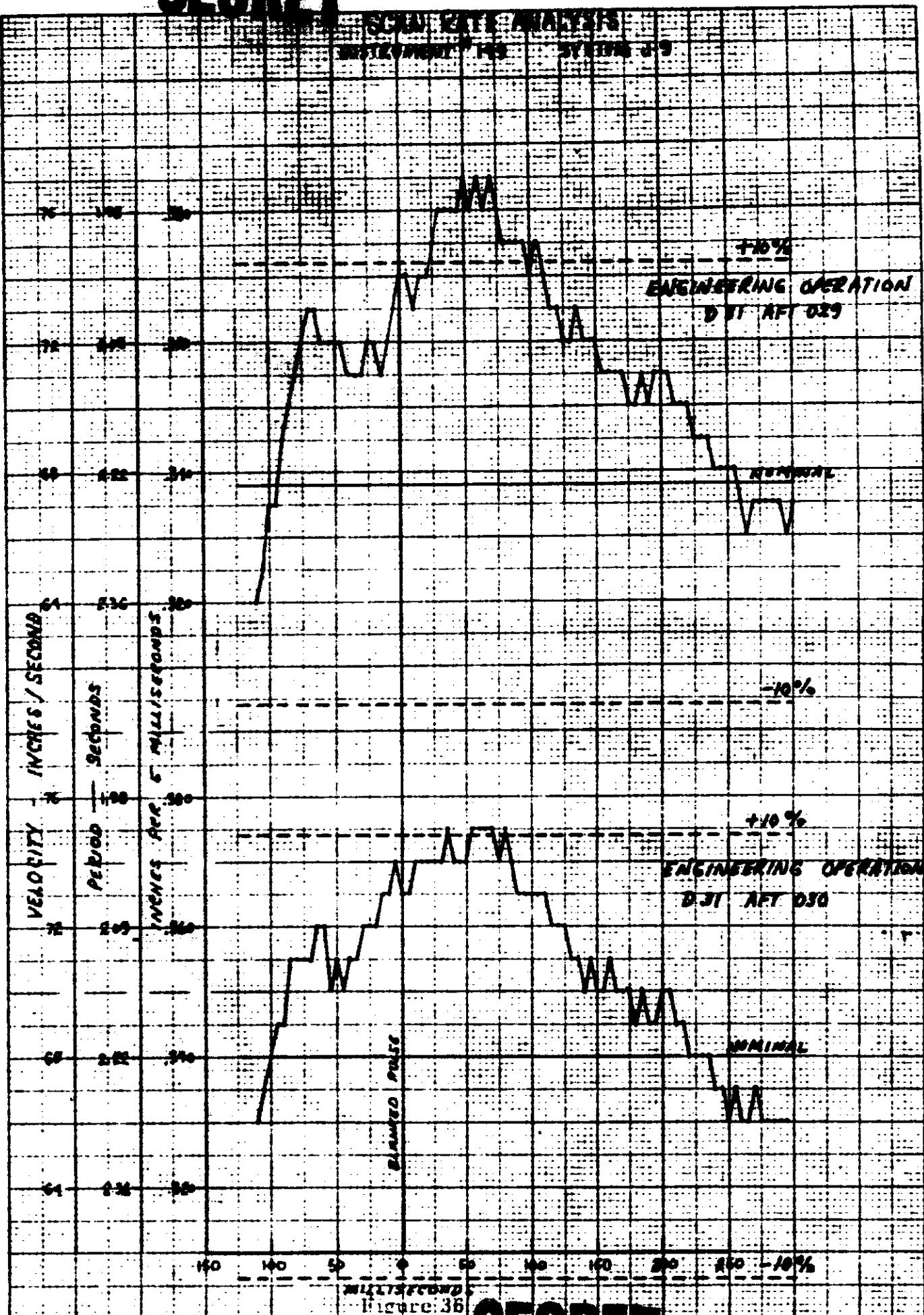


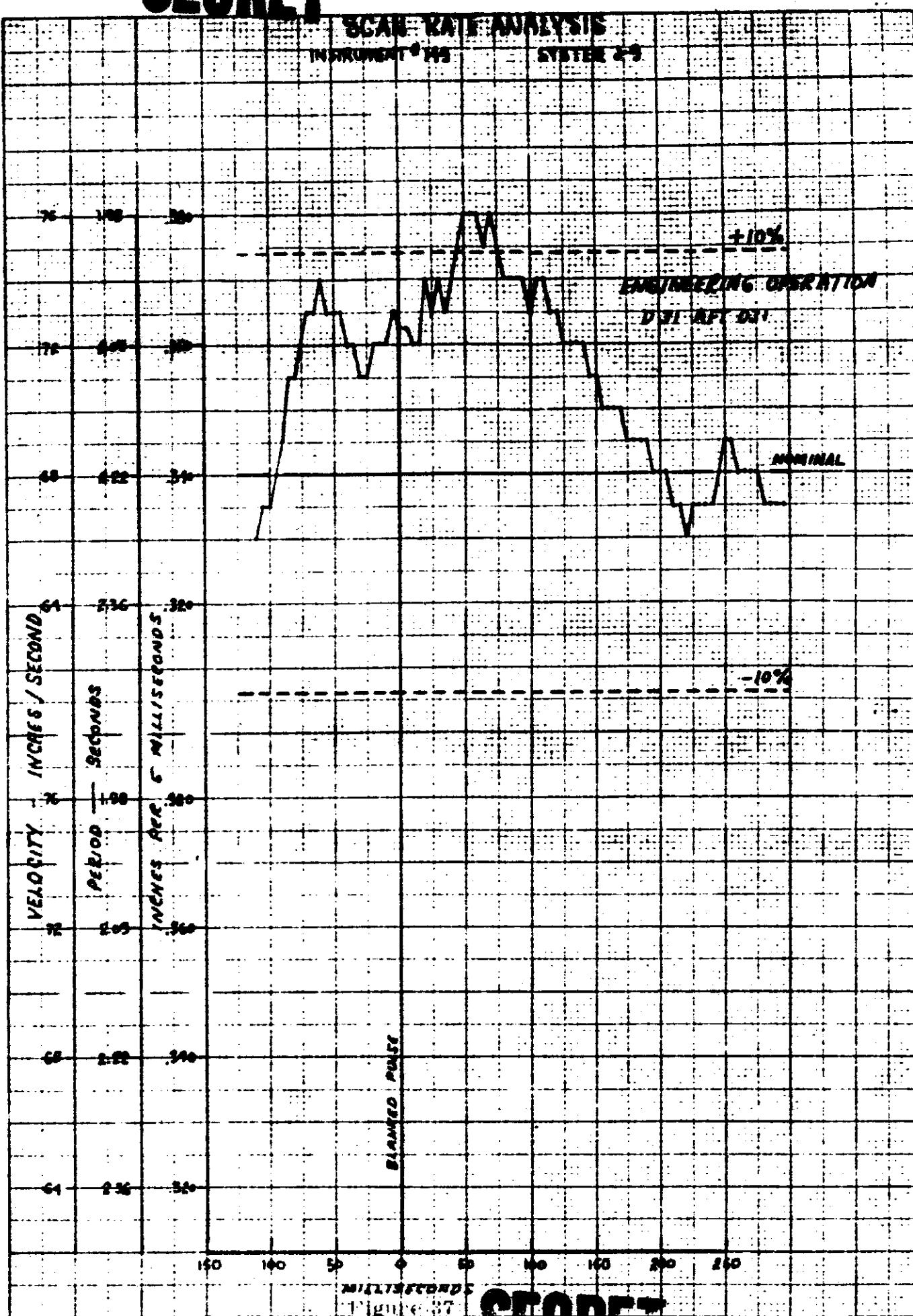












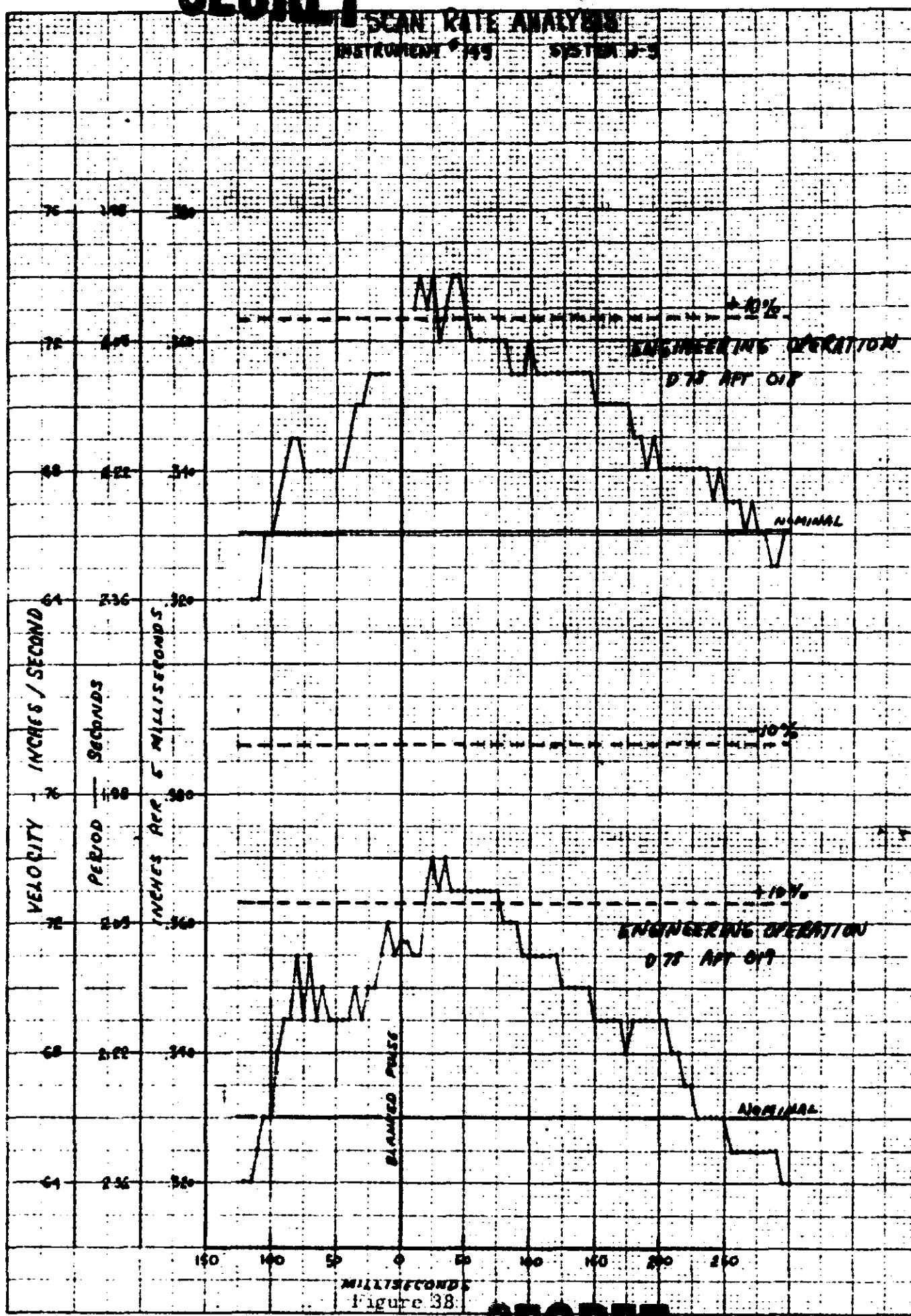
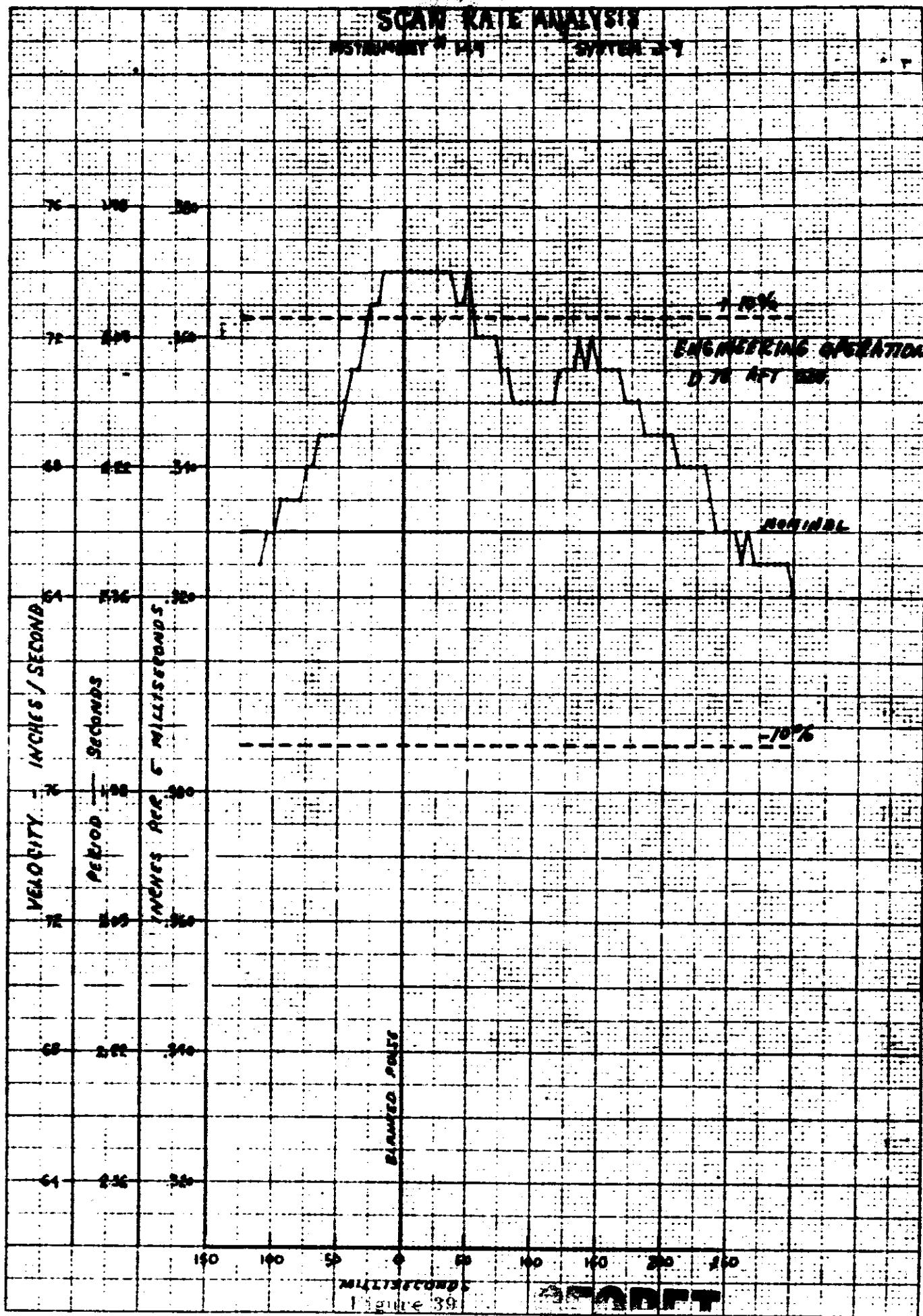


Figure 38.

#### **SCAWK LEAVES**

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- b. Master camera (#148); three consecutive frames of intermediate speed operation during HATS test, Figures 30 and 31.
- c. Master camera (#148); three consecutive frames of high speed operation during Mission 1006-1, Figures 32 and 33.
- d. Master camera (#148); three consecutive frames of high speed operation during Mission 1006-2, Figures 34 and 35.
- e. Slave camera (#149); three consecutive frames of high speed operation during Mission 1006-1, Figures 36 and 37.
- f. Slave camera (#149); three consecutive frames of high speed operation during Mission 1006-2, Figures 38 and 39.

Only high speed operation data is shown on flight film since, except for night engineering operations, nearly all operations for this mission were at high camera speeds due to the abnormal orbit.

The graphs of Figures 28 through 39 show the scan head velocity as measured by the timing mark spacing as a function of scan time. The ordinate or scan time scale is referenced to the blanked pulse which occurs when the center-of-format switch is actuated. The amplitude is a direct measure of the five millisecond incremental scan velocity and is shown as a linear velocity as well as an equivalent cycle period.

#### E. MASTER PANORAMIC CAMERA

Figures 28 and 29 show velocity profiles for four consecutive frames of high speed operation of camera #148 during West Coast high altitude chamber tests. These can be compared with Figures 32 through 35 showing three frame sequences of similar high speed operation of the same camera during Missions 1006-1 and 1006-2.

The first profile of Figure 28 is typical of the high speed velocity profiles collected to date. At the starting edge of the format there is a continuation of the rapid acceleration that should have been completed before the scan head reached the format region. The velocity quickly reaches a value above nominal and continues to increase to some point near the middle of the format, after which it declines gradually to some value below the nominal. Superimposed on the gross pattern is evidence of a secondary cyclical pattern. The frequency of this secondary pattern is about 8 cps which is similar to other instruments examined to date. A third order effect is minor variations observed near

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regions of roughly constant velocity. It may be surmised that such values are due merely to the measurement precision used. However, it has been verified that such regions do contain significant variations in the spaces between successive time marks.

Figures 30 and 31 show velocity profiles for three consecutive frames of intermediate speed operation during West Coast high altitude chamber tests. Here the velocity pattern has changed to show an increase near the end of scan. In these cases the percentage deviation of velocity values is almost the same as in the high speed cases. Operating speeds during Missions 1006-1 and 2 were consistently higher than these test values.

#### F. SLAVE PANORAMIC CAMERA

Figures 36 through 39 show three frame sequences of high speed operation of camera #149 during Missions 1006-1 and 2. Each of these frame sequences was made at the same time as those shown for the master camera in Figures 32 through 35. It will be observed that the velocity profiles of the two cameras are very similar. For the frames shown, the percentage deviation of velocities of the slave camera appear to be slightly smaller.

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